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# NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

TECHNICAL NOTE

No. 1389

DESIGN CHARTS FOR FLAT COMPRESSION PANELS HAVING  
LONGITUDINAL EXTRUDED Y-SECTION STIFFENERS

AND COMPARISON WITH PANELS HAVING  
FORMED Z-SECTION STIFFENERS

By Norris F. Dow and William A. Hickman

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SUMMARY

Design charts are presented for 24S-T (bare sheet) and 75S-T (Alclad sheet) aluminum-alloy flat compression panels with longitudinal extruded Y-section stiffeners. In addition, comparisons are made among panels designed from these charts and 24S-T aluminum-alloy panels having formed Z-section stiffeners designed from available design charts. The comparisons indicate that, if the ratio of intensity of loading to sheet thickness is relatively high, the charts presented may be used to design a Y-stiffened panel in either 24S-T or 75S-T material which is lighter in weight than a 24S-T Z-stiffened panel designed from the available charts to meet the same conditions. The amount of weight saving depends upon the specific design conditions and is greatest for the 75S-T Y-stiffened panels. The comparisons also indicate that the 24S-T Y-stiffener will have a height somewhat greater than the comparable 24S-T Z-stiffener or 75S-T Y-stiffener; the height of the 24S-T Z-stiffener generally is the smallest. In addition, the comparisons indicate that the average spacing of rivet lines is generally somewhat less for the 24S-T Y-stiffened panels than for the 24S-T Z-stiffened panels or for the 75S-T Y-stiffened panels; the average spacing generally is greatest for the 24S-T Z-stiffened panels. If the ratio of intensity of loading to sheet thickness is relatively low, however, the comparative designs indicate that a 24S-T Z-stiffened panel designed from the available charts will be slightly lighter in weight than a Y-stiffened panel of either 24S-T or 75S-T material. If the present design charts are extended to lower values of the ratio of stiffener thickness to sheet thickness to cover the region of heavy sheet thickness more thoroughly (where the ratio of intensity of loading to sheet thickness is relatively low), a Y-stiffened-panel design in this region will probably compare more favorably with a Z-stiffened-panel design than the charts presented indicate. If no sheet thickness is specified so that the design may have optimum proportions, it is concluded that both the 24S-T and 75S-T Y-stiffened panels will be lighter than the 24S-T Z-stiffened panels throughout the range of design conditions investigated.

## INTRODUCTION

The problem of the design of wing compression panels of minimum weight is one that has confronted aircraft structural engineers since the advent of stressed-skin construction. Although the final solution of this problem has not yet been achieved, progress has been made toward its solution as the cumulative result of numerous theoretical and experimental studies to determine "optimum proportions" and "efficient" stiffener shapes.

Recently two such studies (references 1 and 2) have established a type of plot which appears particularly useful in connection with the design of wing compression panels of minimum weight. Reference 1 presented a theoretical comparison of the efficiencies of various stiffener shapes by plotting the average stress at failure - an inverse measure of the weight - against a parameter containing the main design conditions, the load per chordwise inch of panel, and the effective length of panel. Reference 2 used the same type of plot to provide design charts for Z-stiffened panels based on extensive test data from which the optimum proportions can be determined for a particular design.

Study of references 1 and 2 reveals that if panels with longitudinal stiffeners are to have high structural efficiency a stiffener shape is required which has both high-column strength and local-buckling strength. Because a stiffener in the shape of a Y appeared more nearly to meet this requirement than the Z-section or hat-section stiffeners of references 2 and 3, an investigation was made in the Langley structures research laboratory of the National Advisory Committee for Aeronautics to determine the compressive strength of panels having Y-section stiffeners. Both 24S-T (bare sheet) and 75S-T (Alclad sheet) aluminum-alloy panels were tested in this investigation. The results of these tests are presented herein in the form of design charts similar to the design charts for panels with Z-section stiffeners of reference 2.

In order to show the relative structural efficiencies of Y-section and Z-section stiffeners, comparisons are also presented of panels of both types designed to have the minimum weight required to meet a large range of loading conditions.

## SYMBOLS

The symbols used to represent the various dimensions of the panels are shown in figure 1. In addition, the following symbols are used:

$\bar{\sigma}_F$	average stress at failing load, ksi
$\sigma_{cr}$	stress for local buckling of the sheet, ksi
$\sigma_{cy}$	compressive yield stress, ksi
$P_1$	compressive load per inch of panel width, kips per inch
$c$	coefficient of end fixity as used in Euler column formula
$A_1$	cross-sectional area per inch of panel width, or equivalent thickness of panel, inches
$\bar{\epsilon}_F$	shortening per unit length at failing load
$\rho$	radius of gyration, inches
$I_1$	moment of inertia per inch of panel width, cubic inches

## TEST SPECIMENS AND PROCEDURE

The test specimens were constructed with six stiffeners and five bays as shown in figure 1. Three sizes of stiffeners were used that corresponded to values of  $b_W/t_W$  of 20, 25, and 30 with the nominal value of  $t_W$  held constant at 0.064 inch (see fig. 2) and various values of  $t_W/t_S$  were obtained by varying the sheet thickness. The stiffeners were riveted to the sheets with Al7S-T flat-head rivets (AN442AD) on all panels.

Values of the with-grain compressive yield stress for the material used for the sheets (bare sheets were used for the 24S-T panels and Alclad sheets for the 75S-T panels) and for the extrusions are given in table 1. The values of compressive yield stress for the extrusions represent the average values for specimens cut from the three webs and the outstanding flange of the Y-section extrusions at the locations shown in figure 3. Values of the compressive yield stress for the material used to construct the Z-stiffened panels of reference 2 are also given in table 1 for comparison.

The test procedure was essentially the same as that used in other panel tests in the Langley structures research laboratory. (See references 2 and 3.) The panels were tested flat-ended without side support (because their transverse stiffness was small) in a hydraulic testing machine having an accuracy of one-half of 1 percent of the load (See fig. 4.) The stress for local buckling of the sheet was determined by the "strain-reversal method." A discussion of this and other methods of experimentally determining the stress for local buckling is given in reference 4. For panels having a greater width of sheet under the Y's than between the Y's, strain gages were mounted inside the stiffeners, as indicated in figure 5. The ends of the panels were ground flat and parallel, and the method of alignment in the testing machine was such as to insure uniform bearing on the ends of the specimens. An end fixity coefficient of 3.75 has been indicated for such panel tests in this machine, and this value was therefore used in reducing the test data.

Proportions of the specimens and test data - including values of the ratios of rivet diameter to sheet thickness  $d/t_s$  and pitch to sheet thickness  $p/t_s$ , average stress at failing load  $\bar{\sigma}_f$ , and unit shortening at failing load  $\bar{\epsilon}_f$  - are given in tables 2 to 4. The unit shortening was measured as the average of the strains indicated by four,  $6\frac{1}{2}$ -inch gage length, resistance-type wire strain gages mounted on the quarter points of the second and fifth stiffeners, as may be seen in figure 4.

Figure 6 shows a 24S-T aluminum-alloy Y-stiffened panel and its 75S-T counterpart after failure. There tended to be a greater shattering of the 75S-T panels than of the 24S-T panels.

#### DESIGN CHARTS

Design charts for panels with extruded Y-section stiffeners are presented in figures 7 to 11 for 24S-T and in figures 12 to 16 for 75S-T aluminum alloy. These charts were prepared from the test data of tables 2 to 4 in a manner similar to that described in the appendix of reference 2 for Z-stiffened panels. The use of design charts of this type is described fully in reference 2, and a procedure similar to that given in reference 2 for designing a panel for maximum structural efficiency (minimum weight) by use of the charts is included in appendix A of the present paper. This design procedure makes it possible to achieve the balance for given values of  $P_1$ ,  $L/\sqrt{C}$ , and  $t_s$  between the proportions which

will produce the highest average stress at failure and the proportions which will make the area such that the failing stress is just reached at the design load.

A comparison of the curves of the design charts and the test data from which the curves were derived indicates that the stresses given by the curves of the design charts for both the Y-stiffened panels of the present paper and the Z-stiffened panels of reference 2 are on the whole very slightly less than the stresses given by the test data. For both stiffener types, however, there are regions on the design charts in which the curves are interpolated or extrapolated far from the test data, and in these regions the accuracy of the charts is probably less than that indicated by the comparisons of curves and data. The region of the Y-stiffened-panel charts for which there

is the least test data is that for  $\frac{t_w}{t_s} = 0.40$  at wide stiffener

spacings. (See figs. 7 and 12.) Slightly greater caution should be exercised in the use of the charts in this region than elsewhere in the design charts. The region of the Z-stiffened-panel charts for

which there is the least test data is that for  $\frac{t_w}{t_s} = 1.00$  at close

stiffener spacings. In this region, additional unpublished test data have indicated that the curves may be as much as 5 or 6 percent too high.

#### GENERALIZED COMPARISON OF Y-STIFFENED PANELS

#### AND Z-STIFFENED PANELS

Without restrictions on the sheet thickness.- If there are no restrictions on the sheet thickness that may be used, Y-stiffened and Z-stiffened panels may be compared by envelope curves faired over the curves of their design charts. Such a comparison of envelope curves is shown in figure 17. Because the average stresses at failing load for the envelope curves for the Y-stiffened panels of both 24S-T and 75S-T material are above those for the Z-stiffened panels, the Y-stiffened panels of optimum proportions are evidently lighter in weight than the Z-stiffened panels of optimum proportions throughout the range covered by the present design charts.

With restrictions on sheet thickness.- The sheet thickness needed to achieve the stresses of the envelope curves of figure 17 are fixed for any given intensity of loading by the proportions required by the

envelope curves. In the design of wing compression panels, however, the sheet thickness is often fixed by other considerations such as torsional stiffness of the wing. Accordingly, curves which show the effect of a variation in sheet thickness should provide a more useful evaluation of the relative structural efficiencies of Y-stiffened and Z-stiffened panels than do the envelopes of figure 17; therefore, figures 18 to 20 were prepared. In these figures, the average stresses at failure  $\bar{\sigma}_F$  carried by Y-stiffened and Z-stiffened-panel designs, selected for minimum weight according to the procedure given in appendix A, are plotted against the parameter  $\frac{P_1}{t_S}$  for a series of values of  $\frac{P_1}{L/\sqrt{c}}$ . A discussion of this type of plot is given in appendix B.

The chief importance of figures 18 to 20 is that the figures indicate directly the average stress at failure  $\bar{\sigma}_F$  carried by the minimum-weight designs of Y-stiffened or Z-stiffened panels which can be achieved within the large range of proportions covered by the design charts for given values of  $P_1$ ,  $L/\sqrt{c}$ , and  $t_S$ . The effect of a change in any one of the variables  $\bar{\sigma}_F$ ,  $P_1$ ,  $L/\sqrt{c}$ , and  $t_S$  on any of the others, therefore, may be studied from these figures. For example, consider the effect of a change in  $t_S$  on the value of  $\bar{\sigma}_F$ . The relative flatness of the curves at the higher values of  $P_1/t_S$  indicates that the sheet thickness can be varied over a rather large range with very little change in the value of  $\bar{\sigma}_F$  which can be achieved.

A comparison of figures 18 to 20 brings out the following facts:

(1) Minimum-weight designs of both 24S-T and 75S-T Y-stiffened panels are lighter in weight (carry higher stresses) than minimum-weight designs of 24S-T Z-stiffened panels in the region of high values of  $P_1/t_S$  (thin sheet); but the Z-stiffened designs are of slightly lighter weight in the region of low values of  $P_1/t_S$  (thick sheet). No sharply defined boundary exists between these two regions. Instead, there is a range of values of  $P_1/t_S$ , which varies

with  $\frac{P_1}{L/\sqrt{c}}$ , for which the curves of figures 18 to 20 coincide.

(2) The actual amount by which the Y-stiffened-panel design is lighter than the Z-stiffened panel (or vice versa) varies somewhat

erratically as the design conditions  $\frac{P_1}{L/\sqrt{c}}$  and  $\frac{P_1}{t_S}$  are varied because of the cusped nature of the curves.

(3) The value of  $\frac{t_W}{t_S}$ , which produces the minimum-weight design for given values of  $\frac{P_1}{L/\sqrt{c}}$  and  $\frac{P_1}{t_S}$ , is smallest for the 75S-T Y-stiffened panels and largest for the 24S-T Z-stiffened panels.

#### COMPARISON OF MINIMUM-WEIGHT DESIGNS OF Y-STIFFENED PANELS AND Z-STIFFENED PANELS

Although figures 18 to 20 show in a general way the relative structural efficiencies of Y- and Z-stiffened panels, probably the best way to evaluate two types of panel construction is to compare panels of each type designed to meet the same conditions. A comparison of this nature permits consideration of each of the many factors which influence the choice of the most desirable construction for a given situation, such as the number of rivet lines, the space required for the stiffeners, and the distance from the outside surface of the sheet to the axis of the center of gravity of the panel. A series of comparative designs of Y- and Z-stiffened panels, therefore, was made in a manner similar to that used in making the designs from which figures 18 to 20 were prepared. Four values of  $P_1$ , namely, 2.0, 3.0, 5.0, and 8.0 kips per inch and also four values of  $L$ , namely, 10, 20, 30, and 40 inches were used for the comparative designs. The end fixity coefficient  $c$  was assumed equal to 1 in all cases.

In making the comparative designs, obtainment of extruded Y-stiffeners in the thicknesses required by the designs was assumed possible. A minimum thickness in which these shapes can be successfully extruded exists, however, and this minimum thickness is probably above the thickness required for many of the designs for which  $P_1$  is equal to or less than 3.0 kips per inch. The reasons for retaining these designs are (1) they may be scaled up for higher intensities of loading for which the minimum thickness that can be extruded is no longer a limitation, and (2) to emphasize the fact that if the intensity of loading is low, the Y-stiffener will not be satisfactory simply because it cannot be obtained.

Numerical values of the properties of the comparative designs for all values of  $t_W/t_S$  covered by the design charts are given in tables 5 to 8. The values for the particular ratio of  $t_W/t_S$  for



which minimum weight is achieved are enclosed in parentheses. In order to show graphically the general variation of the proportions of these designs as the panel length and the sheet thickness are varied, figures 21 and 22 have been prepared. These figures present cross-sectional views, drawn to scale, of some of the minimum-weight designs of Y-stiffened and Z-stiffened panels for  $P_1 = 5.0$  kips per inch (table 7).

The comparative designs were made according to the procedure given in appendix A except that all values of  $t_w/t_s$  given by the design charts were investigated for each design. Because the design charts cover only a limited range of proportions, the comparisons between the designs are in some cases affected by the limited range of proportions covered by the charts. With this qualification, comparisons of the minimum-weight designs of tables 5 to 8 and figures 21 and 22 show that:

(1) At relatively high values of  $P_1/t_s$ , which are associated with thin sheets, the average stresses at failure  $\bar{\sigma}_F$  for both the 24S-T and the 75S-T Y-stiffened panels are greater than those for the 24S-T Z-stiffened panels, and these stresses indicate that less weight is required for the Y-stiffened than for the Z-stiffened panels, the least weight being required for the 75S-T Y-stiffened panels. On the other hand, at relatively low values of  $P_1/t_s$ , which are associated with thick sheets, the average stresses at failure  $\bar{\sigma}_F$  for both the 24S-T and 75S-T Y-stiffened panels are slightly less than those for the 24S-T Z-stiffened panels, and these stresses indicate that the Z-stiffened panel is slightly lighter in weight. The magnitude of the difference in weight between the two types of panel varies with the values of  $P_1/t_s$  and  $\frac{P_1}{L/\sqrt{c}}$ .

(2) The height of the stiffeners  $H$  is generally somewhat greater and, hence, consumes more space inside the wing for the 24S-T Y-stiffened panels than for the 24S-T Z-stiffened panels or for the 75S-T Y-stiffened panels; the height of the 24S-T Z-stiffened panel generally is the smallest.

(3) The average spacing of rivet lines  $S$  is generally somewhat less and, hence, requires more rivets for the 24S-T Y-stiffened panels than for the 24S-T Z-stiffened panels or for the 75S-T Y-stiffened panels; the average spacing generally is greatest for the 24S-T Z-stiffened panels.

(4) Only if the values of both  $P_1/t_S$  and  $\frac{P_1}{L/\sqrt{C}}$  are relatively high does the value of the stress for local buckling of the sheet  $\sigma_{cr}$  tend to be higher for the 24S-T or 75S-T Y-stiffened panels than for the 24S-T Z-stiffened panels.

(5) The distance from the outside surface of the sheet to the axis of the center of gravity of the panel  $\bar{h}$ , which tends to reduce the effectiveness of the panel to resist bending of the wing, is generally greater for the 24S-T Y-stiffened panels than for the 24S-T Z-stiffened panels or for the 75S-T Y-stiffened panels; the distance  $\bar{h}$  generally tends to be smallest for the 24S-T Z-stiffened panels at low values of  $P_1/t_S$  (thick sheet) and smallest for the 75S-T Y-stiffened panels at high values of  $P_1/t_S$  (thin sheet). (The magnitude of the reduction in effectiveness of the panel to resist bending of the wing depends on the thickness of the wing. The thinner the wing, the greater the reduction.)

(6) The value of the radius of gyration  $\rho$  is generally greater (and also the value of  $\rho^2 A_1 = I_1$  is generally greater) for the 24S-T Y-stiffened panel than for the 24S-T Z-stiffened panel or for the 75S-T Y-stiffened panel; generally,  $\rho$  tends to be smallest for the 24S-T Z-stiffened panels at low values of  $P_1/t_S$  (thick sheet) and smallest for the 75S-T Y-stiffened panels at high values of  $P_1/t_S$  (thin sheet). (The greater the value of  $\rho^2 A_1$ , the greater the effectiveness of the panel to resist local air loads.)

#### EFFECT OF SMALL DIFFERENCES IN TEST SPECIMENS ON THE COMPARISONS OF 24S-T Y-STIFFENED AND Z-STIFFENED PANELS

Only small differences occurred between the test specimens for the 24S-T Y-stiffened and Z-stiffened panels. Differences occurred in material properties, diameter and pitch of rivets, and range of proportions of the elements of the panels actually tested and hence the proportions covered by the resulting design charts.

The effect of these differences on the comparisons of 24S-T Y-stiffened and Z-stiffened panels are discussed in the following sections.

Effect of material properties. - If the material properties of the Y-stiffened panels and the Z-stiffened panels had been identical, would the comparisons have been more or less favorable to the Y-stiffened panels? Table 1 indicates that the average compressive

yield stress of the material used for the Z-stiffened panels and that of the material used for the sheets of the 24S-T Y-stiffened panels were identical but that the average compressive yield stress of the extruded Y-stiffeners as measured was between 3 and 4 percent less than that of the Z-stiffeners before forming. Because forming tends to raise the compressive yield stress (see reference 5), the average properties of the formed Z-stiffeners were probably more than 3 percent above those for the extruded Y-stiffeners. Accordingly, it may be inferred that if the Y-stiffeners and the Z-stiffeners had had identical properties - as might have been obtained if extruded Z-stiffeners had been used, for example - the 24S-T Y-stiffened panels tested would have increased in strength relative to the Z-stiffened panels, and the comparisons would have been more favorable to the Y-stiffened panels.

Effect of riveting. - If the riveting of the Y-stiffened panels and the Z-stiffened panels had been identical, would the comparisons have been more or less favorable to the Y-stiffened panels? A comparison of rivet proportions listed in tables 2 to 4 with those of reference 2 indicates that the Y-stiffened panels were more strongly riveted than the Z-stiffened panels. Reference 6 shows that the strength of short panels having close stiffener spacing increased with an increase in the diameter of the rivets and also increased with a decrease in the pitch of the rivets. Subsequent tests have indicated that as the length of the panel is increased the size and pitch of rivets have progressively less effect on the strength of the panel until the panel strength may actually decrease with an increase in the strength of riveting. If the Y-stiffened panels and Z-stiffened panels had had identical riveting, therefore, the comparative designs would probably have come out less favorable to the Y-stiffened panel

in the case of the short panels (high values of  $\frac{P_1}{L/\sqrt{C}}$ ) and possibly very slightly more favorable to the Y-stiffened panel in the case of the long panels (low values of  $\frac{P_1}{L/\sqrt{C}}$ ).

Effect of panel proportions. - If proportions of Y-stiffened panels or Z-stiffened panels different from those tested and, hence, those covered by the resulting design charts had been considered, would the comparisons have been more or less favorable to the Y-stiffened panels? It can be seen by inspection of tables 5 to 8 that:

(1) The lightest weight 24S-T Y-stiffened-panel design for a given set of design conditions often requires a stiffener which is 2 sheet gages thinner than that for the comparable Z-stiffened-

panel design. (This agrees with the fact that the value of  $t_W/t_S$  for minimum weight is smaller for the Y-stiffened panel than for the Z-stiffened panel. See figs. 18 to 20.) Also, the present charts do not cover a large enough range of proportions to permit a Y-stiffener more than 1 gage thinner than a Z-stiffener in the region of heavy sheet thickness. If the design charts were extended to cover lower values of the ratio  $t_W/t_S$  so that a Y-stiffened-panel design could always be made which had a stiffener 2 gages thinner than the stiffener for the best Z-stiffened-panel design, then the Y-stiffened-panel design would probably be less inferior to the Z-stiffened-panel design in the region of heavy sheet thickness. Similarly, if the charts were extended in the other direction so that in all cases a Z-stiffened-panel design with a Z-stiffener 2 sheet gages thicker than the comparable Y-stiffened-panel design could be made, possibly the Y-stiffened panel would be less superior to the Z-stiffened panel in the region of very light sheet thickness.

(2) The lightest weight Y-stiffened-panel designs - in far more cases than for the Z-stiffened-panel designs - are obtained at the maximum or minimum values of  $b_W/t_W$  given by the design charts. Extending the range of proportions covered to higher and lower values of  $b_W/t_W$  would be likely, therefore, to result in lighter weight designs of Y-stiffened panels in more cases than in lighter designs of Z-stiffened panels.

Because a very extensive test program was run to establish optimum proportions for the Z-stiffener ( $\frac{b_F}{b_W} = 0.3 \text{ to } 0.5$ ), and no such program has been run to establish optimum proportions for the Y-stiffener, the proportions of the Y-stiffener possibly could be improved and, hence, the comparative designs made more favorable to the Y-stiffened panel for all sheet thicknesses. Among the changes in proportions of the Y-stiffened panels which might result in overall improvements in their structural efficiencies are: (1) a change in the angle included between the legs of the Y-stiffeners in order to effect a better balance between the width of sheet under the Y-stiffeners and between adjacent Y-stiffeners, (2) a change in relative proportions of the outstanding "T" part of the Y-stiffeners, and (3) a reduction in the width of attachment flanges of the Y-stiffeners, particularly for  $\frac{t_W}{t_S} = 1.00$ .

## GENERAL TRENDS INDICATED BY MINIMUM-WEIGHT DESIGNS

In addition to the comparisons of Y-stiffened and Z-stiffened panels afforded by the designs of tables 5 to 8 and figures 21 and 22, there are several general trends indicated by the designs and by figures 18 to 20 which apply to both types of construction. These general trends are in some cases affected by the limited range of proportions covered by the present design charts. These trends as well as the comparisons between the two types of construction, are also strictly for minimum-weight designs. With the foregoing qualifications, the comparative designs show that:

For given values of  $P_1$  and  $L/\sqrt{C}$

- (1) The weight of panel generally increases ( $\bar{\sigma}_F$  decreases) with an increase in sheet thickness, but the lightest panel is often obtained not at the thinnest sheet gage at which a design can be achieved but with the sheet 1 or 2 gages thicker than the minimum.
- (2) The stress for local buckling of the sheet  $\sigma_{cr}$  and also the ratio  $\sigma_{cr}/\bar{\sigma}_F$  generally decreases with an increase in sheet thickness, but the maximum value of the stress for local buckling of the sheet is often obtained not at the thinnest sheet gage at which a design can be achieved but with the sheet 1 or 2 gages thicker than the minimum.
- (3) The average spacing of rivet lines  $S$  increases (requiring fewer rivets) with an increase in sheet thickness.
- (4) The distance from the outside surface of the sheet to the axis of the center of gravity of the panel  $\bar{h}$ , which tends to decrease the effectiveness of the panel to resist bending of the wing, generally decreases with an increase in sheet thickness.

And for given values of  $P_1$  and  $t_S$

- (1) The weight of panel increases ( $\bar{\sigma}_F$  decreases) with an increase in the value of  $L/\sqrt{C}$ .
- (2) The stress for local buckling of the sheet  $\sigma_{cr}$ , but not necessarily the ratio  $\sigma_{cr}/\bar{\sigma}_F$ , generally decreases with an increase in the value of  $L/\sqrt{C}$ , except at the heavy sheet thicknesses.
- (3) The height of the stiffeners  $H$  increases with an increase in the value of  $L/\sqrt{C}$ .

(4) The average spacing of rivet lines  $S$  generally increases (again requiring fewer rivets) with an increase in the value of  $L/\sqrt{C}$ , except at the heavy sheet thicknesses.

(5) The distance from the outside surface of the sheet to the axis of the center of gravity of the panel  $h$ , which tends to decrease the effectiveness of the panel to resist bending of the wing, generally increases with an increase in the value of  $L/\sqrt{C}$ .

(6) The radius of gyration  $\rho$  increases (not necessarily increasing the effectiveness of the panel to resist local air loads) with an increase in the value of  $L/\sqrt{C}$ .

#### CONCLUDING REMARKS

In this paper, charts have been presented from which 24S-T (bare sheet) and 75S-T (Alclad sheet) aluminum-alloy flat compression panels having longitudinal extruded Y-section stiffeners may be designed to have the minimum weight required to carry a given intensity of loading at a given effective length of panel with a given sheet thickness. Comparisons have been made of panels designed from these charts and similar designs of Z-stiffened panels, in order to bring out the differences in characteristics of 24S-T and 75S-T and of Y- and Z-stiffened-panel designs. In the case of actual wing compression panels, however, there are often additional factors to be considered which have been neglected for the comparisons, such as the effects of local air loads, the distance from the neutral axis of the wing to the center of gravity of the cross section of the panel, the sheet curvature, the edge support, and the shear combined with the compression, or the effects on the design procedure of specifying stiffener height or spacing in addition to sheet thickness. The labor involved in the introduction of so many additional variables into the comparisons, however, is obviously prohibitive. In fact some of the variables cannot be introduced because the necessary research has not been done. Because in any particular design some such additional factor may be important, the choice of a type of construction in most cases is best made by evaluating the characteristics of panels of several types designed to meet all the requirements of the actual application. The design charts of the present paper (figs. 7 to 17) together with the tables of section properties (tables 9 to 13) may be used as an aid in such an evaluation of the characteristics of a 24S-T or 75S-T Y-stiffened panel.

## APPENDIX A

## METHOD OF DESIGNING A Y-STIFFENED PANEL FOR MINIMUM WEIGHT

The following procedure, which is similar to that given in reference 2 for Z-stiffened panels, permits the selection of the minimum-weight Y-stiffened panel for given values of the design conditions  $P_1$ ,  $L/\sqrt{c}$ , and  $t_S$ . In this procedure, the conditions  $P_1$ ,  $L/\sqrt{c}$ , and  $t_S$  are first combined to determine the values of the parameters  $\frac{P_1}{L/\sqrt{c}}$  and  $\frac{P_1}{t_S}$ . Next, from figures 18 or 20 the value of  $t_W/t_S$  is found for which the minimum-weight design will be achieved. Then a study is made of all the curves of the design chart for that ratio of  $t_W/t_S$  at the given value of  $\frac{P_1}{L/\sqrt{c}}$ . From this study, a plot is made of the variation of the stress at failure with stiffener spacing for panels having all the proportions covered by the chart. Because the chart gives  $\bar{\sigma}_F$  in terms of relative proportions (dimension ratios), the absolute size is established for each set of panel proportions by computing the sheet thickness required to make the design load  $P_1$  divided by the area  $A_1$  equal to the failing stress  $\bar{\sigma}_F$ . The variation of these sheet thicknesses, calculated as  $\frac{P_1}{\bar{\sigma}_F \frac{A_1}{t_S}}$  is then plotted against stiffener spacing. This second plot makes the establishment of stiffener spacings associated with the design value of the sheet thickness for each of the panel proportions possible. Reference to the first plot permits the determination of the stresses corresponding to these proportions and the selection of the proportions (usually by interpolation) which give the highest stress (minimum weight) at the given sheet thickness.

As an example of this procedure, the values and quantities for the 24S-T design shown in figure 23 for  $P_1 = 5.0$  kips per inch,  $L = 20$  inches,  $c = 1$ , and  $t_S = 0.102$  inch are given in table 14 and are employed in the following steps:

- (1) Compute  $\frac{P_1}{L/\sqrt{c}}$  and  $\frac{P_1}{t_S}$ .

(2) From figures 18 or 20 (in the example, fig. 18 for 24S-T is used) determine the value or values of  $t_W/t_S$  which should be investigated to find the minimum-weight design at the values of  $\frac{P_1}{L/\sqrt{c}}$  and  $\frac{P_1}{t_S}$  determined in step (1) (in the example,  $\frac{t_W}{t_S} = 0.40$ ).

(3) From the curves for the particular value of  $t_W/t_S$  determined in step (2) (in the example, fig. 7), pick off for each value of  $b_W/t_W$  and  $b_S/t_S$  the value of  $\bar{\sigma}_F$  corresponding to the value of  $\frac{P_1}{L/\sqrt{c}}$  given by step (1).

(4) Pick from tables 9 to 13 (in the example, table 9) the values of  $A_1/t_S$  corresponding to the ratios used in step (3).

(5) Compute the sheet thickness that would be required to make the design load  $P_1$  divided by the area  $A_1$  equal to the failing stress  $\bar{\sigma}_F$  in each case, thus  $t_S = \frac{P_1}{\bar{\sigma}_F \frac{A_1}{t_S}}$ .

(6) Plot the values of  $\frac{P_1}{\bar{\sigma}_F \frac{A_1}{t_S}}$  and  $\bar{\sigma}_F$  against  $b_S/t_S$  for each value of  $b_W/t_W$  and mark the values of  $\bar{\sigma}_F$  at the value of  $b_S/t_S$  for which  $\frac{P_1}{\bar{\sigma}_F \frac{A_1}{t_S}}$  equals the design value of  $t_S$  (in the example, 0.102 in.). The plots of this step for the example under consideration are given as the two lower plots in figure 23. For ease in interpolating to find the value of  $b_W/t_W$  for the design, a curve of  $b_W/t_W$  against  $b_S/t_S$  is also conveniently established by plotting the consecutive values of  $b_W/t_W$  (18, 21, 24, and so forth) at the values of  $b_S/t_S$  for which  $\frac{P_1}{\bar{\sigma}_F \frac{A_1}{t_S}}$  equals the design value of  $t_S$  (the upper plot in fig. 23).

(7) After step (6) has been completed for all the values of  $b_W/t_W$ , draw curves of stress and of  $b_W/t_W$  against  $b_S/t_S$  through the points determined in step (5) (heavy curves in fig. 23).



(8) Each of the curves drawn in step (7) represents a series of designs, all of which have the required value of  $t_S$  (in the example, 0.102 in.). The maximum point on the curve of  $\bar{\sigma}_F$  against  $b_S/t_S$  indicates the design for minimum weight. Note this maximum value of  $\bar{\sigma}_F$ , the value of  $b_S/t_S$  at which it is reached, and the corresponding value of  $b_W/t_W$  which can be picked from the curve of  $b_W/t_W$  against  $b_S/t_S$ .

(9) Check computations by picking from tables 9 to 13 the value of  $A_1/t_S$  corresponding to the ratios selected for minimum weight in step (8). If computations and plots are correct,

$$P_i = \bar{\sigma}_F \frac{A_1}{t_S}$$

(10) Compute the following panel dimensions from the proportions determined by this design procedure with the aid of tables 9 to 13:

$$t_W = \frac{t_W}{t_S} t_S$$

$$t_L = \frac{t_L}{t_W} t_W$$

$$b_S = \frac{b_S}{t_S} t_S$$

$$b_L = \frac{b_L}{b_W} b_W$$

$$b_W = \frac{b_W}{t_W} t_W$$

$$t_F = \frac{t_F}{t_W} t_W$$

$$b_A = \frac{b_A}{t_W} t_W$$

$$b_F = \frac{b_F}{b_W} b_W$$

$$b_Y = \frac{b_Y}{b_W} b_W$$

$$r = \frac{r}{t_W} t_W$$

$$H = \left( 1.79 \frac{b_W}{t_W} + 1.6 \right) t_W$$

$$S = 0.5 \frac{b_S}{t_S} t_S + \left( 0.52 \frac{b_W}{t_W} + 2.3 \right) t_W$$

$$\bar{h} = \frac{\bar{h}}{t_S} t_S$$

$$\rho = \frac{\rho}{t_S} t_S$$

(11) Compute the diameter and pitch of rivets from the proportions listed in tables 9 to 13, as

$$d = \frac{d}{t_S} t_S$$

$$p = \frac{p}{t_S} t_S$$

(12) Find  $\sigma_{cr}$  by interpolation between the short horizontal lines in figures 7 to 16.

If the values of  $\frac{P_1}{L/\sqrt{c}}$  and  $\frac{P_1}{t_S}$  computed in step (1) are such that the point on figure 18 or 20 corresponding to these values is near a boundary between two values of  $t_W/t_S$ , it is advisable to follow the design procedure of steps (1) to (12) for both values of  $t_W/t_S$ .

## APPENDIX B

## DESIGN CHART FOR DETERMINING THE STRUCTURAL EFFICIENCY

If a chart is to be drawn which will provide a direct measure of the structural efficiency of a wing compression panel, that chart must contain in its parameters all the design conditions which apply to the panel. In references 1 and 2 the parameter  $\frac{P_1}{L/\sqrt{G}}$ , which contains the design conditions of compressive load and effective length of panel, was used for charts that measure directly the structural efficiency when those are the design conditions.

The trend toward higher speeds and thinner wings and the accompanying requirement of high torsional stiffness, however, tends to establish a minimum acceptable sheet thickness for the panel. It therefore appears desirable to include the sheet thickness  $t_s$  within the parameters used for preparing charts indicative of the structural efficiency of panels.

A suitable parameter incorporating the sheet thickness appears to be  $P_1/t_s$ . This parameter, which represents the load divided by the area of sheet alone, denotes the upper limit of stress that can be carried by a panel for a given sheet thickness because any stiffeners added to the sheet must increase the panel area and reduce the stress below that determined as  $P_1/t_s$ . This upper limit is shown in figures 18 to 20 as the dashed line. Besides indicating the upper limit of stress, this line also represents the stress that would be carried by a panel having a value of  $\frac{t_w}{t_s} = 0$  (pure shell construction), but only if such a panel could actually carry the indicated stress without failing.

As the value of  $t_w/t_s$  for the panel is increased from zero, the stress carried will decrease from that equal to the value of  $P_1/t_s$ . The actual magnitude of the highest stress that can be achieved for each value of  $t_w/t_s$  given by the design charts can be determined by assuming values of  $\frac{P_1}{L/\sqrt{G}}$  and  $\frac{P_1}{t_s}$  and by examining all the

individual curves of the design charts in a manner similar to the minimum-weight-design procedure at the assumed values of  $t_w/t_s$ ,  $\frac{P_1}{L/\sqrt{c}}$ , and  $\frac{P_1}{t_s}$ . (Values of  $\bar{\sigma}_f \frac{A_1}{t_s}$ , which are equal to  $\frac{P_1}{t_s}$ , instead of

$\frac{P_1}{\bar{\sigma}_f \frac{A_1}{t_s}}$  are computed in step (5) and plotted in step (6) of the

procedure. See appendix A. Also designs are made for a series of values of  $P_1/t_s$ , corresponding to a series of design values of  $t_s$ , from each plot of step (6).)

The foregoing procedure was used to establish the curves given in figures 18 to 20, which indicate the stresses attainable by minimum-weight designs as  $P_1/t_s$  is varied for chosen values of

$\frac{P_1}{L/\sqrt{c}}$ . The stress for any point on one of these curves is therefore

a direct measure of the structural efficiency of the best design that can be made to meet the given design conditions  $P_1$ ,  $L/\sqrt{c}$ , and  $t_s$ .

Because the design charts are drawn for definite values of  $t_w/t_s$ , the curves of figures 18 to 20 contain cusps which correspond to the intersection of the curves resulting from the use of the design charts for consecutive values of  $t_w/t_s$ . Light lines have been drawn in figures 18 to 20 connecting these cusps, thus dividing the figures into regions in which the indicated values of  $t_w/t_s$  produce the minimum-weight designs. As previously noted, the region for  $\frac{t_w}{t_s} = 0$  is the dashed line, for which  $\bar{\sigma}_f = \frac{P_1}{t_s}$ .

For given values of  $P_1$ ,  $L/\sqrt{c}$ , and  $t_s$ , the value of  $t_w/t_s$  that will produce the lightest weight Y-stiffened or Z-stiffened panel may be determined directly from figures 18 to 20. Since very slight variations in  $\bar{\sigma}_f$  near the cusps of the curves could cause an appreciable shift in the location of the cusps in many cases, the light lines should be considered as only approximate boundaries. If the point corresponding to a particular design being considered lies near a boundary between two values of  $t_w/t_s$ , it might be wise to investigate both values of  $t_w/t_s$  in making that design.

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TABLE 1

VALUES OF THE COMPRESSIVE YIELD STRESS FOR THE MATERIALS  
 USED FOR CONSTRUCTING THE Z-STIFFENED PANELS  
 AND THE Y-STIFFENED PANELS

	$\sigma_{cy}$ (ksi)	Sheet (bare)	Stiffeners (bare sheet before forming)
24S-T Z-stiffened panels (from reference 2)	Maximum	46.5	46.5
	Average	44.0	44.0
	Minimum	41.0	41.0
24S-T Y-stiffened panels	$\sigma_{cy}$ (ksi)	Sheet (bare)	Stiffeners (extrusions)
	Maximum	47.3	48.0
	Average	44.0	42.3
	Minimum	42.4	38.4
75S-T Y-stiffened panels	$\sigma_{cy}$ (ksi)	Sheet (Alclad)	Stiffeners (extrusions)
	Maximum	69.7	86.5
	Average	67.3	78.2
	Minimum	64.7	67.6

TABLE 2

TEST DATA AND PROPORTIONS OF SPECIMENS HAVING  $\frac{t_w}{t_s} = 0.40$ 

[Nominal proportions are given in parentheses]

(a) 24S-T SHEET AND STIFFENERS

Proportions of test specimens													Test data				
$t_w$ (in.)	$\frac{t_w}{t_s}$	$\frac{b_s}{t_s}$	$\frac{b_w}{t_w}$	$\frac{b_A}{t_w}$	$\frac{b_W}{t_W}$	$\frac{t_w}{t_L}$	$\frac{b_w}{b_L}$	$\frac{t_w}{t_F}$	$\frac{b_w}{b_F}$	$\frac{r}{t_w}$	$\frac{d}{t_s}$	$\frac{p}{t_s}$	$\frac{L}{b_w}$ (a)	$\sigma_{or}$ (ksi)	$\sigma_f$ (ksi)	$\frac{P_f}{L/\sqrt{a}}$ (ksi/in.)	$\bar{\epsilon}_f$
(0.064)	(0.40)	(25)	(20)	(9.3)	(0.96)	(0.94)	(1.07)	(0.47)	(1.14)	(1)	(15.4)	(4.6)	12.5				
0.064	0.399	25.0	20.0	9.30		0.921		0.487					24.9	40.7	42.7	1.222	$672 \times 10^{-5}$
.066	.409	25.3	19.5	9.05		.988		.486					43.8		41.3	.592	589
.067	.406	24.3	19.4	8.85		.946		.503					74.8		35.0	.292	362
.065	.397	24.6	19.7	9.14		.931		.488					74.8		20.8	.101	218
.066	.396	23.9	24.1	8.96		.936		.488					12.9	40.7	43.0	1.028	584
.068	.416	24.6	23.7	8.79		.947		.497					25.8	39.2	41.9	.496	575
.065	.390	24.1	24.8	8.79		.930		.461					45.0		34.9	.239	328
.065	.407	25.1	24.5	9.11		.906		.486					77.3		24.3	.095	232
.066	.409	25.0	29.1	9.01		.953		.471					13.2	41.8	43.3	.855	541
.066	.407	24.8	29.1	9.00		.971		.472					26.3		41.1	.407	507
.066	.405	24.5	29.1	9.03		.971		.469					46.1		34.7	.199	367
.064	.394	24.5	30.0	9.30		.945		.457					79.1		23.8	.078	225
.067	.416	35.1	19.1	8.85		.936		.501					11.7	33.3	36.9	1.055	727
.066	.411	35.0	19.3	8.96		.994		.491					23.5	34.0	35.6	.504	391
.066	.416	35.2	19.3	8.99		.947		.496					41.2	33.4	34.7	.279	346
.066	.409	34.4	19.2	8.93		.937		.501					70.7		20.7	.099	196
.066	.401	33.9	24.2	8.99		.923		.490					12.3	32.7	36.0	.838	552
.067	.406	34.2	24.0	8.89		.945		.492					24.6	33.8	35.7	.410	521
.066	.403	34.4	24.4	9.05		.904		.484					42.9	32.9	34.8	.230	331
.067	.410	35.0	24.0	8.92		.959		.490					73.6		21.4	.082	201
.063	.378	34.1	30.6	9.49		.953		.455					12.6	30.3	36.2	.696	516
.062	.373	34.1	30.9	9.57		.947		.454					25.2	34.4	36.3	.354	394
.063	.378	33.7	30.5	9.46		.949		.460					44.1	33.9	34.5	.193	325
.064	.393	34.6	30.0	9.30		.973		.468					75.7		21.1	.068	204
.067	.410	49.3	19.1	8.87		.932		.497					10.8	19.1	32.0	.933	1148
.067	.409	49.5	19.2	8.91		1.010		.493					21.8	19.9	29.2	.424	591
.066	.400	48.9	19.3	8.97		.946		.497					37.3	20.1	29.0	.242	
.066	.407	49.7	19.6	9.04		.940		.490					65.5	21.2	22.6	.109	211
.067	.408	49.2	24.0	8.91		.942		.491					11.4	19.5	31.4	.726	713
.067	.410	49.2	23.9	8.87		.938		.490					23.0	19.8	31.5	.359	627
.066	.403	49.7	24.4	9.05		.925		.487					40.1	20.2	29.2	.190	325
.067	.421	50.6	23.9	8.86		.970		.496					68.7	20.0	20.5	.077	192
.062	.393	51.0	30.7	9.52		.934		.457					11.9	16.7	30.6	.567	565
.061	.379	50.5	31.6	9.79		.905		.440					23.8	19.1	30.4	.286	426
.064	.405	50.7	29.9	9.25		.976		.469					41.6	20.3	29.5	.158	320
.065	.409	50.5	29.4	9.11		.985		.475					71.3	19.8	22.5	.070	214

\* Lengths are for the actual test specimens for which  $\sigma = 3.75$  approximately.NATIONAL ADVISORY  
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TABLE 2.- Concluded

TEST DATA AND PROPORTIONS OF SPECIMENS HAVING  $\frac{t_w}{t_s} = 0.40$  - Concluded

(b) ALCLAD 75S-T SHEET AND 75S-T STIFFENERS

Proportions of test specimens														Test data			
$t_w$	$\frac{t_w}{t_s}$	$\frac{b_s}{t_s}$	$\frac{b_w}{t_w}$	$\frac{b_A}{t_w}$	$\frac{b_w}{b_Y}$	$\frac{t_w}{t_L}$	$\frac{b_w}{b_L}$	$\frac{t_w}{t_F}$	$\frac{b_w}{b_F}$	$\frac{r}{t_w}$	$\frac{d}{t_s}$	$\frac{p}{t_s}$	$\frac{L}{b_w}$ (in.)	$\sigma_{cr}$ (ksi)	$\sigma_f$ (ksi)	$\frac{P_f}{L/\sqrt{a}}$ $\frac{\text{ksi}\sqrt{\text{in.}}}{\text{in.}}$	$\bar{e}_f$
(0.064)	(0.40)	(25)	(20)	(9.3)	(0.99)	(0.94)	(1.07)	(0.47)	(1.14)	(1)	(1.54)	(4.6)	12.4	49.8	57.6	1.630	$640 \times 10^5$
0.060	0.380	25.6	21.4	9.96		0.884		0.442					24.9	51.5	58.1	.834	602
.067	.422	25.3	19.2	8.92		.967		.423					43.7	---	55.0	.443	526
.064	.411	25.8	20.0	9.27		.955		.481					75.0	---	55.0	.443	261
.060	.370	24.7	21.3	9.90		.832		.431							52.9	.125	
.064	0.402	25.2	25.1	9.31		.988		.464					12.8	51.9	57.9	1.351	589
.063	.395	25.2	25.3	9.40		.966		.470					25.6	55.0	58.2	.675	565
.066	.418	25.4	24.1	8.96		.938		.486					45.0	---	52.9	.352	535
.062	0.404	(35)	(20)	9.58		.887		.440					11.7	30.7	52.3	1.468	535
.062	.389	37.1	20.6	9.63		.870		.439					23.5	30.5	50.3	.720	667
.064	.402	35.3	20.0	9.30		.955		.478					41.2	33.9	45.8	.369	582
.064	.402	35.4	20.1	9.34		.951		.479					70.8	---	24.7	.115	243
.064	.398	35.2	25.1	9.31		.957		.461					12.2	28.3	48.4	1.102	615
.066	.411	35.2	24.3	9.03		.929		.488					24.5	33.4	47.7	.543	640
.064	.404	35.3	25.0	9.28		.896		.464					42.8	35.7	49.1	.319	620
.066	.416	35.8	24.4	9.04		.937		.468					73.6	---	27.1	.102	265
.066	.422	35.9	28.9	8.95		.921		.477					12.6	30.8	47.4	.906	567
.063	.394	35.4	30.5	9.46		.937		.453					25.2	33.0	47.1	.449	520
.067	.429	36.0	28.8	8.91		.929		.471					44.2	34.5			
.066	.422	35.7	28.9	8.96		.931		.477					75.8				
.065	.418	(50)	(20)	9.17		.940		.489					10.1	17.7	43.9	1.320	690
.064	.410	52.2	19.7	9.25		.925		.485					21.8	16.6	42.3	.594	740
.066	.421	51.6	19.9	9.03		.939		.491					38.1	17.1	40.2	.324	569
.061	.393	51.9	19.4	9.74		.860		.441					65.4	17.5	23.8	.111	258
.065	.411	51.1	24.7	9.17		.967		.481					11.4	15.8	42.0	.944	539
.066	.417	51.4	24.4	9.07		.919		.482					22.8	16.6	43.6	.491	590
.065	.416	51.3	24.5	9.08		.949		.469					40.0	17.2	41.6	.266	585
.065	.415	51.3	24.5	9.08		.941		.485					68.7	17.9	22.9	.085	264

<sup>a</sup>Lengths are for the actual test specimens for which  $c = 3.75$  approximately.NATIONAL ADVISORY  
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TABLE 3  
TEST DATA AND PROPORTIONS OF SPECIMENS HAVING  $\frac{t_w}{t_s} = 0.63$   
[Nominal proportions are given in parentheses]

## (a) 24S-T SHEET AND STIFFENERS

Proportions of test specimens														Test data			
$t_w$ (in.)	$\frac{t_w}{t_s}$	$\frac{b_s}{t_s}$	$\frac{b_w}{t_w}$	$\frac{b_s}{t_s}$	$\frac{b_w}{t_w}$	$\frac{t_w}{L}$	$\frac{b_w}{L}$	$\frac{t_w}{P}$	$\frac{b_w}{P}$	$\frac{r}{t_s}$	$\frac{d}{t_s}$	$\frac{p}{t_s}$	$\frac{L}{b_w}$ (a)	$\sigma_{cr}$ (ksi)	$\delta_f$ (ksi)	$\frac{F_t}{L/\sqrt{a}}$ (ksi/in.)	$\bar{\epsilon}_f$
(0.064)	(0.63)	(25)	(20)	(9.3)	(0.96)	(0.94)	(1.07)	(0.47)	(1.44)	(1)	(1.84)	(6.1)	14.0	42.6	45.1	1.013	715 x 10 <sup>5</sup>
0.067	0.687	25.2	19.0	8.83		.948		.498					20.0	41.3	42.6	.484	610
0.067	.662	24.5	19.1	8.88		.946		.499					14.8		33.8	.216	515
0.067	.690	25.6	19.1	8.85		.949		.499					85.9		21.2	.079	213
0.066	.687	26.0	19.2	8.95		.931		.494									
0.067	.661	24.7	23.8	8.83		.927		.500					14.2	39.2	41.5	.788	540
0.066	.625	23.1	24.3	9.01		.950		.469					28.4		40.6	.398	513
0.066	.647	24.6	24.2	8.99		.961		.487					49.8		34.5	.187	329
0.066	.674	25.3	24.1	8.95		.935		.482					85.8		23.9	.074	226
0.066	.665	25.4	29.1	9.05		.972		.473					14.3		41.0	.663	472
0.065	.661	25.1	29.3	9.08		.978		.468					28.7	40.0	40.7	.331	481
0.065	.697	25.4	29.6	9.18		.931		.473					50.5		35.5	.165	360
0.066	.665	25.2	29.0	8.97		.959		.476					86.1		22.5	.062	220
0.067	.685	35.6	19.0	8.84		.952		.501					11.1	35.6	40.6	1.018	652
0.067	.654	34.5	19.1	8.87		.937		.501					27.1	35.9	38.4	.406	571
0.065	.664	36.2	19.6	9.08		.933		.484					47.7		33.1	.194	333
0.067	.681	35.9	19.1	8.89		.936		.499					81.8		22.4	.077	220
0.067	.635	33.0	23.8	8.83		.935		.494					13.9	33.3	37.4	.670	518
0.066	.638	34.1	24.2	8.97		.944		.487					27.9	36.0	37.7	.332	483
0.067	.654	34.2	24.0	8.92		.951		.492					48.8		34.7	.173	319
0.064	.647	35.6	25.1	9.35		.887		.465					85.3		25.1	.072	237
0.064	0.627	34.3	29.7	9.20		.982		.471					14.1	33.7	36.8	.551	530
0.065	.607	33.1	29.5	9.14		.953		.473					28.3	35.4	37.0	.284	458
0.066	.640	34.3	29.0	8.97		.941		.480					49.5		33.9	.148	345
0.064	.649	36.0	29.8	9.24		.926		.461					84.5		22.6	.056	225
0.061	.521	48.2	20.9	9.69		.896		.465					12.8	21.5	34.6	.688	794
0.067	.621	47.7	19.1	8.89		1.010		.496					26.3	22.2	32.5	.319	568
0.068	.632	47.6	18.9	8.79		.956		.504					45.2	26.0	30.2	.171	329
0.067	.637	48.0	19.1	8.87		.934		.505					78.8		22.5	.074	216
0.068	.657	49.3	23.7	8.80		.930		.498					13.5	20.3	34.8	.563	644
0.066	.649	49.4	24.1	8.93		.934		.490					27.0	23.1	34.1	.270	465
0.066	.644	49.2	24.2	8.97		.944		.486					47.3	23.1	30.8	.139	323
0.066	.650	49.6	24.2	8.97		.962		.487					81.0		22.0	.059	211
0.061	.574	47.3	31.3	9.67		.919		.449					13.8	20.2	33.6	.466	505
0.062	.583	47.6	30.9	9.57		.944		.453					27.5	23.4	32.9	.230	517
0.064	.649	51.1	29.8	9.24		.977		.471					48.1	20.3	31.1	.119	350
0.061	.618	51.8	31.5	9.75		.917		.443					82.7	21.4	22.0	.049	208
0.062	.624	76.3	20.6	9.57		.899		.471					11.9	10.0	30.0	.559	762
0.066	.673	77.4	19.4	9.00		1.010		.493					24.6	9.5	28.7	.256	499
0.066	.665	76.4	19.2	9.2		.938		.502					43.1	10.9	26.5	.137	343
0.066	.664	76.5	19.4	9.01		.933		.498					74.0	10.1	20.2	.062	225
0.068	.666	74.9	23.6	8.75		.933		.497					12.8	9.2	30.1	.452	616
0.067	.662	75.0	23.8	8.85		.937		.495					25.6	8.3	30.6	.229	496
0.067	.668	75.5	23.9	8.87		.957		.495					44.9	10.8	27.2	.116	347
0.067	.667	76.0	24.0	8.91		.941		.487					76.6	10.4	19.2	.048	215
0.067	.654	74.4	28.7	8.89		.956		.480					13.1	9.5	30.6	.393	514
0.065	.643	75.4	29.6	9.18		.967		.474					26.3	9.9	30.3	.190	500
0.066	.648	77.4	29.2	9.05		.979		.478					46.0	9.3	27.9	.100	370
0.065	.643	77.7	30.4	9.45		.950		.461					79.1	10.4	19.8	.041	208

\*Lengths are for the actual test specimens for which  $c = 3.75$  approximately.

TABLE 3.- Concluded  
 TEST DATA AND PROPORTIONS OF SPECIMENS HAVING  $\frac{t_w}{t_s} = 0.63$  - Concluded

(b) ALCLAD 75S-T SHEET AND 75S-T STIFFENERS

Proportions of test specimens														Test data			
$t_w$	$\frac{t_w}{t_s}$	$\frac{b_s}{t_s}$	$\frac{b_w}{t_w}$	$\frac{b_A}{t_w}$	$\frac{b_w}{b_Y}$	$\frac{t_w}{t_L}$	$\frac{b_w}{b_L}$	$\frac{t_w}{t_F}$	$\frac{b_w}{b_F}$	$\frac{x}{t_w}$	$\frac{d}{t_s}$	$\frac{p}{t_s}$	$\frac{L}{b_w}$ (in.)	$\sigma_r$ (ksi)	$\bar{\sigma}_r$ (ksi)	$\frac{P_1}{L/\sqrt{R}}$ (ksi/in.)	$\epsilon_r$
(0.064) .066 .067 .066 .062	(0.63) 0.633 0.643 0.641 0.605	(25) 23.9 23.8 24.4 24.3	(20) 19.2 19.1 19.4 20.5	(9.3) 8.93 8.85 9.03 9.52	(0.86) 0.965 0.937 0.948 0.881	(0.94) 0.941 0.937 0.948 0.881	(1.07) 1.07 1.07 1.07 1.07	(0.67) 0.67 0.67 0.67 0.67	(1.14) 1.14 1.14 1.14 1.14	(1.14) 1.14 1.14 1.14 1.14	(1.84) 1.84 1.84 1.84 1.84	(6.1) 6.1 6.1 6.1 6.1	14.0 27.9 48.9 48.9 84.0	58.2 63.2 57.9 57.9 54.4	63.3 60.5 57.9 57.9 54.4	1.458 .726 .376 .376 .097	619 x 10 <sup>-5</sup> 620 562 562 256
.064 .067 .066 .064	.621 .657 .657 .615	24.5 25.2 25.2 24.3	25.1 24.1 24.1 25.2	9.32 8.96 8.96 9.35	.981 .935 .935 .971	.475 .493 .493 .461							14.2 28.4 49.8	57.9 56.3 54.6	57.9 56.3 54.6	1.104 .553 .294	563 573 524
.062 .064 .068 .064	.596 .616 .656 .614	33.8 33.9 34.2 33.7	20.5 20.0 18.9 19.9	9.53 9.26 8.79 9.22	.866 .917 .960 .928	.456 .481 .486 .488							13.6 27.3 47.6 81.8	35.9 43.7 37.5 23.6	54.7 57.2 52.2 23.6	1.197 .607 .315 .084	669 646 555 255
.064 .065 .065	.623 .646 .640	34.6 35.0 34.9	24.8 24.7 24.5	9.21 9.15 9.10	.976 .997 .934	.479 .471 .497							13.9 27.8 48.8	38.1 36.3 38.9	55.1 53.5 49.3	.953 .462 .244	590 602 519
.065 .066 .064	.662 .654 .652	35.7 34.6 34.6	29.3 29.0 29.8	9.08 8.97 9.24	.906 .904 .903	.472 .473 .465							14.1 28.2 49.3	32.3 35.1 35.2	48.6 49.9 47.1	.736 .378 .204	533 548 489
.065 .063 .063 .061	.616 .593 .603 .583	48.0 47.5 48.8 48.2	19.7 20.3 20.5 21.0	9.24 9.43 9.50 9.74	.938 .867 .866 .836	.444 .446 .444 .437							13.0 26.2 45.9 78.7	20.4 20.9 21.9 22.7	51.0 50.9 46.2 23.7	1.019 .509 .260 .078	648 689 587 237
.065 .067 .066 .064	.623 .632 .628 .610	48.8 47.7 48.3 48.1	24.7 24.0 24.2 25.0	9.18 8.91 8.98 9.28	.939 .933 .936 .938	.481 .478 .476 .464							13.5 27.0 47.3 81.0	19.9 20.8 21.6 21.9	50.0 51.2 46.6 23.7	.802 .423 .219 .065	652 617 572 234
.066 .067 .066 .063	.638 .666 .654 .624	48.6 50.4 49.9 49.9	29.0 28.8 28.9 30.3	8.99 8.91 8.96 9.40	.938 .937 .925 .898	.476 .490 .479 .463							13.8 27.5 48.2 82.6	19.5 19.5 19.7 21.9	45.6 46.6 44.3 23.6	.641 .325 .176 .055	566 561 531 256
.061 .066 .065 .063	.579 .629 .619 .602	72.4 72.9 72.7 72.6	21.0 19.4 19.7 20.3	9.77 9.01 9.17 9.41	.909 .956 .938 .868	.435 .434 .433 .450							12.2 24.6 43.1 74.0	10.0 8.7 9.9 9.9	43.5 43.6 40.2 22.2	.824 .413 .217 .070	739 686 576 279
.066 .065 .065 .066	.635 .650 .641 .642	74.0 76.2 75.7 73.7	24.4 24.6 24.8 24.1	9.04 9.12 9.20 8.95	.922 .904 .946 .924	.487 .478 .478 .485							12.8 25.8 44.8 76.8	7.7 7.9 9.5 8.4	43.8 42.2 40.2 22.6	.665 .313 .172 .058	634 593 582 276
.067 .064 .066 .066	.663 .637 .656 .659	75.4 75.8 75.6 75.7	28.7 30.0 29.1 29.0	8.89 9.30 9.01 8.97	.929 .921 .906 .927	.480 .459 .474 .476							13.2 26.4 46.0 79.0	7.4 8.9 8.7 10.6	39.7 39.6 39.5 22.1	.509 .252 .144 .047	560 532 559 280

<sup>a</sup>Lengths are for the actual test specimens for which  $c = 3.75$  approximately.

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TABLE 4  
TEST DATA AND PROPORTIONS OF SPECIMENS HAVING  $\frac{t_w}{t_s} = 1.00$

[Nominal proportions are given in parentheses]

(a) 24S-T SHEET AND STIFFENERS

Proportions of test specimens													Test data				
$t_w$ (in.)	$\frac{t_w}{t_s}$	$\frac{b_s}{t_s}$	$\frac{b_w}{t_w}$	$\frac{b_A}{t_w}$	$\frac{b_w}{b_y}$	$\frac{t_w}{t_L}$	$\frac{b_w}{b_L}$	$\frac{t_w}{t_F}$	$\frac{b_w}{b_F}$	$\frac{r}{t_w}$	$\frac{d}{t_s}$	$\frac{p}{t_s}$	$\frac{L}{b_w}$ (in.)	$\sigma_{cr}$ (ksi)	$\bar{\sigma}_r$ (ksi)	$\frac{P_1}{L/\sqrt{a}}$ (ksi/in.)	$\bar{\epsilon}_r$
(0.064)	(1.00)	(25)	(20)	(9.3)	(0.96)	(0.94)	(1.07)	(0.47)	(1.44)	(1)	(2.44)	(7.8)					
0.067	1.035	23.4	19.0	8.83		0.939	.500						13.9		42.4	.966	567x10 <sup>-5</sup>
.063	.981	23.4	20.2	9.37		.916	.482						29.5		41.2	.473	497
.068	1.080	24.8	18.9	9.49		.944	.504						50.9		33.9	.206	342
.069	1.099	25.1	18.6	8.63		1.006	.508						67.6		21.2	.073	209
		(25)															
.068	1.045	22.9	23.7	8.79		.942	.496						14.7	35.6	39.1	.713	590
.067	1.028	23.0	23.9	8.87		.938	.494						29.3		38.8	.352	490
.066	1.059	22.9	24.1	8.95		.935	.490						51.3		35.4	.183	336
.067	1.065	23.4	24.0	8.91		.930	.499						67.9		23.4	.070	227
		(20)															
.067	1.039	32.3	19.1	8.89		.949	.499						13.5	38.5	41.1	.843	505
.068	1.027	32.3	18.9	8.77		.942	.506						28.9		39.9	.384	455
.063	.946	34.9	20.2	9.37		.903	.474						50.7		35.1	.189	355
.067	1.075	32.9	19.0	8.84		1.006	.497						67.1		21.4	.065	245
		(25)															
.067	1.028	33.1	24.0	8.89		.941	.492						14.6	37.9	39.1	.639	612
.067	1.047	32.6	23.9	8.88		.953	.494						29.2		38.5	.312	447
.067	1.059	33.5	24.0	8.89		.940	.488						51.1		34.7	.158	349
.066	1.064	33.7	24.1	8.95		.925	.496						67.5		24.3	.064	235
		(30)															
.064	0.971	35.2	29.9	9.25		.958	.467						14.7	33.3	37.6	.536	440
.066	1.003	35.1	29.2	9.05		.988	.477						29.3	32.8	36.1	.257	385
.063	.972	35.6	30.3	9.37		.935	.461						51.3	33.2	33.9	.138	357
.064	1.021	35.6	30.0	9.28		.930	.459						68.0		22.7	.053	218
		(50)															
.067	1.052	46.8	19.1	8.85		.940	.501						14.4	25.7	38.7	.646	613
.063	.960	49.8	20.3	9.44		.897	.473						28.6	28.1	37.1	.308	521
.065	1.030	48.7	19.8	9.18		.934	.480						50.1	27.6	32.2	.151	359
.067	1.032	47.0	19.1	8.87		.938	.499						66.0		21.3	.059	215
		(25)															
.067	1.032	47.0	24.0	8.89		.933	.487						14.5	25.5	37.0	.532	630
.063	1.023	46.7	23.7	8.79		.931	.498						28.9	27.8	36.3	.261	649
.065	1.020	48.6	24.3	9.01		.939	.483						50.7	27.2	32.3	.130	312
.066	1.025	47.9	24.4	9.05		.945	.486						66.9		22.3	.052	227
		(30)															
.068	1.061	46.3	28.2	8.74		.952	.488						14.5	23.2	35.3	.448	363
.067	1.039	46.6	28.6	8.85		.974	.484						29.2	24.4	34.7	.221	450
.065	1.003	48.2	29.3	9.08		.960	.475						50.7	27.8	31.9	.115	325
.063	1.002	50.0	30.4	9.43		.957	.463						67.6		21.9	.044	205
		(75)															
.068	1.023	69.6	18.9	8.77		.938	.505						13.8	10.0	34.2	.513	625
.067	1.026	71.1	19.1	8.85		.941	.503						26.5	12.9	34.5	.268	505
.064	1.002	73.9	20.0	9.28		.914	.478						48.9	12.5	29.2	.121	543
.066	1.023	72.0	19.2	8.93		.927	.503						64.0	13.5	19.5	.048	207
		(25)															
.068	1.010	70.5	23.7	8.80		.932	.494						14.3	11.5	34.2	.430	520
.067	1.020	70.7	23.9	8.88		.984	.492						28.4	11.9	33.0	.206	490
.067	1.040	72.0	23.9	8.88		.956	.490						49.8	12.2	29.6	.104	337
.067	1.042	71.6	24.0	8.89		.990	.488						65.5	12.8	20.0	.041	210
		(30)															
.063	.972	74.8	30.2	9.40		.921	.462						14.4	10.5	31.7	.340	496
.064	.980	73.8	29.7	9.21		.979	.470						26.6	13.1	32.8	.178	381
.066	1.002	72.1	29.2	9.04		.963	.477						50.2	12.6	29.5	.092	323
.062	.986	76.6	30.8	9.53		.947	.454						66.5	11.8	20.5	.036	209

\*Lengths are for the actual test specimens for which  $c = 3.75$  approximately.

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TABLE 4.- Concluded  
 TEST DATA AND PROPORTIONS OF SPECIMENS HAVING  $\frac{t_w}{t_s} = 1.00$  - Concluded

(b) ALCLAD 75S-T SHEET AND 75S-T STIFFENERS

Proportions of test specimens														Test data			
$t_w$	$\frac{t_h}{t_s}$	$\frac{b_s}{t_s}$	$\frac{b_w}{t_w}$	$\frac{b_A}{t_w}$	$\frac{b_w}{b_y}$	$\frac{t_w}{t_L}$	$\frac{b_w}{b_L}$	$\frac{t_w}{t_F}$	$\frac{b_w}{b_F}$	$\frac{r}{t_w}$	$\frac{d}{t_s}$	$\frac{p}{t_s}$	$\frac{L}{b_w}$ (in.)	$\sigma_{or}$ (ksi)	$\bar{\sigma}_f$ (ksi)	$\frac{F_1}{L/\sqrt{a}}$ (ksi/in.)	$\bar{E}_f$
(0.064)	(1.00)	(25)	(20)	(9.3)	(0.96)	(0.94)	(1.07)	(0.47)	(1.44)	(1)	(2.44)	(7.8)					
.064	1.003	24.5	20.0	9.31		0.876		0.452					14.6	60.8	64.0	1.350	615 × 10 <sup>5</sup>
.065	1.002	24.8	20.3	9.43		.933		.475					29.2	61.1	61.6	.658	593
.064	1.019	24.9	19.9	9.24		.927		.484					50.9	58.0	58.0	.345	551
.065	1.027	24.2	19.7	9.14		.939		.491					87.6	56.2	56.2	.087	237
.065	.949	24.0	24.6	9.14		.902		.479					14.7	48.9	57.4	1.048	583
.064	.973	23.6	24.9	9.24		.921		.471					29.4	42.4	58.7	.533	600
.065	.991	24.2	24.7	9.18		.907		.477					51.3	52.9	52.9	.269	524
.062	.955	34.3	20.8	9.66		.833		.467					14.5	47.7	58.3	1.095	650
.064	.937	34.5	20.0	9.30		.896		.455					23.0	48.7	56.6	.534	547
.065	1.006	34.2	19.8	9.18		.945		.481					50.7	45.6	52.3	.278	497
.064	.998	34.1	20.1	9.35		.946		.461					87.1	49.4	49.4	.061	249
.064	1.013	34.8	25.0	9.27		.912		.478					14.6	42.4	54.8	.877	592
.065	.989	33.7	24.8	9.21		.936		.477					29.2	44.7	53.9	.429	542
.066	1.027	35.1	24.4	9.07		.928		.483					51.1	47.4	49.4	.223	522
.065	1.008	33.6	29.3	9.08		.901		.471					14.7	33.3	48.3	.690	560
.065	1.011	34.4	29.4	9.10		.903		.468					29.3	27.5	48.3	.343	520
.065	1.019	34.5	29.6	9.15		.896		.468					51.3	26.2	44.7	.179	457
.061	.944	48.8	21.0	9.74		.852		.429					14.3	23.3	55.6	.925	704
.062	.967	49.1	20.6	9.58		.858		.452					28.6	25.3	57.0	.476	541
.062	.963	48.9	20.5	9.52		.927		.466					50.2	23.7	50.1	.235	524
.062	.946	48.5	20.7	9.61		.858		.452					86.0	21.6	24.8	.069	255
.065	.951	45.9	24.5	9.08		.910		.483					14.5	25.7	53.1	.769	666
.065	.986	48.3	24.7	9.18		.956		.485					29.2	24.3	50.6	.356	551
.063	.952	47.4	25.2	9.37		.922		.470					50.7	24.8	48.5	.197	511
.064	.949	47.5	25.2	9.35		.968		.461					87.0	23.3	24.8	.058	250
.065	1.014	50.2	29.7	9.21		.896		.482					14.6	23.8	47.1	.586	517
.064	1.022	50.3	29.9	9.25		.908		.462					29.1	21.9	45.9	.286	503
.065	1.002	49.1	29.7	9.21		.901		.463					50.6	21.9	44.5	.161	465
.063	.989	49.8	30.5	9.46		.904		.454					87.4	23.3	24.9	.051	260
.062	.960	74.2	20.7	9.63		.848		.437					14.0	9.2	49.3	.717	638
.061	.953	74.6	21.1	9.82		.833		.427					28.0	11.9	49.6	.360	593
.067	1.059	75.2	19.1	8.88		.964		.499					48.9	11.1	46.9	.194	581
.062	.967	74.6	20.8	9.64		.866		.436					84.1	13.4	22.9	.055	236
.064	.933	70.4	25.1	9.32		.974		.471					14.3	11.7	47.7	.590	839
.064	1.002	74.5	25.0	9.28		.910		.472					28.5	10.4	47.2	.289	558
.064	.961	71.6	25.1	9.32		.967		.460					49.8	11.7	43.8	.155	543
.064	.973	71.7	24.9	9.25		.946		.461					84.2	12.0	23.2	.049	260
.067	1.026	73.3	28.8	8.93		.921		.479					14.4	11.1	43.5	.477	545
.064	1.005	73.9	29.8	9.22		.887		.462					28.8	10.7	42.6	.232	539
.066	1.022	74.3	29.2	9.04		.907		.472					50.4	11.8	43.4	.133	489
.060	.926	73.0	31.8	9.85		.878		.442					86.5	11.6	22.9	.041	258

\*Lengths are for the actual test specimens for which  $c = 3.75$  approximately.

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TABLE 5.- COMPARATIVE DESIGNS OF Y- AND Z-STIFFENED PANELS ( $P_c = 2.0$  kips/inch;  $e = 1$ ; minimum-weight designs in parentheses)

[illegible]

TABLE 5.-COMPARATIVE DESIGNS OF Y- AND Z-CORNERED PANELS - Continued

TABLE 5.-COMPARATIVE DESIGN OF X- AND Z-CORRODED PANELS - Continued																					
$\frac{t_y}{t_z}$		Panel	$L = 30$ in.										$L = 40$ in.								
			$t_z$ in.										$t_z$ in.								
$\sigma_r$ (ksi)	$\sigma_z$	$\sigma_x$	0.025	0.050	0.075	0.100	0.125	0.150	0.175	0.200	0.225	0.250	0.275	0.300	0.325	0.350	0.375	0.400	0.425	0.450	0.475
			0.025	0.050	0.075	0.100	0.125	0.150	0.175	0.200	0.225	0.250	0.275	0.300	0.325	0.350	0.375	0.400	0.425	0.450	0.475
.40																					
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TABLE 6. -- COMPARATIVE DESIGNS OF Y- AND Z-STIFFENED PANELS [ $P_c = 5.0$  kips/inch;  $\sigma = 1$ ; minimum-weight designs in parentheses]

	$t_w$ (in.)	Panel	$b = 10$ in.										$b = 20$ in.									
			$t_w$ , in.										$t_w$ , in.									
$\sigma_r$ (ksi)	0.40	Y 753-T	0.020	0.025	0.032	0.040	0.051	0.064	0.081	0.102	0.125	0.027	0.032	0.040	0.051	0.064	0.081	0.102	0.125	0.154		
		Z 213-T	(.49.5)	(.49.5)	(.49.5)	(.49.5)	(.49.5)	(.49.5)	(.49.5)	(.49.5)	(.49.5)	(.49.5)	(.49.5)	(.49.5)	(.49.5)	(.49.5)	(.49.5)	(.49.5)	(.49.5)	(.49.5)		
	.51	Y 753-T																				
		Z 213-T																				
	.65	Y 753-T																				
		Z 213-T																				
	.79	Y 753-T																				
		Z 213-T																				
	1.00	Y 753-T																				
		Z 213-T																				
$\sigma_{cr}$ (ksi)	.40	Y 753-T																				
		Z 213-T																				
	.51	Y 753-T																				
		Z 213-T																				
	.65	Y 753-T																				
		Z 213-T																				
	.79	Y 753-T																				
		Z 213-T																				
	1.00	Y 753-T																				
		Z 213-T																				
$K$ (in.)	.40	Y 753-T																				
		Z 213-T																				
	.51	Y 753-T																				
		Z 213-T																				
	.65	Y 753-T																				
		Z 213-T																				
	.79	Y 753-T																				
		Z 213-T																				
	1.00	Y 753-T																				
		Z 213-T																				
$S$ (in.)	.40	Y 753-T																				
		Z 213-T																				
	.51	Y 753-T																				
		Z 213-T																				
	.65	Y 753-T																				
		Z 213-T																				
	.79	Y 753-T																				
		Z 213-T																				
	1.00	Y 753-T																				
		Z 213-T																				
$K$ (in.)	.40	Y 753-T																				
		Z 213-T																				
	.51	Y 753-T																				
		Z 213-T																				
	.65	Y 753-T																				
		Z 213-T																				
	.79	Y 753-T																				
		Z 213-T																				
	1.00	Y 753-T																				
		Z 213-T																				
$\rho$ (in.)	.40	Y 753-T																				
		Z 213-T																				
	.51	Y 753-T																				
		Z 213-T																				
	.65	Y 753-T																				
		Z 213-T																				
	.79	Y 753-T																				
		Z 213-T																				
	1.00	Y 753-T																				
		Z 213-T																				
$\rho_w$	.40	Y 753-T																				
		Z 213-T																				
	.51	Y 753-T																				
		Z 213-T																				
	.65	Y 753-T																				
		Z 213-T																				
	.79	Y 753-T																				
		Z 213-T																				
	1.00	Y 753-T																				
		Z 213-T																				
$\sigma_{cr}$	.40	Y 753-T																				
		Z 213-T																				
	.51	Y 753-T																				
		Z 213-T																				
	.65	Y 753-T																				
		Z 213-T																				
	.79	Y 753-T																				
		Z 213-T																				
	1.00	Y 753-T																				
		Z 213-T																				

TABLE 6.- COMPARATIVE DESIGNS OF Y- AND Z-STIFFENED PANELS - Concluded

t/y	Panel	L = 30 in. Fig. 1a.												L = 40 in. Fig. 1b.											
		0.025	0.032	0.040	0.050	0.063	0.080	0.100	0.125	0.156	0.188	0.032	0.040	0.050	0.063	0.080	0.100	0.125	0.156	0.188					
0.40	755-H																								
0.51	755-H																								
0.63	755-H																								
0.79	755-H																								
1.00	755-H																								
0.40	755-H																								
0.51	755-H																								
0.63	755-H																								
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0.40	755-H																								
0.51	755-H																								
0.63	755-H																								
0.79	755-H																								
1.00	755-H																								
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0.51	755-H																								
0.63	755-H																								
0.79	755-H																								
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0.51	755-H																								
0.63	755-H																								
0.79	755-H																								
1.00	755-H																								
0.40	755-H																								
0.51	755-H																								
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0.79	755-H																								
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0.51	755-H																								
0.63	755-H																								
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TABLE 7.- COMPARATIVE DESIGNS OF Y- AND Z-STIFFERED PANELS [ $E_s = 3.0$  kips/inch;  $\nu = 1$ ; minimum-weight designs in parentheses]

Panel		$L = 10$ in.										$L = 20$ in.									
$t_y$ in.	$t_z$ in.	$t_x$ in.										$t_x$ in.									
		0.032	0.040	0.051	0.064	0.082	0.102	0.125	0.156	0.188	0.204	0.040	0.051	0.064	0.082	0.102	0.125	0.156	0.188	0.204	
$\sigma_r$ (ksi)	0.40				(34.1)	(46.6)	(58.9)	(71.2)	(83.5)	(95.8)	(108.1)				(48.0)	(63.3)	(78.5)	(93.8)	(109.0)	(124.2)	
	.51			(57.3)	51.5	44.5	37.7	30.9	25.2	20.0	15.7				(46.8)	(61.7)	(76.6)	(91.5)	(106.4)	(121.3)	
	.63			(60.1)	55.3	49.6	43.9	38.2	32.5	26.8	21.1			(49.8)	(64.1)	(78.3)	(92.5)	(106.7)	(120.9)	(135.1)	
	.79				57.3	52.4	46.9	41.2	35.5	29.8	24.1			(51.2)	(65.4)	(79.6)	(93.8)	(108.0)	(122.2)	(136.4)	
	1.00	(58.5)	54.4	49.2	43.7	38.0	32.3	26.6	20.9	15.2	9.5	(50.5)	47.5	44.5	41.5	38.5	35.5	32.5	29.5	26.5	
$\sigma_z$ (ksi)	.40				(50.1)	(67.5)	(84.9)	(102.3)	(119.7)	(137.1)	(154.5)				(42.0)	(56.0)	(70.0)	(84.0)	(98.0)	(112.0)	
	.51			(55.5)	51.0	45.2	39.4	33.6	27.8	22.0	16.2				(47.5)	(61.5)	(75.5)	(89.5)	(103.5)	(117.5)	
	.63			(58.6)	53.2	47.4	41.6	35.8	30.0	24.2	18.4			(49.1)	(63.1)	(77.1)	(91.1)	(105.1)	(119.1)	(133.1)	
	.79				54.5	50.0	45.5	41.0	36.5	32.0	27.5			(51.2)	(65.2)	(79.2)	(93.2)	(107.2)	(121.2)	(135.2)	
	1.00	(42.0)	21.0	10.5	16.7	10.8	6.5	2.2	1.7	1.2	0.7	(42.0)	21.0	10.5	16.7	10.8	6.5	2.2	1.7	1.2	
$\sigma_x$ (ksi)	.40				(1.08)	(1.35)	(1.72)	(2.10)	(2.48)	(2.86)	(3.24)				(1.36)	(1.69)	(2.02)	(2.35)	(2.68)	(3.01)	
	.51			(.84)	1.08	1.25	1.42	1.59	1.76	1.93	2.10				(1.46)	(1.77)	(2.08)	(2.39)	(2.70)	(3.01)	
	.63			(.88)	1.08	1.25	1.42	1.59	1.76	1.93	2.10			(1.43)	(1.74)	(2.05)	(2.36)	(2.67)	(2.98)	(3.29)	
	.79				1.08	1.25	1.42	1.59	1.76	1.93	2.10			(1.32)	(1.63)	(1.94)	(2.25)	(2.56)	(2.87)	(3.18)	
	1.00	(1.08)	1.35	1.72	2.10	2.48	2.86	3.24	3.62	4.00	4.38	(1.49)	1.72	1.95	2.18	2.41	2.64	2.87	3.10	3.33	
$\sigma_y$ (ksi)	.40				(1.15)	(1.35)	(1.55)	(1.75)	(1.95)	(2.15)	(2.35)				(1.23)	(1.43)	(1.63)	(1.83)	(2.03)	(2.23)	
	.51			(.93)	1.58	1.75	1.92	2.09	2.26	2.43	2.60				(1.59)	(1.79)	(1.99)	(2.19)	(2.39)	(2.59)	
	.63			(.76)	1.13	1.21	1.29	1.37	1.45	1.53	1.61			(1.33)	(1.53)	(1.73)	(1.93)	(2.13)	(2.33)	(2.53)	
	.79				1.07	1.84	1.75	1.66	1.57	1.48	1.39			(1.03)	(1.72)	(1.41)	(1.10)	(.79)	(.48)	(.17)	
	1.00	(.96)	1.54	2.60	2.78	2.96	3.14	3.32	3.50	3.68	3.86	(1.49)	2.58	2.76	2.94	3.12	3.30	3.48	3.66	3.84	
$\sigma_z$ (ksi)	.40				(.209)	.222	.238	.254	.270	.286	.302				(.346)	.378	.410	.442	.474	.506	
	.51					.261	.280	.299	.318	.337	.356			(.391)	.420	.449	.478	.507	.536	.565	
	.63			(.249)	.261	.280	.299	.318	.337	.356	.375			(.412)	.440	.469	.498	.527	.556	.585	
	.79				.304	.337	.370	.403	.436	.469	.502			(.512)	.540	.569	.598	.627	.656	.685	
	1.00	(.347)	.394	.441	.488	.535	.582	.629	.676	.723	.770	(.512)	.540	.569	.598	.627	.656	.685	.714	.743	
$\sigma_x$ (ksi)	.40				(.305)	.360	.420	.480	.540	.600	.660				(.523)	.600	.677	.754	.831	.908	
	.51					.451	.524	.597	.670	.743	.816			(.501)	.570	.643	.716	.789	.862	.935	
	.63			(.523)	.381	.451	.524	.597	.670	.743	.816			(.501)	.570	.643	.716	.789	.862	.935	
	.79				.396	.486	.579	.672	.765	.858	.951			(.501)	.570	.643	.716	.789	.862	.935	
	1.00	(.413)	.506	.599	.692	.785	.878	.971	.1064	.1157	.1250	(.501)	.570	.643	.716	.789	.862	.935	.1028	.1121	
$\sigma_y$ (ksi)	.40				(18.0)	(18.0)	(18.0)	(18.0)	(18.0)	(18.0)	(18.0)				(25.1)	(25.1)	(25.1)	(25.1)	(25.1)	(25.1)	
	.51				(18.0)	(18.0)	(18.0)	(18.0)	(18.0)	(18.0)	(18.0)				(24.5)	(24.5)	(24.5)	(24.5)	(24.5)	(24.5)	
	.63			(18.7)	18.0	18.0	18.0	18.0	18.0	18.0	18.0			(24.0)	21.9	18.0	18.0	18.0	18.0	18.0	
	.79				18.0	18.0	18.0	18.0	18.0	18.0	18.0			(22.1)	20.5	18.0	18.0	18.0	18.0	18.0	
	1.00	(18.0)	18.0	18.0	18.0	18.0	18.0	18.0	18.0	18.0	18.0	(20.0)	18.0	18.0	18.0	18.0	18.0	18.0	18.0	18.0	
$\sigma_z$ (ksi)	.40				(24.9)	38.0	54.4	70.8	87.2	103.6	120.0				(34.6)	44.0	53.4	62.8	72.2	81.6	
	.51					31.6	44.0	56.4	68.8	81.2	93.6			(33.4)	40.0	47.6	55.2	62.8	70.4	78.0	
	.63			(23.0)	36.8	44.0	51.2	58.4	65.6	72.8	80.0			(33.4)	40.0	47.6	55.2	62.8	70.4	78.0	
	.79				34.8	53.6	72.4	91.2	110.0	128.8	147.6			(29.4)	47.0	55.6	64.2	72.8	81.4	90.0	
	1.00	(36.7)	53.4	78.5	103.6	128.8	153.9	179.0	204.1	229.2	254.3	(49.0)	70.7	91.6	112.5	133.4	154.3	175.2	196.1	217.0	

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TABLE 7.-COMPARATIVE DESIGNS OF X- AND K-STIFFENED PANELS - Continued  
L = 40 in.

Panel	$\frac{t_w}{t_s}$	$t_s$ , in.										$t_s$ , in.									
		0.040	0.050	0.060	0.080	0.100	0.125	0.150	0.180	0.200		0.040	0.050	0.060	0.080	0.100	0.125	0.150	0.180	0.200	0.250
$\frac{t_w}{t_s}$	0.40	(1.0)	(1.0)	(1.0)	(1.0)	(1.0)	(1.0)	(1.0)	(1.0)	(1.0)		(1.0)	(1.0)	(1.0)	(1.0)	(1.0)	(1.0)	(1.0)	(1.0)	(1.0)	(1.0)
	.51			(42.4)	38.5	34.7	33.7	29.5	27.6	26.1					36.0	32.2	28.0	26.0	24.5	23.0	(17.3)
	.65			(43.6)	41.7	38.6	34.7	33.2	30.2	27.2				37.5	(32.8)	(25.8)	22.9	20.8	20.8	19.8	
	.79		42.3	42.8	41.1	38.2	34.7	33.2	30.2	27.5				38.3	(38.1)	35.2	28.0	26.1	22.9		
	1.00		(44.1)	43.0	41.0	38.2	34.7	33.2	30.2	27.5				(38.4)	(39.2)	37.6	30.0	27.5	25.6		
$\frac{t_w}{t_s}$	.40				(18.5)	18.8	11.8	10.8	8.3							(19.0)	14.1	13.0	12.3	10.2	(7.4)
	.51			(32.5)	21.0	12.8	12.0	11.9	11.9	9.1						25.6	14.8	10.5	10.5	8.4	13.2
	.65			(38.7)	21.9	11.2	11.2	11.2	12.2							(32.8)	(12.8)	10.5	8.8	15.8	
	.79		31.5	27.0	13.8	9.7	9.7	11.1	12.5						27.1	(13.8)	9.5	10.9			
	1.00		(29.0)	13.0	12.6	9.7	9.7	12.5							(30.0)	(15.5)	10.9	13.3			
$\frac{t_w}{t_s}$	.40				(1.94)	2.02	1.92	2.41	2.74	(2.74)						(2.42)	2.74	2.86	(2.74)	(2.74)	(3.43)
	.51			(1.94)	2.17	2.40	2.40	2.11	2.74	(2.74)						2.42	2.74	3.56	2.74	(2.74)	(2.63)
	.65			(1.77)	2.17	2.43	2.43	2.11	2.63	2.63						2.42	2.93	2.86	3.50	2.63	
	.79		1.94	2.08	2.28	2.14	2.14	2.63							2.42	(2.80)	2.79	2.65	2.65		
	1.00		(2.00)	2.23	2.73	2.83	2.83	2.63							(2.23)	(2.58)	2.68	2.65	2.65		
$\frac{t_w}{t_s}$	.40				(1.64)	2.61	3.80	4.58	6.24	11.6						2.42	3.80	5.27	6.28	8.60	11.8
	.51			(1.25)	2.21	3.41	5.04	5.89	7.27	11.2						1.95	3.41	5.04	6.28	8.60	11.8
	.65			(1.25)	2.21	3.41	5.04	5.89	7.27	11.2						1.95	3.41	5.04	6.28	8.60	11.8
	.79		1.08	1.81	2.86	4.07	5.76	6.76	8.15						1.70	(2.97)	4.21	5.04	6.28	8.60	11.8
	1.00		(1.56)	2.18	3.32	5.04	6.76	8.15							(1.40)	(2.43)	3.28	5.22	7.34		
$\frac{t_w}{t_s}$	.40				(.522)	.492	.568	.568	.568	.568						.608	.565	.565	.565	.565	.565
	.51				(.522)	.492	.568	.568	.568	.568						.608	.565	.565	.565	.565	.565
	.65			(.551)	.573	.515	.583	.583	.583	.583						.718	.710	.597	.597	.597	.597
	.79		.698	.689	.588	.588	.588	.588	.588	.588					.824	.786	.689	.689	.689	.689	.689
	1.00		(.690)	.672	.589	.589	.589	.589	.589	.589					(.834)	(.837)	.732	.673	.552		
$\frac{t_w}{t_s}$	.40				(.730)	.778	.792	.792	.792	.792						.901	.901	.901	.901	.901	.901
	.51				(.730)	.778	.792	.792	.792	.792						.901	.901	.901	.901	.901	.901
	.65			(.687)	.807	.838	.838	.838	.838	.838						.930	1.06	.931	.931	.931	.931
	.79		.756	.803	.833	.833	.833	.833	.833	.833					.955	(1.06)	.993	.993	.993	.993	.993
	1.00		(.784)	.852	.852	.852	.852	.852	.852	.852					(.887)	(1.01)	.995	.995	.995	.995	.995
$\frac{t_w}{t_s}$	.40				(33.0)	29.4	25.3	25.3	25.3	25.3						33.0	29.2	26.0	26.0	26.0	26.0
	.51				(33.0)	29.4	25.3	25.3	25.3	25.3						33.0	29.2	26.0	26.0	26.0	26.0
	.65			(30.0)	29.4	25.8	25.8	25.8	25.8	25.8						33.0	29.2	26.0	26.0	26.0	26.0
	.79		37.0	25.0	24.2	24.2	24.2	24.2	24.2	24.2					37.0	25.0	24.2	24.2	24.2	24.2	24.2
	1.00		(27.0)	25.6	22.9	22.9	22.9	22.9	22.9	22.9					(30.3)	(27.5)	22.4	22.4	22.4	22.4	22.4
$\frac{t_w}{t_s}$	.40				(37.5)	47.0	59.0	60.0	67.6	73.2						(37.5)	47.0	59.0	60.0	67.6	73.2
	.51				(31.7)	47.0	59.0	60.0	67.6	73.2						(31.7)	47.0	59.0	60.0	67.6	73.2
	.65			(26.2)	47.0	70.0	70.0	70.0	70.0	70.0						36.0	(41.1)	81.0	81.0	81.0	81.0
	.79		23.0	41.5	60.0	62.0	62.0	62.0	62.0	62.0					36.0	(41.1)	81.0	81.0	81.0	81.0	81.0
	1.00		(45.1)	68.0	62.0	62.0	62.0	62.0	62.0	62.0					(33.6)	(62.0)	81.0	81.0	81.0	81.0	81.0

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TABLE 8.- COMPARATIVE DESIGNS OF Y- AND Z-STIFFENED PANELS [ $P_1 = 8.0$  kips/inch;  $\alpha = 1$ ; minimum-weight designs in parentheses]

TABLE 8.- COMPARATIVE DESIGNS OF 1- AND 20-IN. DIFFERENTIAL PANELS $T_1 = 8.0$ KIPS/INCH, $\sigma = 1$ , MINIMUM WEIGHT DESIGNS IN PARENTHESES											
L = 20 in.											
L = 20 in.											
Panel	$t_w$	$t_b$ , in.									
		0.040	0.051	0.062	0.084	0.108	0.135	0.164	0.204	0.255	0.325
$\sigma_c$ (in.)	0.40	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	.51	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	.63	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	.79	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	1.00	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	.40	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	.51	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	.63	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	.79	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	1.00	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
$\sigma_c$ (in.)	.40	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	.51	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	.63	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	.79	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	1.00	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	.40	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	.51	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	.63	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	.79	758- 248- 248- 248- 248- 248- 248- 248- 248- 248-	(55.3)	(57.2)	(58.2)	(58.1)	(58.2)	(58.2)	(58.2)	(58.2)	(58.2)
	1.00	758- 248- 248- 248- 248- 248- 2									

TABLE 7. - COMPARATIVE DESIGNS OF Y- AND Z-STIFFENED PANELS - Continued

[illegible]

TABLE 8.- COMPARATIVE DESIGNS OF Y- AND Z-STIFFENED PANELS [ $P_1 = 5.0$  kips/inch;  $a = 1$ ; minimum-weight designs in parentheses]

TABLE 3. - COMPARATIVE DESIGN OF L- AND A-TYPE SUPPORTED PANELS (L = 20 in., t = .010 in., t <sub>w</sub> = .010 in., t <sub>s</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = .010 in., t <sub>h</sub> = 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TABLE 8.- COMPARATIVE DESIGNS OF X- AND Z-STIFFENED PANELS - Continued

	w t/s	Panel	L = 30 in. t <sub>z</sub> , in.								L = 40 in. t <sub>z</sub> , in.									
			0.051	0.064	0.081	0.102	0.125	0.156	0.184	0.204	0.252	0.051	0.064	0.081	0.102	0.125	0.156	0.184	0.204	0.252
E <sub>x</sub> (ksi)	0.40	Y 750-24	(49.0)	(44.5)	(41.8)	(38.8)	(36.3)	(34.1)	(32.0)	(30.7)	(28.0)	(42.9)	(37.7)	(34.9)	(32.9)	(30.8)	(28.5)	(26.0)	(24.5)	(22.0)
	.51	Y 750-24	(47.4)	(42.8)	(40.1)	(37.1)	(34.6)	(32.4)	(30.2)	(28.0)	(25.4)	(41.7)	(36.5)	(33.8)	(31.8)	(29.5)	(27.1)	(24.6)	(23.1)	(20.6)
	.63	Y 750-24	(46.8)	(42.2)	(39.5)	(36.5)	(34.0)	(31.8)	(29.6)	(27.4)	(24.8)	(41.2)	(36.0)	(33.3)	(31.3)	(29.0)	(26.6)	(24.1)	(22.6)	(20.1)
	.79	Y 750-24	(45.2)	(40.6)	(37.9)	(34.9)	(32.4)	(30.2)	(28.0)	(25.8)	(23.2)	(40.6)	(35.4)	(32.7)	(30.7)	(28.4)	(26.0)	(23.5)	(22.0)	(19.5)
	1.00	Y 750-24	(44.6)	(40.0)	(37.3)	(34.3)	(31.8)	(29.6)	(27.4)	(25.2)	(22.6)	(40.0)	(34.8)	(32.1)	(30.1)	(27.8)	(25.4)	(22.9)	(21.4)	(18.9)
G <sub>or</sub> (ksi)	.40	Y 750-24	(37.1)	(32.5)	(30.0)	(27.0)	(24.5)	(22.3)	(20.1)	(17.9)	(15.4)	(34.5)	(29.3)	(26.6)	(24.6)	(22.3)	(19.8)	(17.3)	(15.8)	(13.3)
	.51	Y 750-24	(35.5)	(30.9)	(28.4)	(25.4)	(22.9)	(20.7)	(18.5)	(16.3)	(13.8)	(32.9)	(27.7)	(25.0)	(23.0)	(20.7)	(18.2)	(15.7)	(14.2)	(11.7)
	.63	Y 750-24	(34.9)	(30.3)	(27.8)	(24.8)	(22.3)	(20.1)	(17.9)	(15.7)	(13.2)	(32.3)	(27.1)	(24.4)	(22.4)	(20.1)	(17.6)	(15.1)	(13.6)	(11.1)
	.79	Y 750-24	(33.3)	(28.7)	(26.2)	(23.2)	(20.7)	(18.5)	(16.3)	(14.1)	(11.6)	(31.7)	(26.5)	(23.8)	(21.8)	(19.5)	(17.0)	(14.5)	(13.0)	(10.5)
	1.00	Y 750-24	(32.7)	(28.1)	(25.6)	(22.6)	(20.1)	(17.9)	(15.7)	(13.5)	(11.0)	(31.1)	(25.9)	(23.2)	(21.2)	(18.9)	(16.4)	(13.9)	(12.4)	(9.9)
H (in.)	.40	Y 750-24	(2.15)	(2.16)	(2.16)	(2.16)	(2.16)	(2.16)	(2.16)	(2.16)	(2.16)	(2.70)	(2.85)	(2.95)	(3.05)	(3.15)	(3.25)	(3.35)	(3.45)	(3.55)
	.51	Y 750-24	(2.15)	(2.16)	(2.16)	(2.16)	(2.16)	(2.16)	(2.16)	(2.16)	(2.16)	(2.70)	(2.85)	(2.95)	(3.05)	(3.15)	(3.25)	(3.35)	(3.45)	(3.55)
	.63	Y 750-24	(2.09)	(2.15)	(2.16)	(2.16)	(2.16)	(2.16)	(2.16)	(2.16)	(2.16)	(2.69)	(2.84)	(2.94)	(3.04)	(3.14)	(3.24)	(3.34)	(3.44)	(3.54)
	.79	Y 750-24	(1.94)	(2.07)	(2.14)	(2.15)	(2.16)	(2.16)	(2.16)	(2.16)	(2.16)	(2.64)	(2.79)	(2.89)	(2.99)	(3.09)	(3.19)	(3.29)	(3.39)	(3.49)
	1.00	Y 750-24	(1.94)	(2.07)	(2.14)	(2.15)	(2.16)	(2.16)	(2.16)	(2.16)	(2.16)	(2.64)	(2.79)	(2.89)	(2.99)	(3.09)	(3.19)	(3.29)	(3.39)	(3.49)
s (in.)	.40	Y 750-24	(2.00)	(2.01)	(2.01)	(2.01)	(2.01)	(2.01)	(2.01)	(2.01)	(2.01)	(2.59)	(2.74)	(2.84)	(2.94)	(3.04)	(3.14)	(3.24)	(3.34)	(3.44)
	.51	Y 750-24	(1.99)	(2.00)	(2.00)	(2.00)	(2.00)	(2.00)	(2.00)	(2.00)	(2.00)	(2.58)	(2.73)	(2.83)	(2.93)	(3.03)	(3.13)	(3.23)	(3.33)	(3.43)
	.63	Y 750-24	(1.92)	(1.99)	(2.00)	(2.00)	(2.00)	(2.00)	(2.00)	(2.00)	(2.00)	(2.57)	(2.72)	(2.82)	(2.92)	(3.02)	(3.12)	(3.22)	(3.32)	(3.42)
	.79	Y 750-24	(1.86)	(1.94)	(1.97)	(1.99)	(2.00)	(2.00)	(2.00)	(2.00)	(2.00)	(2.56)	(2.71)	(2.81)	(2.91)	(3.01)	(3.11)	(3.21)	(3.31)	(3.41)
	1.00	Y 750-24	(1.86)	(1.94)	(1.97)	(1.99)	(2.00)	(2.00)	(2.00)	(2.00)	(2.00)	(2.56)	(2.71)	(2.81)	(2.91)	(3.01)	(3.11)	(3.21)	(3.31)	(3.41)
E (in.)	.40	Y 750-24	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)
	.51	Y 750-24	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)
	.63	Y 750-24	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)
	.79	Y 750-24	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)
	1.00	Y 750-24	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.655)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)	(.692)
p (in.)	.40	Y 750-24	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)
	.51	Y 750-24	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)
	.63	Y 750-24	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)
	.79	Y 750-24	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)
	1.00	Y 750-24	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)	(.761)
E <sub>y</sub> (ksi)	.40	Y 750-24	(20.0)	(18.0)	(16.0)	(14.0)	(12.0)	(10.0)	(8.0)	(6.0)	(4.0)	(20.0)	(18.0)	(16.0)	(14.0)	(12.0)	(10.0)	(8.0)	(6.0)	(4.0)
	.51	Y 750-24	(20.0)	(18.0)	(16.0)	(14.0)	(12.0)	(10.0)	(8.0)	(6.0)	(4.0)	(20.0)	(18.0)	(16.0)	(14.0)	(12.0)	(10.0)	(8.0)	(6.0)	(4.0)
	.63	Y 750-24	(20.0)	(18.0)	(16.0)	(14.0)	(12.0)	(10.0)	(8.0)	(6.0)	(4.0)	(20.0)	(18.0)	(16.0)	(14.0)	(12.0)	(10.0)	(8.0)	(6.0)	(4.0)
	.79	Y 750-24	(20.0)	(18.0)	(16.0)	(14.0)	(12.0)	(10.0)	(8.0)	(6.0)	(4.0)	(20.0)	(18.0)	(16.0)	(14.0)	(12.0)	(10.0)	(8.0)	(6.0)	(4.0)
	1.00	Y 750-24	(20.0)	(18.0)	(16.0)	(14.0)	(12.0)	(10.0)	(8.0)	(6.0)	(4.0)	(20.0)	(18.0)	(16.0)	(14.0)	(12.0)	(10.0)	(8.0)	(6.0)	(4.0)
E <sub>z</sub> (ksi)	.40	Y 750-24	(30.0)	(28.0)	(26.0)	(24.0)	(22.0)	(20.0)	(18.0)	(16.0)	(14.0)	(30.0)	(28.0)	(26.0)	(24.0)	(22.0)	(20.0)	(18.0)	(16.0)	(14.0)
	.51	Y 750-24	(30.0)	(28.0)	(26.0)	(24.0)	(22.0)	(20.0)	(18.0)	(16.0)	(14.0)	(30.0)	(28.0)	(26.0)	(24.0)	(22.0)	(20.0)	(18.0)	(16.0)	(14.0)
	.63	Y 750-24	(30.0)	(28.0)	(26.0)	(24.0)	(22.0)	(20.0)	(18.0)	(16.0)	(14.0)	(30.0)	(28.0)	(26.0)	(24.0)	(22.0)	(20.0)	(18.0)	(16.0)	(14.0)
	.79	Y 750-24	(30.0)	(28.0)	(26.0)	(24.0)	(22.0)	(20.0)	(18.0)	(16.0)	(14.0)	(30.0)	(28.0)	(26.0)	(24.0)	(22.0)	(20.0)	(18.0)	(16.0)	(14.0)
	1.00	Y 750-24	(30.0)	(28.0)	(26.0)	(24.0)	(22.0)	(20.0)	(18.0)	(16.0)	(14.0)	(30.0)	(28.0)	(26.0)	(24.0)	(22.0)	(20.0)	(18.0)	(16.0)	(14.0)

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TABLE 9.- X-PANEL PROPERTIES  $\frac{t}{b} = 0.40; \frac{b}{c} = 9.3; \frac{b}{d} = 1.04; \frac{t}{c} = 1.06; \frac{b}{d} = 0.94; \frac{t}{c} = 2.13; \frac{b}{d} = 0.69; \frac{t}{c} = 1; \frac{d}{t} = 1.5; \frac{b}{c} = 4.6$ 

$\frac{t}{b}$	$\frac{b}{c}$	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33
23	23	1.496	1.512	1.527	1.542	1.557	1.571	1.585	1.598	1.612	1.624	1.637	1.649	1.661	1.671	1.685	1.696
24	24	1.481	1.497	1.512	1.526	1.541	1.555	1.568	1.582	1.595	1.608	1.620	1.632	1.644	1.655	1.667	1.678
25	25	1.467	1.482	1.497	1.511	1.525	1.539	1.553	1.566	1.579	1.591	1.604	1.616	1.628	1.639	1.650	1.661
26	26	1.454	1.468	1.483	1.497	1.511	1.525	1.539	1.553	1.566	1.579	1.591	1.604	1.616	1.628	1.639	1.650
27	27	1.441	1.455	1.470	1.484	1.498	1.511	1.525	1.539	1.552	1.565	1.578	1.591	1.604	1.616	1.628	1.639
28	28	1.429	1.444	1.458	1.472	1.485	1.498	1.511	1.524	1.536	1.548	1.560	1.572	1.585	1.597	1.609	1.616
29	29	1.418	1.432	1.446	1.460	1.473	1.486	1.498	1.511	1.523	1.535	1.547	1.558	1.569	1.580	1.591	1.602
30	30	1.407	1.421	1.435	1.448	1.461	1.474	1.486	1.498	1.510	1.522	1.534	1.546	1.558	1.569	1.580	1.591
31	31	1.397	1.411	1.424	1.437	1.450	1.463	1.475	1.487	1.499	1.510	1.522	1.533	1.545	1.556	1.567	1.578
32	32	1.388	1.401	1.414	1.427	1.440	1.452	1.464	1.476	1.488	1.499	1.510	1.521	1.532	1.543	1.554	1.565
33	33	1.378	1.391	1.404	1.417	1.430	1.442	1.454	1.465	1.477	1.488	1.499	1.510	1.521	1.531	1.542	1.552
34	34	1.370	1.382	1.395	1.408	1.420	1.432	1.444	1.455	1.467	1.478	1.489	1.500	1.510	1.521	1.531	1.541
35	35	1.361	1.374	1.386	1.399	1.411	1.423	1.434	1.446	1.457	1.468	1.479	1.489	1.500	1.510	1.520	1.530
36	36	1.352	1.365	1.377	1.390	1.402	1.414	1.425	1.436	1.447	1.458	1.469	1.479	1.490	1.500	1.510	1.520
37	37	1.344	1.356	1.368	1.380	1.392	1.404	1.415	1.426	1.437	1.448	1.459	1.469	1.479	1.490	1.500	1.510
38	38	1.336	1.348	1.360	1.372	1.384	1.395	1.406	1.417	1.428	1.439	1.450	1.461	1.471	1.481	1.491	1.500
39	39	1.328	1.340	1.352	1.364	1.376	1.387	1.398	1.409	1.420	1.431	1.442	1.452	1.462	1.472	1.482	1.491
40	40	1.320	1.332	1.344	1.356	1.368	1.379	1.390	1.401	1.412	1.423	1.434	1.444	1.454	1.464	1.474	1.482
41	41	1.312	1.324	1.336	1.348	1.360	1.371	1.382	1.393	1.404	1.415	1.426	1.436	1.446	1.456	1.466	1.474
42	42	1.304	1.316	1.328	1.340	1.352	1.363	1.374	1.385	1.396	1.407	1.418	1.428	1.438	1.448	1.458	1.466
43	43	1.296	1.308	1.320	1.332	1.344	1.355	1.366	1.377	1.388	1.399	1.410	1.420	1.430	1.440	1.450	1.458
44	44	1.289	1.300	1.311	1.322	1.333	1.344	1.355	1.366	1.377	1.388	1.399	1.409	1.419	1.429	1.439	1.446
45	45	1.281	1.292	1.303	1.314	1.325	1.336	1.347	1.358	1.369	1.379	1.390	1.400	1.410	1.420	1.430	1.437
46	46	1.273	1.284	1.295	1.306	1.317	1.328	1.339	1.350	1.361	1.371	1.382	1.392	1.402	1.412	1.422	1.429
47	47	1.265	1.276	1.287	1.298	1.309	1.320	1.331	1.342	1.352	1.363	1.373	1.384	1.394	1.404	1.414	1.421
48	48	1.257	1.268	1.279	1.290	1.301	1.312	1.323	1.334	1.344	1.355	1.365	1.376	1.386	1.396	1.406	1.413
49	49	1.250	1.260	1.271	1.282	1.293	1.304	1.315	1.325	1.336	1.346	1.357	1.367	1.377	1.387	1.397	1.404
50	50	1.242	1.252	1.263	1.274	1.285	1.296	1.307	1.317	1.328	1.338	1.349	1.359	1.369	1.379	1.389	1.396
51	51	1.234	1.245	1.256	1.267	1.278	1.289	1.299	1.310	1.320	1.331	1.341	1.351	1.361	1.371	1.381	1.388
52	52	1.226	1.237	1.248	1.259	1.270	1.281	1.292	1.302	1.313	1.323	1.334	1.344	1.354	1.364	1.374	1.381
53	53	1.218	1.229	1.240	1.251	1.262	1.273	1.283	1.294	1.304	1.315	1.325	1.335	1.345	1.355	1.365	1.372
54	54	1.210	1.221	1.232	1.243	1.254	1.265	1.275	1.286	1.296	1.307	1.317	1.327	1.337	1.347	1.357	1.364
55	55	1.202	1.213	1.224	1.235	1.246	1.257	1.267	1.278	1.288	1.299	1.309	1.319	1.329	1.339	1.349	1.356
56	56	1.194	1.205	1.216	1.227	1.238	1.249	1.259	1.270	1.280	1.291	1.301	1.311	1.321	1.331	1.341	1.348
57	57	1.186	1.197	1.208	1.219	1.230	1.241	1.251	1.262	1.272	1.283	1.293	1.303	1.313	1.323	1.333	1.340
58	58	1.178	1.189	1.200	1.211	1.222	1.232	1.243	1.253	1.264	1.274	1.284	1.294	1.304	1.314	1.324	1.331
59	59	1.170	1.181	1.192	1.203	1.214	1.224	1.235	1.245	1.256	1.266	1.276	1.286	1.296	1.306	1.316	1.323
60	60	1.162	1.173	1.184	1.195	1.206	1.217	1.227	1.238	1.248	1.259	1.269	1.279	1.289	1.299	1.309	1.316
61	61	1.154	1.165	1.176	1.187	1.198	1.209	1.219	1.230	1.240	1.251	1.261	1.271	1.281	1.291	1.301	1.308
62	62	1.146	1.157	1.168	1.179	1.190	1.201	1.211	1.222	1.232	1.243	1.253	1.263	1.273	1.283	1.293	1.300
63	63	1.138	1.149	1.160	1.171	1.182	1.193	1.203	1.214	1.224	1.235	1.245	1.255	1.265	1.275	1.285	1.292
64	64	1.130	1.141	1.152	1.163	1.174	1.185	1.195	1.206	1.216	1.227	1.237	1.247	1.257	1.267	1.277	1.284
65	65	1.122	1.133	1.144	1.155	1.166	1.177	1.187	1.198	1.208	1.219	1.229	1.239	1.249	1.259	1.269	1.276
66	66	1.114	1.125	1.136	1.147	1.158	1.169	1.179	1.190	1.200	1.211	1.221	1.231	1.241	1.251	1.261	1.268
67	67	1.106	1.117	1.128	1.139	1.150	1.161	1.171	1.182	1.192	1.203	1.213	1.223	1.233	1.243	1.253	1.260
68	68	1.098	1.109	1.120	1.131	1.142	1.153	1.163	1.174	1.184	1.195	1.205	1.215	1.225	1.235	1.245	1.252
69	69	1.090	1.101	1.112	1.123	1.134	1.145	1.155	1.166	1.176	1.187	1.197	1.207	1.217	1.227	1.237	1.244
70	70	1.082	1.093	1.104	1.115	1.126	1.137	1.147	1.158	1.168	1.179	1.189	1.199	1.209	1.219	1.229	1.236
71	71	1.074	1.085	1.096	1.107	1.118	1.129	1.139	1.150	1.160	1.171	1.181	1.191	1.201	1.211	1.221	1.228
72	72	1.066	1.077	1.088	1.099	1.110	1.121	1.131	1.142	1.152	1.163	1.173	1.183	1.193	1.203	1.213	1.220
73	73	1.058	1.069	1.080	1.091	1.102	1.113	1.123	1.134	1.144	1.155	1.165	1.175	1.185	1.195	1.205	1.212
74	74	1.050	1.061	1.072	1.083	1.094	1.105	1.115	1.126	1.136	1.147	1.157	1.167	1.177	1.187	1.197	1.204
75	75	1.042	1.053	1.064	1.075	1.086	1.097	1.107	1.118	1.128	1.139	1.149	1.159	1.169	1.179	1.189	1.196
76	76	1.034	1.045	1.056	1.067	1.078	1.089	1.099	1.110	1.120	1.131	1.141	1.151	1.161	1.171	1.181	1.188
77	77	1.026	1.037	1.048	1.059	1.070	1.081	1.091	1.102	1.112	1.123	1.133	1.143	1.153	1.163	1.173	1.180
78	78	1.018	1.029	1.040	1.051	1.062	1.073	1.083	1.094	1.104	1.115	1.125	1.135	1.145	1.155	1.165	1.172
79	79	1.010	1.021	1.032	1.043	1.054	1.065	1.075	1.086	1.096	1.107	1.117	1.127	1.137	1.147	1.157	1.164
80	80	1.002	1.013	1.024	1.035	1.046	1.057	1.067	1.078	1.088	1.099	1.109	1.119	1.129	1.139	1.149	1.156
81	81	0.994	1.005	1.016	1.027	1.038	1.049	1.059	1.070	1.080	1.091	1.101	1.111	1.121	1.131	1.141	1.148
82	82	0.986	0.997	1.008	1.019	1.030	1.041	1.051	1.062	1.072	1.083	1.093	1.103	1.113	1.123	1.133	1.140
83	83	0.978	0.989	1.000	1.011	1.022	1.033	1.043	1.054	1.064	1.075	1.085	1.095	1.105	1.115	1.125	1.132
84	84	0.970	0.981	0.992	1.003	1.014	1.025	1.035	1.046	1.056	1.067	1.077	1.087	1.097	1.107	1.117	1.124
85	85	0.962	0.973	0.984	0.995	1.006	1.017	1.027	1.038	1.048	1.059	1.069	1.079	1.089	1.099	1.109	1.116
86	86	0.954	0.965	0.976	0.987	0.998	1.009	1.019	1.030	1.040	1.051	1.061	1.071	1.081	1.091	1.101	1.108
87	87	0.946	0.957	0.968	0.979	0.990	1.001	1.011	1.022	1.032	1.043	1.053	1.063	1.073	1.083	1.093	1.100
88	88	0.938	0.949	0.960	0.971	0.982	0.993	1.003	1.014	1.024	1.035	1.045	1.055	1.065	1.075	1.085	1.092
89	89	0.930	0.941	0.952	0.963	0.974	0.985	0.995	1.006	1.016	1.027	1.037	1.047	1.057			



TABLE 10 - Y-PANEL PROPERTIES  $\frac{b}{c} = 0.51$ ;  $\frac{b}{a} = 9.3$ ;  $\frac{b}{d} = 1.04$ ;  $\frac{b}{e} = 1.06$ ;  $\frac{b}{f} = 0.94$ ;  $\frac{b}{g} = 2.13$ ;  $\frac{b}{h} = 0.69$ ;  $\frac{b}{i} = 1$ ;  $\frac{b}{j} = 2.0$ ;  $\frac{b}{k} = 6.0$ 

$\frac{b}{c}$	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33
23	1.744	1.770	1.792	1.812	1.832	1.852	1.871	1.889	1.907	1.924	1.941	1.958	1.974	1.989	2.004	2.019
24	1.728	1.749	1.770	1.789	1.808	1.826	1.844	1.861	1.878	1.894	1.910	1.926	1.941	1.956	1.971	1.986
25	1.708	1.729	1.750	1.770	1.789	1.808	1.826	1.844	1.862	1.880	1.896	1.912	1.928	1.944	1.959	1.974
26	1.689	1.710	1.730	1.750	1.769	1.788	1.807	1.826	1.844	1.862	1.879	1.895	1.911	1.927	1.943	1.958
27	1.671	1.692	1.712	1.731	1.750	1.769	1.787	1.805	1.822	1.839	1.856	1.871	1.887	1.899	1.914	1.929
28	1.654	1.674	1.694	1.713	1.732	1.751	1.769	1.787	1.803	1.820	1.836	1.852	1.868	1.883	1.898	1.913
29	1.638	1.658	1.678	1.697	1.715	1.734	1.751	1.769	1.785	1.802	1.818	1.834	1.849	1.864	1.879	1.893
30	1.623	1.642	1.662	1.681	1.699	1.717	1.734	1.752	1.768	1.785	1.801	1.816	1.832	1.847	1.861	1.875
31	1.608	1.628	1.647	1.665	1.683	1.701	1.718	1.735	1.752	1.768	1.784	1.799	1.815	1.829	1.844	1.858
32	1.594	1.613	1.632	1.650	1.668	1.686	1.703	1.720	1.736	1.752	1.768	1.783	1.798	1.813	1.827	1.841
33	1.581	1.600	1.618	1.636	1.654	1.671	1.688	1.705	1.721	1.737	1.752	1.768	1.783	1.797	1.811	1.826
34	1.568	1.587	1.605	1.623	1.640	1.657	1.674	1.691	1.707	1.722	1.738	1.753	1.767	1.782	1.796	1.810
35	1.556	1.574	1.592	1.610	1.627	1.644	1.661	1.677	1.693	1.708	1.723	1.738	1.753	1.767	1.781	1.795
36	1.544	1.562	1.580	1.598	1.615	1.631	1.648	1.664	1.679	1.695	1.710	1.725	1.739	1.753	1.767	1.781
37	1.533	1.551	1.569	1.586	1.602	1.619	1.635	1.651	1.666	1.682	1.697	1.711	1.726	1.740	1.753	1.767
38	1.522	1.540	1.557	1.574	1.591	1.607	1.623	1.639	1.654	1.669	1.684	1.698	1.713	1.727	1.740	1.754
39	1.512	1.529	1.546	1.563	1.580	1.596	1.611	1.627	1.642	1.657	1.672	1.686	1.700	1.714	1.727	1.741
40	1.502	1.519	1.536	1.553	1.569	1.585	1.600	1.615	1.630	1.645	1.659	1.673	1.687	1.699	1.713	1.727
41	1.494	1.510	1.526	1.542	1.558	1.573	1.589	1.603	1.618	1.633	1.647	1.661	1.675	1.688	1.699	1.713
42	1.486	1.502	1.518	1.534	1.549	1.564	1.579	1.593	1.608	1.622	1.636	1.650	1.664	1.677	1.689	1.702
43	1.480	1.495	1.510	1.525	1.540	1.555	1.569	1.583	1.597	1.611	1.625	1.639	1.652	1.665	1.678	1.690
44	1.475	1.490	1.505	1.520	1.535	1.549	1.563	1.577	1.591	1.605	1.618	1.632	1.645	1.658	1.670	1.682
45	1.470	1.485	1.500	1.515	1.530	1.544	1.558	1.572	1.585	1.599	1.612	1.625	1.638	1.650	1.662	1.674
46	1.466	1.481	1.495	1.510	1.524	1.538	1.552	1.565	1.578	1.591	1.604	1.617	1.630	1.642	1.654	1.666
47	1.462	1.477	1.491	1.505	1.519	1.533	1.546	1.559	1.572	1.585	1.598	1.611	1.624	1.636	1.648	1.660
48	1.459	1.473	1.487	1.501	1.514	1.528	1.541	1.554	1.567	1.580	1.593	1.605	1.618	1.630	1.642	1.654
49	1.456	1.470	1.484	1.497	1.511	1.524	1.537	1.550	1.563	1.575	1.588	1.600	1.612	1.624	1.636	1.648
50	1.453	1.467	1.480	1.493	1.506	1.519	1.532	1.545	1.557	1.570	1.582	1.595	1.607	1.619	1.631	1.643
51	1.450	1.464	1.477	1.490	1.503	1.516	1.528	1.541	1.553	1.565	1.577	1.589	1.601	1.613	1.625	1.637
52	1.448	1.461	1.474	1.487	1.499	1.512	1.524	1.536	1.548	1.560	1.572	1.584	1.595	1.607	1.619	1.631
53	1.446	1.459	1.471	1.484	1.496	1.508	1.520	1.532	1.544	1.556	1.567	1.579	1.590	1.602	1.613	1.625
54	1.444	1.457	1.469	1.481	1.493	1.505	1.517	1.528	1.540	1.551	1.563	1.574	1.585	1.596	1.608	1.619
55	1.442	1.455	1.467	1.479	1.491	1.503	1.514	1.526	1.537	1.548	1.559	1.570	1.581	1.592	1.603	1.614
56	1.440	1.453	1.465	1.477	1.489	1.501	1.512	1.523	1.534	1.545	1.556	1.567	1.578	1.589	1.600	1.611
57	1.438	1.451	1.463	1.475	1.487	1.498	1.510	1.521	1.532	1.543	1.554	1.565	1.575	1.586	1.597	1.608
58	1.436	1.449	1.461	1.473	1.485	1.496	1.508	1.519	1.530	1.541	1.552	1.562	1.573	1.583	1.594	1.605
59	1.434	1.447	1.459	1.471	1.483	1.494	1.506	1.517	1.528	1.539	1.549	1.560	1.570	1.581	1.591	1.602
60	1.432	1.445	1.457	1.469	1.481	1.492	1.504	1.515	1.526	1.537	1.547	1.558	1.568	1.578	1.589	1.600
61	1.430	1.443	1.455	1.467	1.479	1.490	1.502	1.513	1.524	1.535	1.545	1.556	1.566	1.576	1.587	1.598
62	1.428	1.441	1.453	1.465	1.477	1.488	1.499	1.510	1.521	1.532	1.542	1.553	1.563	1.573	1.584	1.595
63	1.426	1.439	1.451	1.463	1.475	1.486	1.497	1.508	1.519	1.530	1.540	1.551	1.561	1.571	1.582	1.593
64	1.424	1.437	1.449	1.461	1.473	1.484	1.495	1.506	1.517	1.527	1.538	1.548	1.558	1.568	1.579	1.590
65	1.422	1.435	1.447	1.459	1.471	1.482	1.493	1.504	1.515	1.525	1.536	1.546	1.556	1.566	1.577	1.588
66	1.420	1.433	1.445	1.457	1.469	1.480	1.491	1.502	1.513	1.523	1.534	1.544	1.554	1.564	1.575	1.586
67	1.418	1.431	1.443	1.455	1.467	1.478	1.489	1.500	1.511	1.521	1.532	1.542	1.552	1.562	1.573	1.584
68	1.416	1.429	1.441	1.453	1.465	1.476	1.487	1.498	1.509	1.519	1.530	1.540	1.550	1.560	1.571	1.582
69	1.414	1.427	1.439	1.451	1.463	1.474	1.485	1.496	1.507	1.517	1.528	1.538	1.548	1.558	1.569	1.580
70	1.412	1.425	1.437	1.449	1.461	1.472	1.483	1.494	1.505	1.515	1.526	1.536	1.546	1.556	1.567	1.578
71	1.410	1.423	1.435	1.447	1.459	1.470	1.481	1.492	1.503	1.513	1.524	1.534	1.544	1.554	1.565	1.576
72	1.408	1.421	1.433	1.445	1.457	1.468	1.479	1.490	1.501	1.511	1.522	1.532	1.542	1.552	1.563	1.574
73	1.406	1.419	1.431	1.443	1.455	1.466	1.477	1.488	1.499	1.509	1.520	1.530	1.540	1.550	1.561	1.572
74	1.404	1.417	1.429	1.441	1.453	1.464	1.475	1.486	1.497	1.507	1.518	1.528	1.538	1.548	1.559	1.570
75	1.402	1.415	1.427	1.439	1.451	1.462	1.473	1.484	1.495	1.505	1.516	1.526	1.536	1.546	1.557	1.568
76	1.400	1.413	1.425	1.437	1.449	1.460	1.471	1.482	1.493	1.503	1.514	1.524	1.534	1.544	1.555	1.566
77	1.398	1.411	1.423	1.435	1.447	1.458	1.469	1.480	1.491	1.501	1.512	1.522	1.532	1.542	1.553	1.564
78	1.396	1.409	1.421	1.433	1.445	1.456	1.467	1.478	1.489	1.500	1.510	1.520	1.530	1.540	1.551	1.562
79	1.394	1.407	1.419	1.431	1.443	1.454	1.465	1.476	1.487	1.497	1.508	1.518	1.528	1.538	1.549	1.560
80	1.392	1.405	1.417	1.429	1.441	1.452	1.463	1.474	1.485	1.495	1.506	1.516	1.526	1.536	1.547	1.558
81	1.390	1.403	1.415	1.427	1.439	1.450	1.461	1.472	1.483	1.493	1.504	1.514	1.524	1.534	1.545	1.556
82	1.388	1.401	1.413	1.425	1.437	1.448	1.459	1.470	1.481	1.491	1.502	1.512	1.522	1.532	1.543	1.554
83	1.386	1.399	1.411	1.423	1.435	1.446	1.457	1.468	1.479	1.489	1.499	1.510	1.520	1.530	1.541	1.552
84	1.384	1.397	1.409	1.421	1.433	1.444	1.455	1.466	1.477	1.487	1.498	1.508	1.518	1.528	1.539	1.550
85	1.382	1.395	1.407	1.419	1.431	1.442	1.453	1.464	1.475	1.485	1.496	1.506	1.516	1.526	1.537	1.548
86	1.380	1.393	1.405	1.417	1.429	1.440	1.451	1.462	1.473	1.483	1.494	1.504	1.514	1.524	1.535	1.546
87	1.378	1.391	1.403	1.415	1.427	1.438	1.449	1.460	1.471	1.481	1.492	1.502	1.512	1.522	1.533	1.544
88	1.376	1.389	1.401	1.413	1.425	1.436	1.447	1.458	1.469	1.479	1.490	1.500	1.510	1.520	1.531	1.542
89	1.374	1.387	1.399	1.411	1.423	1.434	1.445	1.456	1.467	1.477	1.488	1.498	1.508	1.518	1.529	1.540
90	1.372	1.385	1.397	1.409	1.421	1.432	1.443	1.454	1.465	1.475	1.486	1.496	1.506	1.516	1.527	1.538
91	1.370	1.383	1.395	1.407	1.419	1.430	1.441	1.452	1.463	1.473	1.484	1.494	1.504	1.514	1.525	1.536



TABLE 11.-Y-PANEL PROPERTIES  $\frac{b}{c} = 0.63$ ;  $\frac{a}{b} = 9.3$ ;  $\frac{b}{c} = 1.04$ ;  $\frac{L}{b} = 1.06$ ;  $\frac{L}{c} = 0.94$ ;  $\frac{L}{b} = 2.13$ ;  $\frac{L}{c} = 0.69$ ;  $\frac{L}{b} = 1$ ;  $\frac{L}{c} = 1.6$ ;  $\frac{L}{b} = 6.1$

$\frac{L}{b}$	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33
23	2.059	2.080	2.115	2.148	2.167	2.192	2.216	2.240	2.263	2.285	2.306	2.327	2.347	2.366	2.385	2.404
24	2.031	2.059	2.087	2.113	2.135	2.164	2.188	2.211	2.234	2.255	2.277	2.297	2.316	2.335	2.353	2.375
25	2.005	2.033	2.060	2.086	2.112	2.136	2.160	2.184	2.206	2.228	2.249	2.270	2.290	2.309	2.328	2.347
26	1.980	2.008	2.035	2.061	2.086	2.110	2.134	2.157	2.180	2.201	2.222	2.243	2.263	2.282	2.301	2.320
27	1.956	1.984	2.010	2.036	2.061	2.085	2.109	2.132	2.154	2.176	2.197	2.217	2.238	2.257	2.276	2.294
28	1.934	1.961	1.987	2.013	2.038	2.062	2.085	2.108	2.130	2.151	2.172	2.193	2.213	2.232	2.251	2.269
29	1.912	1.939	1.965	1.990	2.015	2.039	2.062	2.085	2.107	2.128	2.149	2.169	2.189	2.208	2.227	2.245
30	1.892	1.918	1.944	1.969	1.993	2.017	2.040	2.062	2.084	2.105	2.126	2.146	2.166	2.185	2.204	2.222
31	1.872	1.898	1.924	1.948	1.972	1.996	2.019	2.041	2.063	2.084	2.104	2.124	2.144	2.163	2.182	2.200
32	1.853	1.879	1.904	1.929	1.953	1.976	1.999	2.021	2.042	2.063	2.084	2.104	2.123	2.142	2.161	2.179
33	1.835	1.861	1.886	1.910	1.934	1.957	1.979	2.001	2.022	2.043	2.064	2.084	2.103	2.122	2.140	2.158
34	1.818	1.843	1.868	1.892	1.915	1.938	1.960	1.982	2.003	2.024	2.044	2.064	2.083	2.102	2.120	2.138
35	1.801	1.826	1.851	1.874	1.898	1.921	1.943	1.965	1.986	2.006	2.026	2.045	2.064	2.083	2.101	2.119
36	1.785	1.810	1.834	1.858	1.881	1.903	1.925	1.946	1.967	1.987	2.007	2.027	2.046	2.064	2.083	2.100
37	1.770	1.794	1.818	1.842	1.864	1.887	1.908	1.929	1.950	1.970	1.990	2.009	2.028	2.047	2.065	2.082
38	1.755	1.779	1.803	1.826	1.849	1.871	1.892	1.913	1.934	1.954	1.973	1.992	2.011	2.029	2.047	2.065
39	1.741	1.765	1.788	1.811	1.834	1.855	1.877	1.897	1.918	1.938	1.957	1.976	1.995	2.013	2.031	2.048
40	1.727	1.751	1.774	1.797	1.819	1.840	1.861	1.882	1.902	1.922	1.941	1.960	1.979	1.997	2.015	2.032
41	1.712	1.736	1.759	1.781	1.803	1.824	1.845	1.865	1.885	1.904	1.923	1.942	1.961	1.979	1.997	2.014
42	1.698	1.722	1.745	1.767	1.788	1.809	1.829	1.849	1.868	1.887	1.906	1.925	1.943	1.961	1.979	1.996
43	1.684	1.708	1.731	1.753	1.774	1.795	1.815	1.835	1.854	1.873	1.892	1.911	1.930	1.948	1.966	1.983
44	1.670	1.694	1.717	1.739	1.760	1.780	1.800	1.819	1.838	1.856	1.875	1.893	1.911	1.929	1.946	1.963
45	1.656	1.680	1.703	1.725	1.745	1.765	1.784	1.803	1.822	1.840	1.858	1.876	1.894	1.911	1.928	1.945
46	1.642	1.666	1.689	1.711	1.731	1.751	1.770	1.789	1.807	1.825	1.843	1.861	1.878	1.895	1.912	1.929
47	1.628	1.652	1.675	1.697	1.717	1.736	1.755	1.773	1.791	1.809	1.826	1.844	1.861	1.878	1.895	1.912
48	1.614	1.638	1.661	1.683	1.703	1.722	1.741	1.759	1.777	1.794	1.811	1.828	1.845	1.862	1.879	1.895
49	1.599	1.623	1.646	1.668	1.688	1.707	1.725	1.743	1.761	1.778	1.795	1.812	1.829	1.845	1.862	1.878
50	1.585	1.609	1.632	1.654	1.674	1.692	1.710	1.728	1.745	1.762	1.779	1.795	1.812	1.828	1.845	1.861
51	1.571	1.595	1.618	1.639	1.659	1.677	1.695	1.712	1.729	1.746	1.762	1.779	1.795	1.811	1.828	1.844
52	1.557	1.581	1.604	1.625	1.645	1.663	1.681	1.698	1.715	1.731	1.748	1.764	1.780	1.796	1.812	1.828
53	1.543	1.567	1.590	1.611	1.631	1.649	1.667	1.684	1.700	1.717	1.733	1.749	1.765	1.781	1.797	1.812
54	1.529	1.553	1.576	1.597	1.617	1.635	1.653	1.670	1.686	1.702	1.718	1.734	1.750	1.766	1.781	1.797
55	1.515	1.539	1.562	1.583	1.603	1.621	1.639	1.656	1.672	1.688	1.704	1.720	1.736	1.751	1.767	1.782
56	1.499	1.523	1.546	1.567	1.587	1.605	1.623	1.640	1.656	1.672	1.688	1.704	1.720	1.735	1.751	1.766
57	1.485	1.509	1.532	1.553	1.573	1.591	1.609	1.626	1.642	1.658	1.674	1.689	1.705	1.720	1.735	1.750
58	1.471	1.495	1.518	1.539	1.559	1.577	1.595	1.612	1.628	1.644	1.659	1.675	1.690	1.705	1.720	1.735
59	1.457	1.481	1.504	1.525	1.545	1.563	1.581	1.598	1.614	1.630	1.645	1.660	1.675	1.690	1.705	1.720
60	1.443	1.467	1.490	1.511	1.531	1.549	1.567	1.584	1.600	1.616	1.631	1.646	1.661	1.676	1.690	1.705
61	1.429	1.453	1.476	1.497	1.517	1.535	1.553	1.570	1.586	1.601	1.616	1.631	1.646	1.661	1.675	1.690
62	1.415	1.439	1.462	1.483	1.503	1.521	1.539	1.556	1.572	1.587	1.602	1.616	1.631	1.645	1.660	1.675
63	1.402	1.426	1.448	1.469	1.489	1.507	1.525	1.542	1.558	1.573	1.588	1.602	1.616	1.631	1.645	1.660
64	1.388	1.412	1.435	1.456	1.476	1.494	1.512	1.529	1.545	1.560	1.575	1.589	1.603	1.617	1.631	1.645
65	1.374	1.398	1.421	1.442	1.462	1.480	1.498	1.515	1.531	1.546	1.560	1.574	1.588	1.602	1.616	1.630
66	1.360	1.384	1.407	1.428	1.448	1.466	1.484	1.501	1.517	1.532	1.546	1.560	1.574	1.588	1.602	1.616
67	1.346	1.370	1.393	1.414	1.434	1.452	1.470	1.487	1.503	1.518	1.532	1.546	1.560	1.574	1.588	1.602
68	1.332	1.356	1.379	1.400	1.420	1.438	1.456	1.473	1.489	1.504	1.518	1.532	1.546	1.560	1.574	1.588
69	1.318	1.342	1.365	1.386	1.406	1.424	1.442	1.459	1.475	1.490	1.504	1.518	1.532	1.546	1.560	1.574
70	1.304	1.328	1.351	1.372	1.392	1.410	1.428	1.445	1.461	1.476	1.490	1.504	1.518	1.532	1.546	1.560
71	1.290	1.314	1.337	1.358	1.378	1.396	1.414	1.431	1.447	1.462	1.476	1.490	1.504	1.518	1.532	1.546
72	1.276	1.300	1.323	1.344	1.364	1.382	1.400	1.417	1.433	1.448	1.462	1.476	1.490	1.504	1.518	1.532
73	1.262	1.286	1.309	1.330	1.350	1.368	1.386	1.403	1.419	1.434	1.448	1.462	1.476	1.490	1.504	1.518
74	1.248	1.272	1.295	1.316	1.336	1.354	1.372	1.389	1.405	1.420	1.434	1.448	1.462	1.476	1.490	1.504
75	1.234	1.258	1.281	1.302	1.322	1.340	1.358	1.375	1.391	1.406	1.420	1.434	1.448	1.462	1.476	1.490
76	1.220	1.244	1.267	1.288	1.308	1.326	1.344	1.361	1.377	1.392	1.406	1.420	1.434	1.448	1.462	1.476
77	1.206	1.230	1.253	1.274	1.294	1.312	1.330	1.347	1.363	1.378	1.392	1.406	1.420	1.434	1.448	1.462
78	1.192	1.216	1.239	1.260	1.280	1.298	1.316	1.333	1.349	1.364	1.378	1.392	1.406	1.420	1.434	1.448
79	1.178	1.202	1.225	1.246	1.266	1.284	1.302	1.319	1.335	1.350	1.364	1.378	1.392	1.406	1.420	1.434
80	1.164	1.188	1.211	1.232	1.252	1.270	1.288	1.305	1.321	1.336	1.350	1.364	1.378	1.392	1.406	1.420
81	1.150	1.174	1.197	1.218	1.238	1.256	1.274	1.291	1.307	1.322	1.336	1.350	1.364	1.378	1.392	1.406
82	1.136	1.160	1.183	1.204	1.224	1.242	1.260	1.277	1.293	1.308	1.322	1.336	1.350	1.364	1.378	1.392
83	1.122	1.146	1.169	1.190	1.210	1.228	1.246	1.263	1.279	1.294	1.308	1.322	1.336	1.350	1.364	1.378
84	1.108	1.132	1.155	1.176	1.196	1.214	1.232	1.249	1.265	1.280	1.294	1.308	1.322	1.336	1.350	1.364
85	1.094	1.118	1.141	1.162	1.182	1.200	1.218	1.235	1.251	1.266	1.280	1.294	1.308	1.322	1.336	1.350

TABLE 12.-Y-PANEL PROPERTIES  $\frac{b}{c} = 0.731$ ;  $\frac{b}{a} = 0.3$ ;  $\frac{b}{d} = 1.04$ ;  $\frac{b}{e} = 1.06$ ;  $\frac{b}{f} = 0.94$ ;  $\frac{b}{g} = 2.13$ ;  $\frac{b}{h} = 0.69$ ;  $\frac{b}{i} = 1$ ;  $\frac{b}{j} = 2.3$ ;  $\frac{b}{k} = 7.7$ 

$\frac{b}{c}$	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33
23	2.517	2.514	2.510	2.506	2.502	2.498	2.494	2.490	2.486	2.482	2.478	2.474	2.470	2.466	2.462	2.458
24	2.511	2.508	2.504	2.500	2.496	2.492	2.488	2.484	2.480	2.476	2.472	2.468	2.464	2.460	2.456	2.452
25	2.505	2.502	2.498	2.494	2.490	2.486	2.482	2.478	2.474	2.470	2.466	2.462	2.458	2.454	2.450	2.446
26	2.499	2.496	2.492	2.488	2.484	2.480	2.476	2.472	2.468	2.464	2.460	2.456	2.452	2.448	2.444	2.440
27	2.493	2.490	2.486	2.482	2.478	2.474	2.470	2.466	2.462	2.458	2.454	2.450	2.446	2.442	2.438	2.434
28	2.487	2.484	2.480	2.476	2.472	2.468	2.464	2.460	2.456	2.452	2.448	2.444	2.440	2.436	2.432	2.428
29	2.481	2.478	2.474	2.470	2.466	2.462	2.458	2.454	2.450	2.446	2.442	2.438	2.434	2.430	2.426	2.422
30	2.475	2.472	2.468	2.464	2.460	2.456	2.452	2.448	2.444	2.440	2.436	2.432	2.428	2.424	2.420	2.416
31	2.469	2.466	2.462	2.458	2.454	2.450	2.446	2.442	2.438	2.434	2.430	2.426	2.422	2.418	2.414	2.410
32	2.463	2.460	2.456	2.452	2.448	2.444	2.440	2.436	2.432	2.428	2.424	2.420	2.416	2.412	2.408	2.404
33	2.457	2.454	2.450	2.446	2.442	2.438	2.434	2.430	2.426	2.422	2.418	2.414	2.410	2.406	2.402	2.398
34	2.451	2.448	2.444	2.440	2.436	2.432	2.428	2.424	2.420	2.416	2.412	2.408	2.404	2.400	2.396	2.392
35	2.445	2.442	2.438	2.434	2.430	2.426	2.422	2.418	2.414	2.410	2.406	2.402	2.398	2.394	2.390	2.386
36	2.439	2.436	2.432	2.428	2.424	2.420	2.416	2.412	2.408	2.404	2.400	2.396	2.392	2.388	2.384	2.380
37	2.433	2.430	2.426	2.422	2.418	2.414	2.410	2.406	2.402	2.398	2.394	2.390	2.386	2.382	2.378	2.374
38	2.427	2.424	2.420	2.416	2.412	2.408	2.404	2.400	2.396	2.392	2.388	2.384	2.380	2.376	2.372	2.368
39	2.421	2.418	2.414	2.410	2.406	2.402	2.398	2.394	2.390	2.386	2.382	2.378	2.374	2.370	2.366	2.362
40	2.415	2.412	2.408	2.404	2.400	2.396	2.392	2.388	2.384	2.380	2.376	2.372	2.368	2.364	2.360	2.356
41	2.409	2.406	2.402	2.398	2.394	2.390	2.386	2.382	2.378	2.374	2.370	2.366	2.362	2.358	2.354	2.350
42	2.403	2.400	2.396	2.392	2.388	2.384	2.380	2.376	2.372	2.368	2.364	2.360	2.356	2.352	2.348	2.344
43	2.397	2.394	2.390	2.386	2.382	2.378	2.374	2.370	2.366	2.362	2.358	2.354	2.350	2.346	2.342	2.338
44	2.391	2.388	2.384	2.380	2.376	2.372	2.368	2.364	2.360	2.356	2.352	2.348	2.344	2.340	2.336	2.332
45	2.385	2.382	2.378	2.374	2.370	2.366	2.362	2.358	2.354	2.350	2.346	2.342	2.338	2.334	2.330	2.326
46	2.379	2.376	2.372	2.368	2.364	2.360	2.356	2.352	2.348	2.344	2.340	2.336	2.332	2.328	2.324	2.320
47	2.373	2.370	2.366	2.362	2.358	2.354	2.350	2.346	2.342	2.338	2.334	2.330	2.326	2.322	2.318	2.314
48	2.367	2.364	2.360	2.356	2.352	2.348	2.344	2.340	2.336	2.332	2.328	2.324	2.320	2.316	2.312	2.308
49	2.361	2.358	2.354	2.350	2.346	2.342	2.338	2.334	2.330	2.326	2.322	2.318	2.314	2.310	2.306	2.302
50	2.355	2.352	2.348	2.344	2.340	2.336	2.332	2.328	2.324	2.320	2.316	2.312	2.308	2.304	2.300	2.296
51	2.349	2.346	2.342	2.338	2.334	2.330	2.326	2.322	2.318	2.314	2.310	2.306	2.302	2.298	2.294	2.290
52	2.343	2.340	2.336	2.332	2.328	2.324	2.320	2.316	2.312	2.308	2.304	2.300	2.296	2.292	2.288	2.284
53	2.337	2.334	2.330	2.326	2.322	2.318	2.314	2.310	2.306	2.302	2.298	2.294	2.290	2.286	2.282	2.278
54	2.331	2.328	2.324	2.320	2.316	2.312	2.308	2.304	2.300	2.296	2.292	2.288	2.284	2.280	2.276	2.272
55	2.325	2.322	2.318	2.314	2.310	2.306	2.302	2.298	2.294	2.290	2.286	2.282	2.278	2.274	2.270	2.266
56	2.319	2.316	2.312	2.308	2.304	2.300	2.296	2.292	2.288	2.284	2.280	2.276	2.272	2.268	2.264	2.260
57	2.313	2.310	2.306	2.302	2.298	2.294	2.290	2.286	2.282	2.278	2.274	2.270	2.266	2.262	2.258	2.254
58	2.307	2.304	2.300	2.296	2.292	2.288	2.284	2.280	2.276	2.272	2.268	2.264	2.260	2.256	2.252	2.248
59	2.301	2.298	2.294	2.290	2.286	2.282	2.278	2.274	2.270	2.266	2.262	2.258	2.254	2.250	2.246	2.242
60	2.295	2.292	2.288	2.284	2.280	2.276	2.272	2.268	2.264	2.260	2.256	2.252	2.248	2.244	2.240	2.236
61	2.289	2.286	2.282	2.278	2.274	2.270	2.266	2.262	2.258	2.254	2.250	2.246	2.242	2.238	2.234	2.230
62	2.283	2.280	2.276	2.272	2.268	2.264	2.260	2.256	2.252	2.248	2.244	2.240	2.236	2.232	2.228	2.224
63	2.277	2.274	2.270	2.266	2.262	2.258	2.254	2.250	2.246	2.242	2.238	2.234	2.230	2.226	2.222	2.218
64	2.271	2.268	2.264	2.260	2.256	2.252	2.248	2.244	2.240	2.236	2.232	2.228	2.224	2.220	2.216	2.212
65	2.265	2.262	2.258	2.254	2.250	2.246	2.242	2.238	2.234	2.230	2.226	2.222	2.218	2.214	2.210	2.206
66	2.259	2.256	2.252	2.248	2.244	2.240	2.236	2.232	2.228	2.224	2.220	2.216	2.212	2.208	2.204	2.200
67	2.253	2.250	2.246	2.242	2.238	2.234	2.230	2.226	2.222	2.218	2.214	2.210	2.206	2.202	2.198	2.194
68	2.247	2.244	2.240	2.236	2.232	2.228	2.224	2.220	2.216	2.212	2.208	2.204	2.200	2.196	2.192	2.188
69	2.241	2.238	2.234	2.230	2.226	2.222	2.218	2.214	2.210	2.206	2.202	2.198	2.194	2.190	2.186	2.182
70	2.235	2.232	2.228	2.224	2.220	2.216	2.212	2.208	2.204	2.200	2.196	2.192	2.188	2.184	2.180	2.176
71	2.229	2.226	2.222	2.218	2.214	2.210	2.206	2.202	2.198	2.194	2.190	2.186	2.182	2.178	2.174	2.170
72	2.223	2.220	2.216	2.212	2.208	2.204	2.200	2.196	2.192	2.188	2.184	2.180	2.176	2.172	2.168	2.164
73	2.217	2.214	2.210	2.206	2.202	2.198	2.194	2.190	2.186	2.182	2.178	2.174	2.170	2.166	2.162	2.158
74	2.211	2.208	2.204	2.200	2.196	2.192	2.188	2.184	2.180	2.176	2.172	2.168	2.164	2.160	2.156	2.152
75	2.205	2.202	2.198	2.194	2.190	2.186	2.182	2.178	2.174	2.170	2.166	2.162	2.158	2.154	2.150	2.146
76	2.199	2.196	2.192	2.188	2.184	2.180	2.176	2.172	2.168	2.164	2.160	2.156	2.152	2.148	2.144	2.140
77	2.193	2.190	2.186	2.182	2.178	2.174	2.170	2.166	2.162	2.158	2.154	2.150	2.146	2.142	2.138	2.134
78	2.187	2.184	2.180	2.176	2.172	2.168	2.164	2.160	2.156	2.152	2.148	2.144	2.140	2.136	2.132	2.128
79	2.181	2.178	2.174	2.170	2.166	2.162	2.158	2.154	2.150	2.146	2.142	2.138	2.134	2.130	2.126	2.122
80	2.175	2.172	2.168	2.164	2.160	2.156	2.152	2.148	2.144	2.140	2.136	2.132	2.128	2.124	2.120	2.116
81	2.169	2.166	2.162	2.158	2.154	2.150	2.146	2.142	2.138	2.134	2.130	2.126	2.122	2.118	2.114	2.110
82	2.163	2.160	2.156	2.152	2.148	2.144	2.140	2.136	2.132	2.128	2.124	2.120	2.116	2.112	2.108	2.104
83	2.157	2.154	2.150	2.146	2.142	2.138	2.134	2.130	2.126	2.122	2.118	2.114	2.110	2.106	2.102	2.098
84	2.151	2.148	2.144	2.140	2.136	2.132	2.128	2.124	2.120	2.116	2.112	2.108	2.104	2.100	2.096	2.092
85	2.145	2.142	2.138	2.134	2.130	2.126	2.122	2.118	2.114	2.110	2.106	2.102	2.098	2.094	2.090	2.086
86	2.139	2.136	2.132	2.128	2.124	2.120	2.116	2.112	2.108	2.104	2.100	2.096	2.092	2.088	2.084	2.080
87	2.133	2.130	2.126	2.122	2.118	2.114	2.110	2.106	2.102	2.098	2.094	2.090	2.086	2.082	2.078	2.074
88	2.127	2.124	2.120	2.116	2.112	2.108	2.104	2.100	2.096	2.092	2.088	2.084	2.080	2.076	2.072	2.068
89	2.121	2.118	2.114	2.110	2.106	2.102	2.098	2.094	2.090	2.086	2.082	2.078	2.074	2.070	2.066	2.062
90	2.115	2.112	2.108	2.104	2.100	2.096	2.092	2.088	2.084	2.080	2.076	2.072	2.068	2.064	2.060	2.056
91	2.109	2.106	2.102	2.098	2.094	2.090	2.086	2.082	2.078	2.074	2.070	2.066	2.062	2.058	2.054	2.050

TABLE 13.- Y-PANEL PROPERTIES  $\left[ \frac{t_V}{t_H} = 1.00; \frac{b_A}{t_H} = 9.3; \frac{b_V}{b_H} = 1.04; \frac{t_B}{t_H} = 1.06; \frac{b_L}{b_H} = 0.94; \frac{t_F}{t_H} = 2.13; \frac{b_F}{b_H} = 0.69; \frac{r}{b_H} = 1; \frac{d}{b_H} = 2.4; \frac{p}{b_H} = 7.8 \right]$

	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33
23	1.177	1.225	1.270	1.313	1.355	1.395	1.433	1.470	1.505	1.539	1.572	1.604	1.634	1.663	1.692	1.720
24	1.171	1.178	1.183	1.187	1.190	1.192	1.194	1.195	1.196	1.197	1.198	1.199	1.200	1.201	1.202	1.203
25	1.086	1.133	1.179	1.222	1.264	1.304	1.343	1.380	1.416	1.451	1.485	1.518	1.550	1.581	1.612	1.642
26	1.089	1.089	1.089	1.089	1.089	1.089	1.089	1.089	1.089	1.089	1.089	1.089	1.089	1.089	1.089	1.089
27	2.963	3.020	3.055	3.098	3.140	3.181	3.219	3.257	3.293	3.328	3.362	3.394	3.426	3.456	3.486	3.514
28	2.925	2.972	3.017	3.060	3.102	3.142	3.181	3.219	3.255	3.290	3.324	3.356	3.388	3.418	3.448	3.476
29	2.888	2.935	2.980	3.023	3.065	3.105	3.144	3.182	3.218	3.253	3.287	3.320	3.352	3.383	3.413	3.442
30	2.851	2.900	2.948	2.998	3.048	3.097	3.145	3.192	3.238	3.283	3.327	3.370	3.412	3.453	3.493	3.531
31	2.815	2.865	2.915	2.965	3.015	3.063	3.110	3.156	3.201	3.245	3.288	3.330	3.371	3.411	3.450	3.488
32	2.780	2.830	2.877	2.923	2.968	3.012	3.054	3.095	3.135	3.174	3.212	3.249	3.285	3.320	3.354	3.388
33	2.745	2.801	2.845	2.888	2.929	2.969	3.008	3.045	3.081	3.117	3.153	3.188	3.221	3.254	3.286	3.318
34	2.710	2.770	2.814	2.857	2.898	2.937	2.974	3.010	3.045	3.079	3.112	3.144	3.175	3.205	3.234	3.262
35	2.675	2.741	2.785	2.827	2.868	2.908	2.946	2.983	3.019	3.053	3.086	3.118	3.149	3.179	3.208	3.236
36	2.640	2.718	2.763	2.804	2.844	2.883	2.921	2.958	2.994	3.028	3.061	3.093	3.124	3.154	3.183	3.211
37	2.605	2.684	2.729	2.769	2.808	2.846	2.883	2.919	2.954	2.988	3.021	3.053	3.084	3.114	3.143	3.171
38	2.570	2.650	2.695	2.735	2.774	2.812	2.849	2.885	2.920	2.953	2.985	3.016	3.046	3.075	3.104	3.132
39	2.535	2.615	2.660	2.700	2.738	2.775	2.812	2.848	2.883	2.916	2.948	2.979	3.009	3.038	3.066	3.094
40	2.500	2.580	2.625	2.665	2.703	2.740	2.776	2.811	2.845	2.878	2.910	2.941	2.971	3.000	3.028	3.056
41	2.465	2.545	2.590	2.630	2.667	2.704	2.740	2.775	2.808	2.840	2.871	2.901	2.930	2.958	2.986	3.014
42	2.430	2.510	2.555	2.595	2.632	2.669	2.705	2.740	2.773	2.805	2.836	2.866	2.895	2.923	2.951	2.979
43	2.395	2.475	2.520	2.560	2.597	2.634	2.670	2.705	2.738	2.769	2.799	2.828	2.856	2.884	2.912	2.940
44	2.360	2.440	2.485	2.525	2.562	2.599	2.635	2.670	2.703	2.735	2.766	2.795	2.823	2.851	2.879	2.907
45	2.325	2.405	2.450	2.490	2.527	2.564	2.600	2.635	2.668	2.699	2.729	2.758	2.786	2.814	2.842	2.870
46	2.290	2.370	2.415	2.455	2.492	2.529	2.564	2.600	2.633	2.664	2.694	2.723	2.751	2.779	2.807	2.835
47	2.255	2.335	2.380	2.420	2.457	2.494	2.529	2.564	2.597	2.628	2.657	2.686	2.714	2.742	2.769	2.797
48	2.220	2.300	2.345	2.385	2.422	2.459	2.494	2.529	2.562	2.593	2.622	2.650	2.678	2.705	2.732	2.759
49	2.185	2.265	2.310	2.350	2.387	2.424	2.459	2.494	2.527	2.557	2.586	2.614	2.641	2.668	2.695	2.722
50	2.150	2.230	2.275	2.315	2.352	2.389	2.424	2.459	2.492	2.522	2.551	2.579	2.606	2.633	2.660	2.687
51	2.115	2.195	2.240	2.280	2.317	2.354	2.389	2.424	2.457	2.487	2.515	2.542	2.569	2.596	2.623	2.650
52	2.080	2.160	2.205	2.245	2.282	2.319	2.354	2.389	2.422	2.452	2.480	2.507	2.534	2.561	2.588	2.615
53	2.045	2.125	2.170	2.210	2.247	2.284	2.319	2.354	2.387	2.417	2.444	2.471	2.498	2.525	2.552	2.579
54	2.010	2.090	2.135	2.175	2.212	2.249	2.284	2.319	2.352	2.382	2.409	2.435	2.461	2.487	2.513	2.539
55	1.975	2.055	2.100	2.140	2.177	2.214	2.249	2.282	2.312	2.341	2.367	2.393	2.418	2.443	2.468	2.493
56	1.940	2.020	2.065	2.105	2.142	2.179	2.214	2.247	2.277	2.306	2.332	2.357	2.382	2.407	2.432	2.457
57	1.905	1.985	2.030	2.070	2.107	2.144	2.179	2.212	2.242	2.271	2.297	2.322	2.347	2.372	2.397	2.422
58	1.870	1.950	1.995	2.035	2.072	2.109	2.144	2.177	2.207	2.236	2.262	2.287	2.312	2.337	2.362	2.387
59	1.835	1.915	1.960	2.000	2.037	2.074	2.109	2.142	2.172	2.201	2.227	2.252	2.277	2.302	2.327	2.352
60	1.800	1.880	1.925	1.965	2.002	2.039	2.074	2.107	2.137	2.166	2.191	2.216	2.241	2.266	2.291	2.316
61	1.765	1.845	1.890	1.930	1.967	2.004	2.039	2.072	2.102	2.131	2.156	2.181	2.206	2.231	2.256	2.281
62	1.730	1.810	1.855	1.895	1.932	1.969	2.004	2.037	2.067	2.096	2.121	2.146	2.171	2.196	2.221	2.246
63	1.695	1.775	1.820	1.860	1.897	1.934	1.969	2.002	2.032	2.061	2.086	2.111	2.136	2.161	2.186	2.211
64	1.660	1.740	1.785	1.825	1.862	1.899	1.934	1.967	1.997	2.026	2.051	2.076	2.101	2.126	2.151	2.176
65	1.625	1.705	1.750	1.790	1.827	1.864	1.899	1.932	1.962	1.991	2.016	2.041	2.066	2.091	2.116	2.141
66	1.590	1.670	1.715	1.755	1.792	1.829	1.864	1.897	1.927	1.956	1.981	2.006	2.031	2.056	2.081	2.106
67	1.555	1.635	1.680	1.720	1.757	1.794	1.829	1.862	1.892	1.921	1.946	1.971	1.996	2.021	2.046	2.071
68	1.520	1.600	1.645	1.685	1.722	1.759	1.794	1.827	1.857	1.886	1.911	1.936	1.961	1.986	2.011	2.036
69	1.485	1.565	1.610	1.650	1.687	1.724	1.759	1.792	1.822	1.851	1.876	1.901	1.926	1.951	1.976	2.001
70	1.450	1.530	1.575	1.615	1.652	1.689	1.724	1.757	1.787	1.816	1.841	1.866	1.891	1.916	1.941	1.966
71	1.415	1.495	1.540	1.580	1.617	1.654	1.689	1.722	1.752	1.781	1.806	1.831	1.856	1.881	1.906	1.931
72	1.380	1.460	1.505	1.545	1.582	1.619	1.654	1.687	1.717	1.746	1.771	1.796	1.821	1.846	1.871	1.896
73	1.345	1.425	1.470	1.510	1.547	1.584	1.619	1.652	1.682	1.711	1.736	1.761	1.786	1.811	1.836	1.861
74	1.310	1.390	1.435	1.475	1.512	1.549	1.584	1.617	1.647	1.676	1.701	1.726	1.751	1.776	1.801	1.826
75	1.275	1.355	1.400	1.440	1.477	1.514	1.549	1.582	1.612	1.641	1.666	1.691	1.716	1.741	1.766	1.791
76	1.240	1.320	1.365	1.405	1.442	1.479	1.514	1.547	1.577	1.606	1.631	1.656	1.681	1.706	1.731	1.756
77	1.205	1.285	1.330	1.370	1.407	1.444	1.479	1.512	1.542	1.571	1.596	1.621	1.646	1.671	1.696	1.721
78	1.170	1.250	1.295	1.335	1.372	1.409	1.444	1.477	1.507	1.536	1.561	1.586	1.611	1.636	1.661	1.686
79	1.135	1.215	1.260	1.300	1.337	1.374	1.409	1.442	1.472	1.501	1.526	1.551	1.576	1.601	1.626	1.651
80	1.100	1.180	1.225	1.265	1.302	1.339	1.374	1.407	1.437	1.466	1.491	1.516	1.541	1.566	1.591	1.616
81	1.065	1.145	1.190	1.230	1.267	1.304	1.339	1.372	1.402	1.431	1.456	1.481	1.506	1.531	1.556	1.581
82	1.030	1.110	1.155	1.195	1.232	1.269	1.304	1.337	1.367	1.396	1.421	1.446	1.471	1.496	1.521	1.546
83	0.995	1.075	1.120	1.160	1.197	1.234	1.269	1.302	1.332	1.361	1.386	1.411	1.436	1.461	1.486	1.511
84	0.960	1.040	1.085	1.125	1.162	1.199	1.234	1.267	1.297	1.326	1.351	1.376	1.401	1.426	1.451	1.476
85	0.925	1.005	1.050	1.090	1.127	1.164	1.199	1.232	1.262	1.291	1.316	1.341	1.366	1.391	1.416	1.441
86	0.890	0.970	1.015	1.055	1.092	1.129	1.164	1.197	1.227	1.256	1.281	1.306	1.331	1.356	1.381	1.406
87	0.855	0.935	0.980	1.020	1.057	1.094	1.129	1.162	1.192	1.221	1.246	1.271	1.296	1.321	1.346	1.371
88	0.820	0.900	0.945	0.985	1.022	1.059	1.094	1.127	1.157	1.186	1.211	1.236	1.261	1.286	1.311	1.336
89	0.785	0.865	0.910	0.950	0.987	1.024	1.059	1.092	1.122	1.151	1.176	1.201	1.226	1.251	1.276	1.301
90	0.750	0.830	0.875	0.915	0.952	0.989	1.024	1.057	1.087	1.116	1.141	1.166	1.191	1.216	1.241	1.266
91	0.715	0.795	0.840	0.880	0.917	0.954	0.989	1.022	1.052	1.081	1.106	1.131	1.156	1.181	1.206	1.231
92	0.680	0.760	0.805	0.845	0.882	0.919	0.954	0.987	1.017	1.046	1.071	1.096	1.121	1.146	1.171	1.196
93	0.645	0.725	0.770	0.810	0.847	0.884	0.919	0.952	0.982	1.011	1.036	1.061	1.086	1.111	1.136	1.161
94	0.610	0.690	0.735	0.775	0.812	0.849	0.884	0.917	0.947	0.976	1.001					

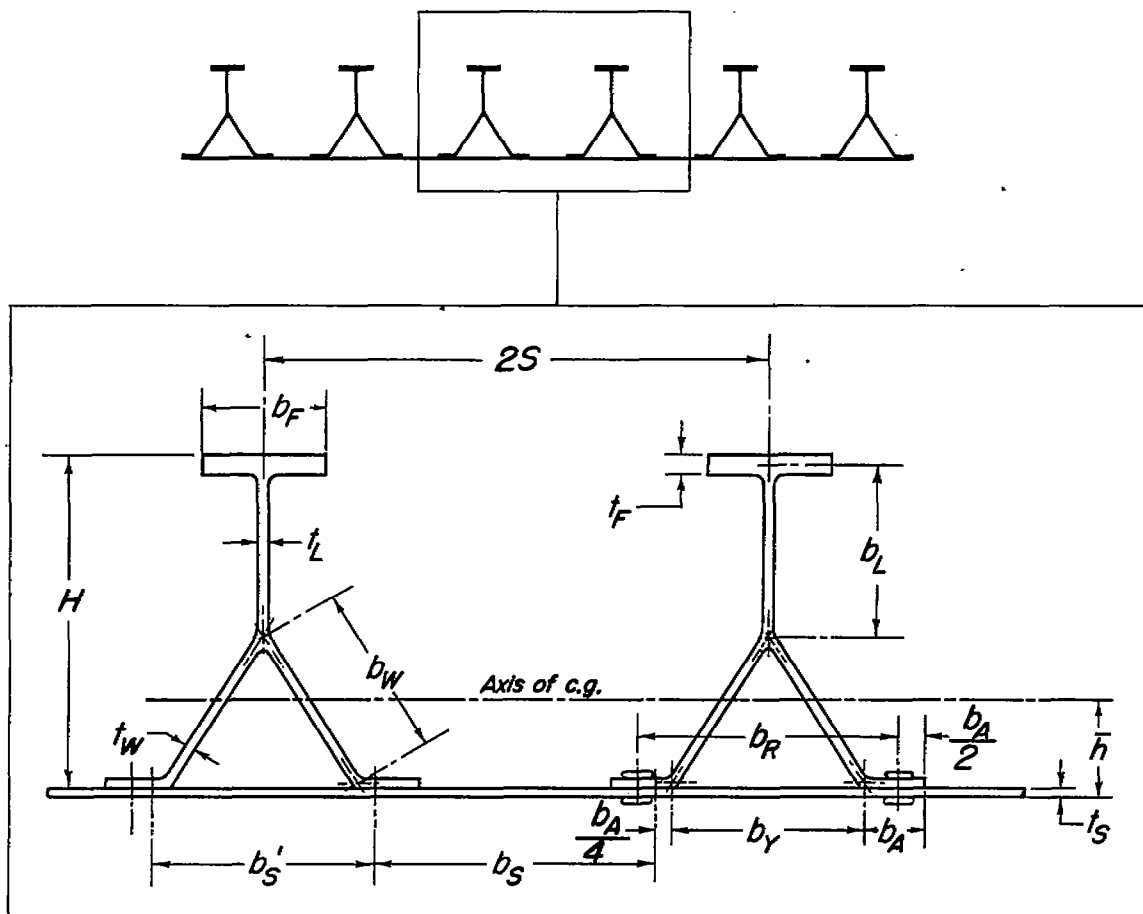
TABLE 14  
VALUES AND COMPUTATIONS FOR OBTAINING DESIGN FOR MAXIMUM STRUCTURAL EFFICIENCY

$$[P_1 = 5.0 \text{ kips/in.}; L = 20 \text{ in.}; c = 1; t_s = 0.102 \text{ in.}; \frac{t_w}{t_s} = 0.40]$$

Step (1)		Step (3)			Step (4)	Step (5)			Step (8)			Step (9)		Step (10)			Step (12)
$\frac{P_1}{L/\sqrt{c}}$ (ksi)	$\frac{P_1}{t_s}$ (ksi)	$\frac{b_w}{t_w}$	$\frac{b_s}{t_s}$	$\bar{\sigma}_f$ (ksi)	$\frac{A_1}{t_s}$	$\frac{P_1}{\bar{\sigma}_f t_s}$	$\frac{b_s}{t_s}$	$\frac{b_w}{t_w}$	$\bar{\sigma}_f$ (ksi)	$\frac{A_1}{t_s}$	$\frac{P_1}{t_s}$ (kips/in.)	$\frac{t_w}{t_s}$	$\frac{P_1}{t_s}$ (kips/in.)	$t_w$ (in.)	$b_s$ (in.)	$b_w$ (in.)	$\sigma_{cr}$ (ksi)
For $t_s = 0.102 \text{ in.}$																	
0.25	49	18	23	31.2	1.496	0.1071	36.6	25.7	34.2	1.439	5.02	0.040	5.02	0.040	3.732	1.028	31.5
			26	31.5	1.454	.1090											
			30	31.6	1.407	.1123											
			35	31.2	1.361	.1177											
			42	29.8	1.312	.1278											
			50	27.8	1.270	.1415											
			60	25.7	1.231	.1580											
			72	22.5	1.196	.1860											
			84	19.2	1.171	.2222											
		21	23	33.3	1.542	.0973											
			26	33.6	1.497	.0995											
			30	33.5	1.448	.1029											
			35	32.9	1.399	.1086											
			42	31.5	1.345	.1179											
			50	29.4	1.300	.1318											
			60	26.7	1.257	.1489											
			72	23.4	1.220	.1750											
			84	19.7	1.192	.2127											
		24	23	35.1	1.585	.0899											
			26	35.4	1.538	.0919											
			30	35.0	1.486	.0962											
			35	34.1	1.434	.1022											
			42	32.3	1.378	.1123											
			50	30.1	1.328	.1250											
			60	27.2	1.283	.1432											
			72	24.0	1.242	.1677											
			84	20.0	1.212	.2065											
		27	23	36.5	1.624	.0843											
			26	36.5	1.576	.0870											
			30	35.9	1.522	.0915											
			35	34.8	1.468	.0978											
			42	32.8	1.408	.1082											
			50	30.2	1.356	.1221											
			60	27.4	1.307	.1397											
			72	24.0	1.264	.1647											
			84	20.2	1.231	.2010											
		30	23	37.7	1.661	.0798											
			26	37.2	1.612	.0834											
			30	36.4	1.556	.0881											
			35	35.0	1.500	.0952											
			42	32.6	1.437	.1067											
			50	30.2	1.383	.1195											
			60	27.1	1.331	.1385											
			72	23.7	1.285	.1642											
			84	20.2	1.250	.1980											
		33	23	39.0	1.696	.0756											
			26	37.3	1.646	.0814											
			30	36.2	1.589	.0870											
			35	34.7	1.530	.0941											
			42	32.3	1.465	.1060											
			50	29.8	1.408	.1191											
			60	26.7	1.354	.1381											
			72	23.4	1.306	.1656											
			84	20.3	1.269	.1942											

\*See appendix A for discussion of steps.

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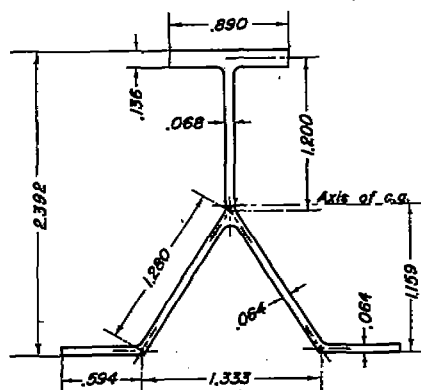
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*r = All fillet radii*  
*d = Rivet diameter*  
*p = Rivet pitch*  
*L = Length of panel*

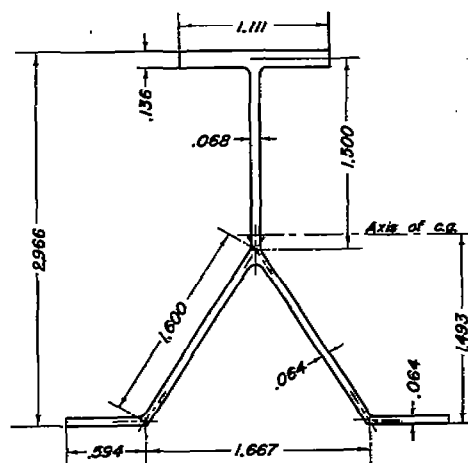
Figure 1. - Symbols for panel dimensions.

$\frac{b_w}{t_w}$ 

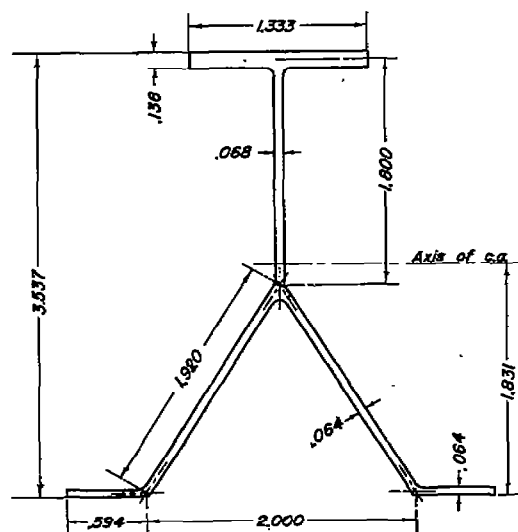
20


 $A=0.446 \text{ sq in.}$ 
 $\rho=0.907 \text{ in.}$ 

25


 $A=0.538 \text{ sq in.}$ 
 $\rho=1.123 \text{ in.}$ 

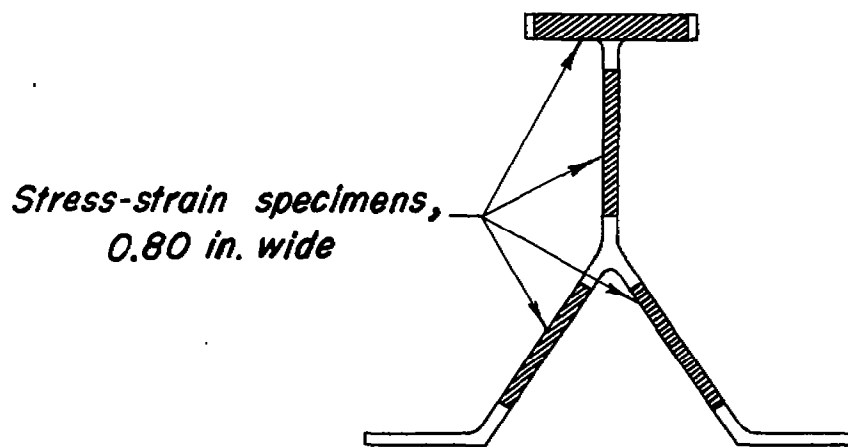
30


 $A=0.629 \text{ sq in.}$ 
 $\rho=1.336 \text{ in.}$ 

All fillet radii = 0.064

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Figure 2.- Dimensions of extrusions used for test specimens.



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*Figure 3.- Locations from which stress-strain  
specimens were cut from Y-section extrusions.*

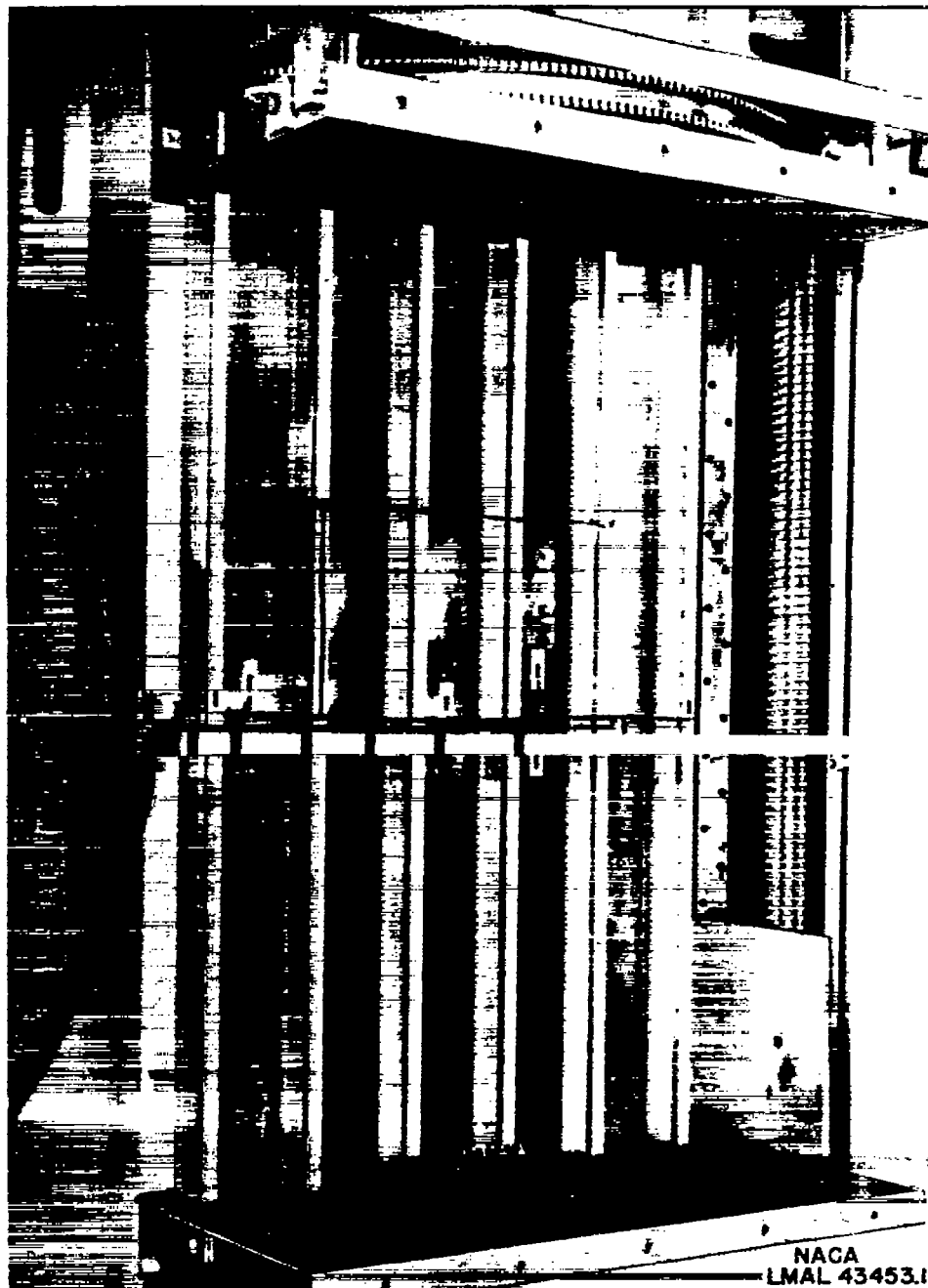


Figure 4.- Test specimen in testing machine.



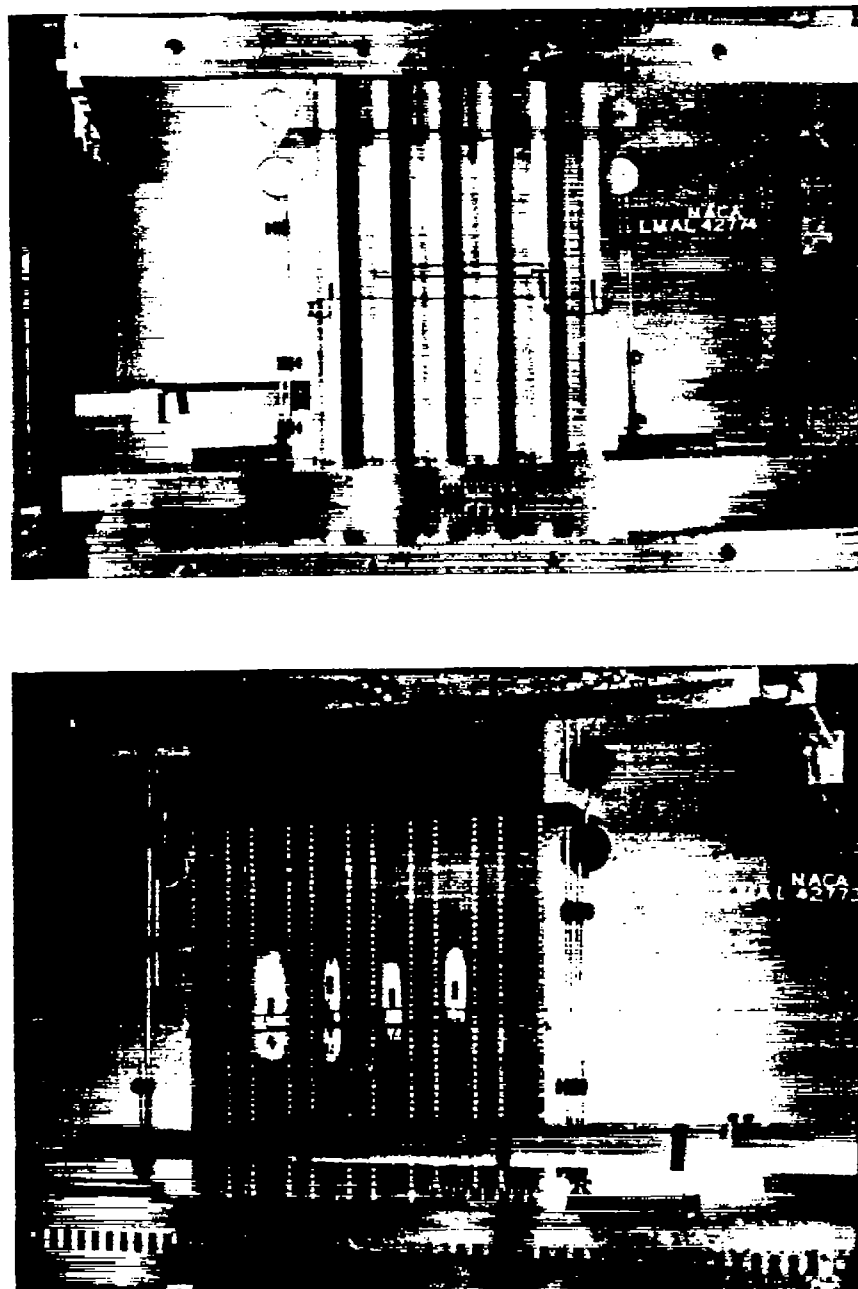


Figure 5.- Test specimen in testing machine with special bearing block to permit gage wires to be introduced inside stiffeners.

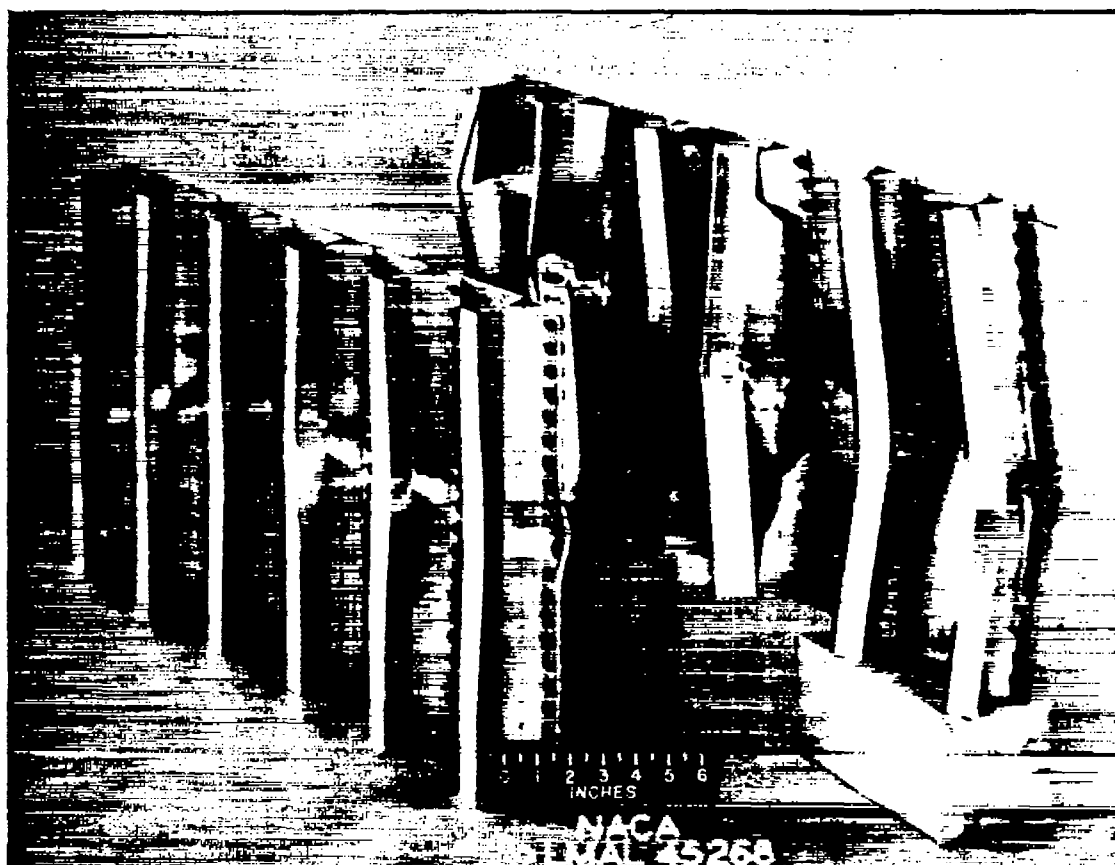


Figure 6.- A 24S-T aluminum-alloy Y-stiffened panel (on the left) and its 75S-T counterpart after failure.

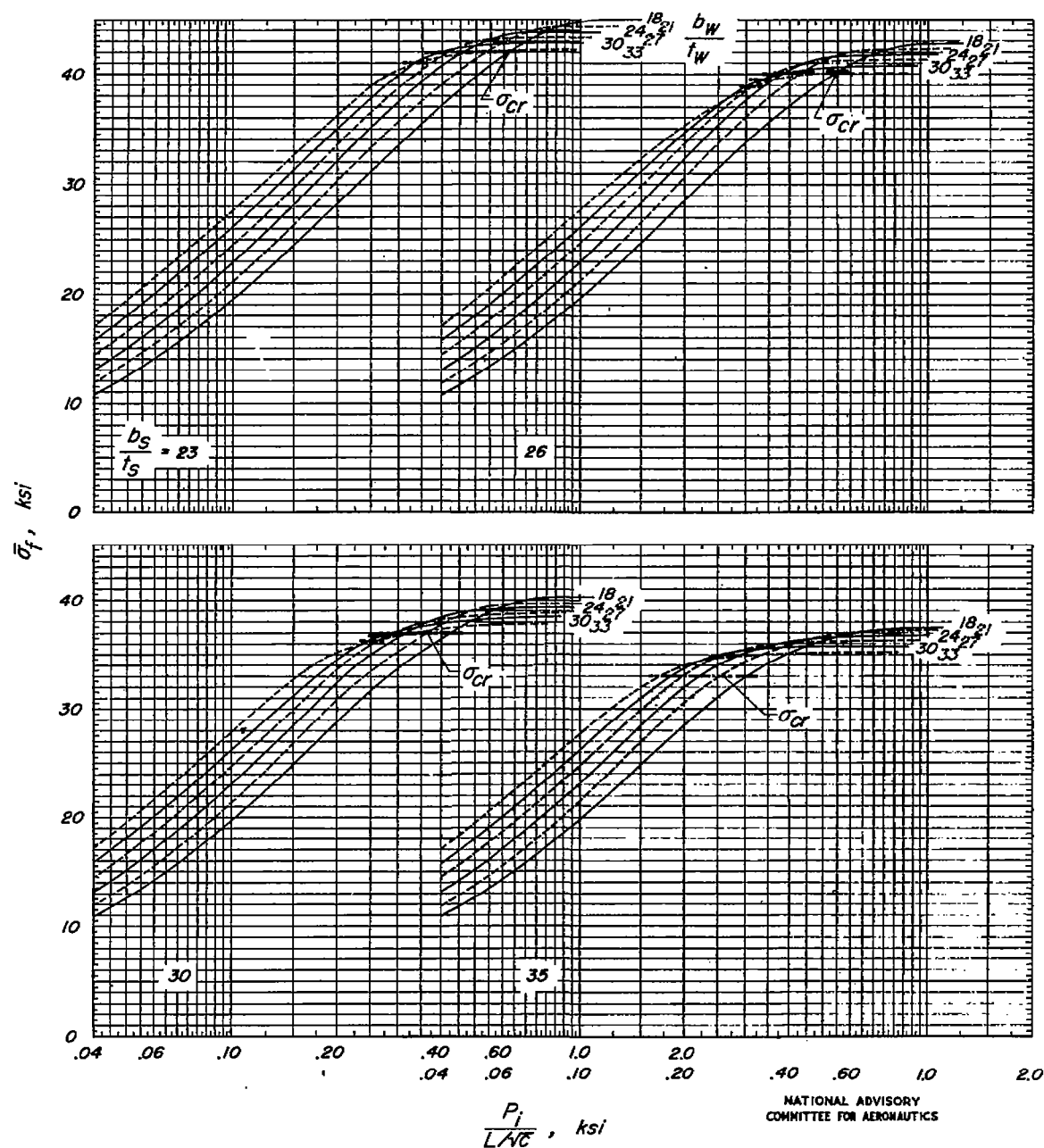


Figure 7.- Design chart for 24S-T Y-panels of the proportions tested.  $\frac{t_w}{t_s} = 0.40$ .

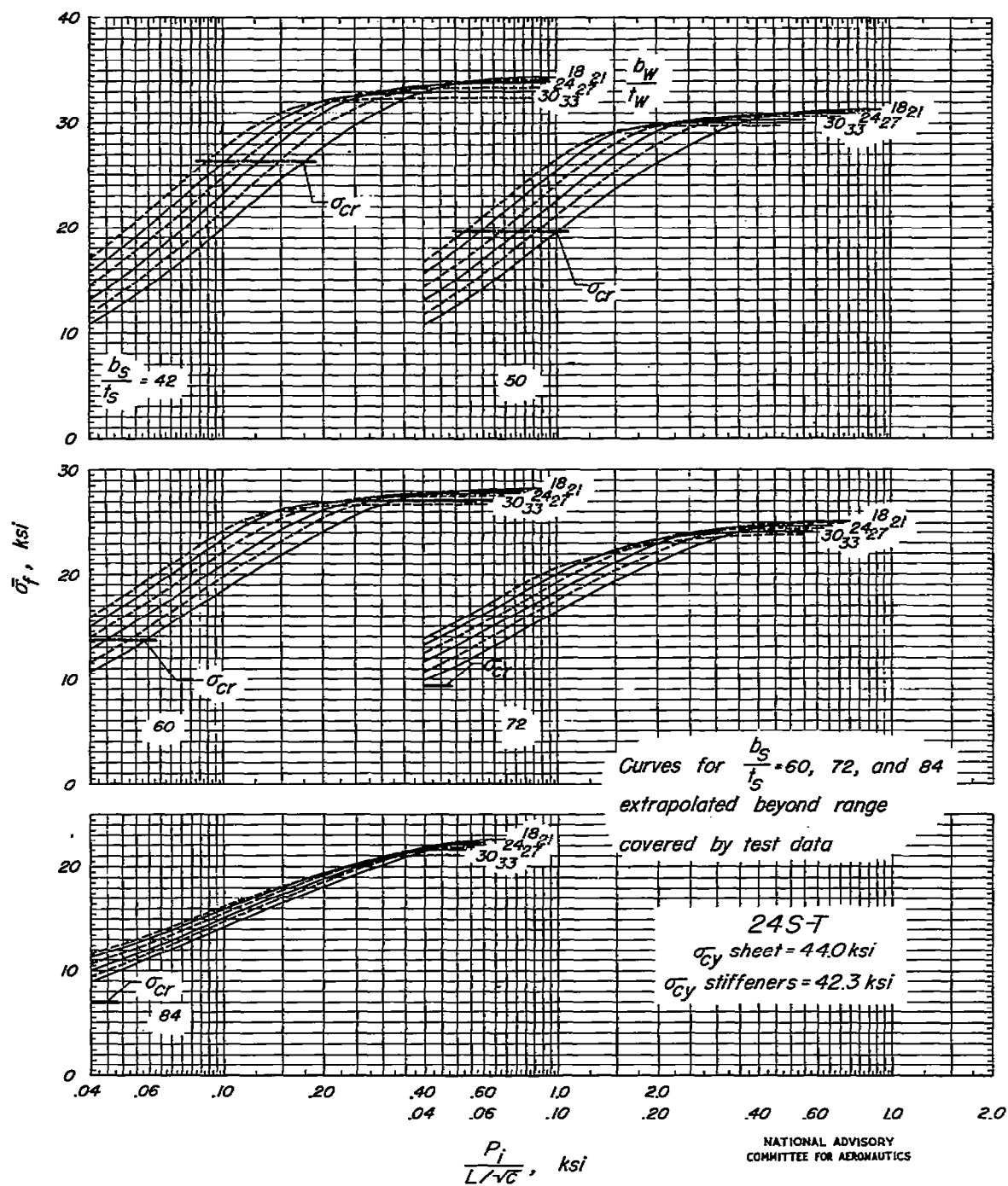


Figure 7.- Concluded.

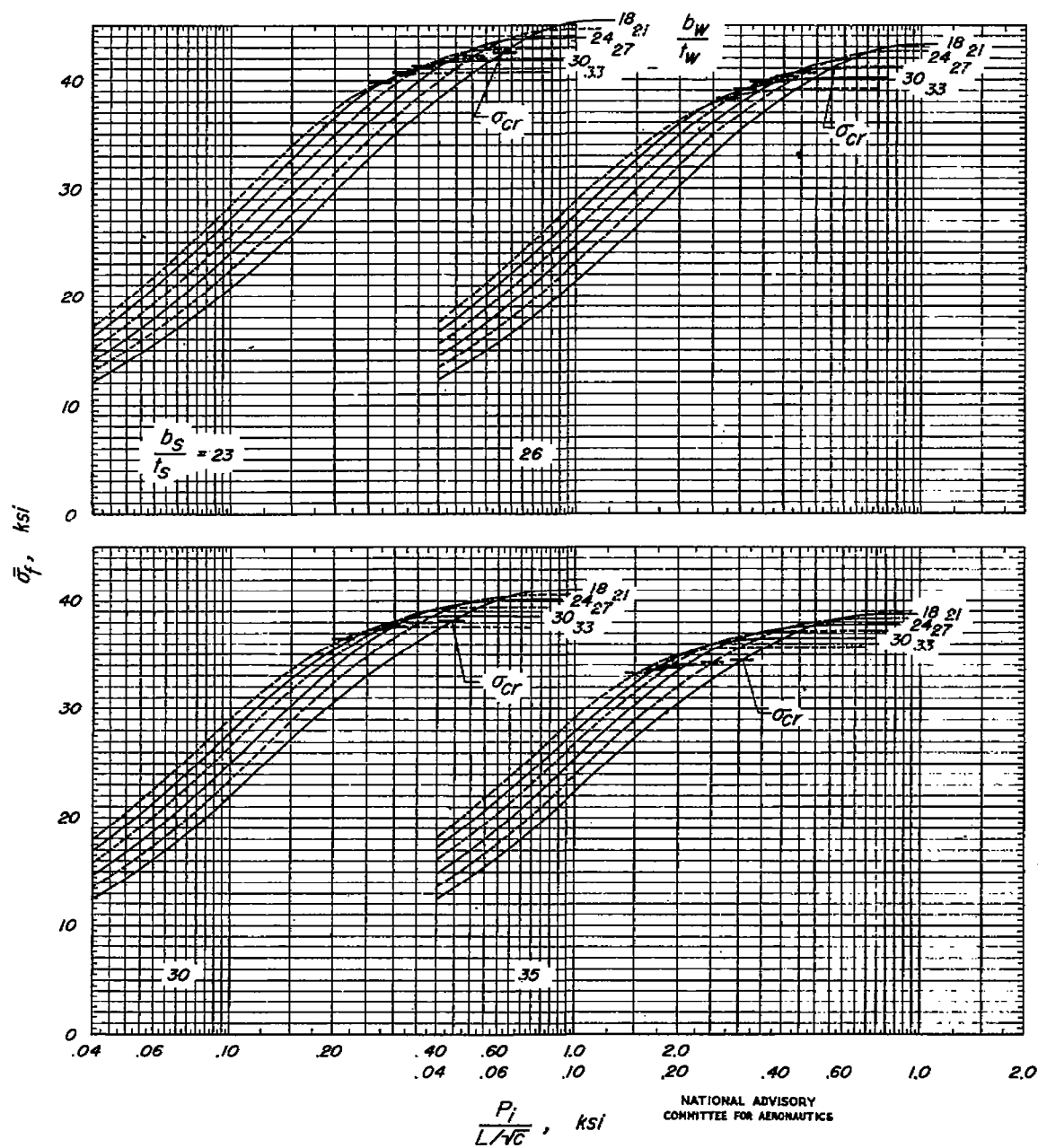


Figure 8.- Design chart for 24S-T Y-panels of the proportions tested.  $\frac{t_w}{t_s} = 0.51$ .

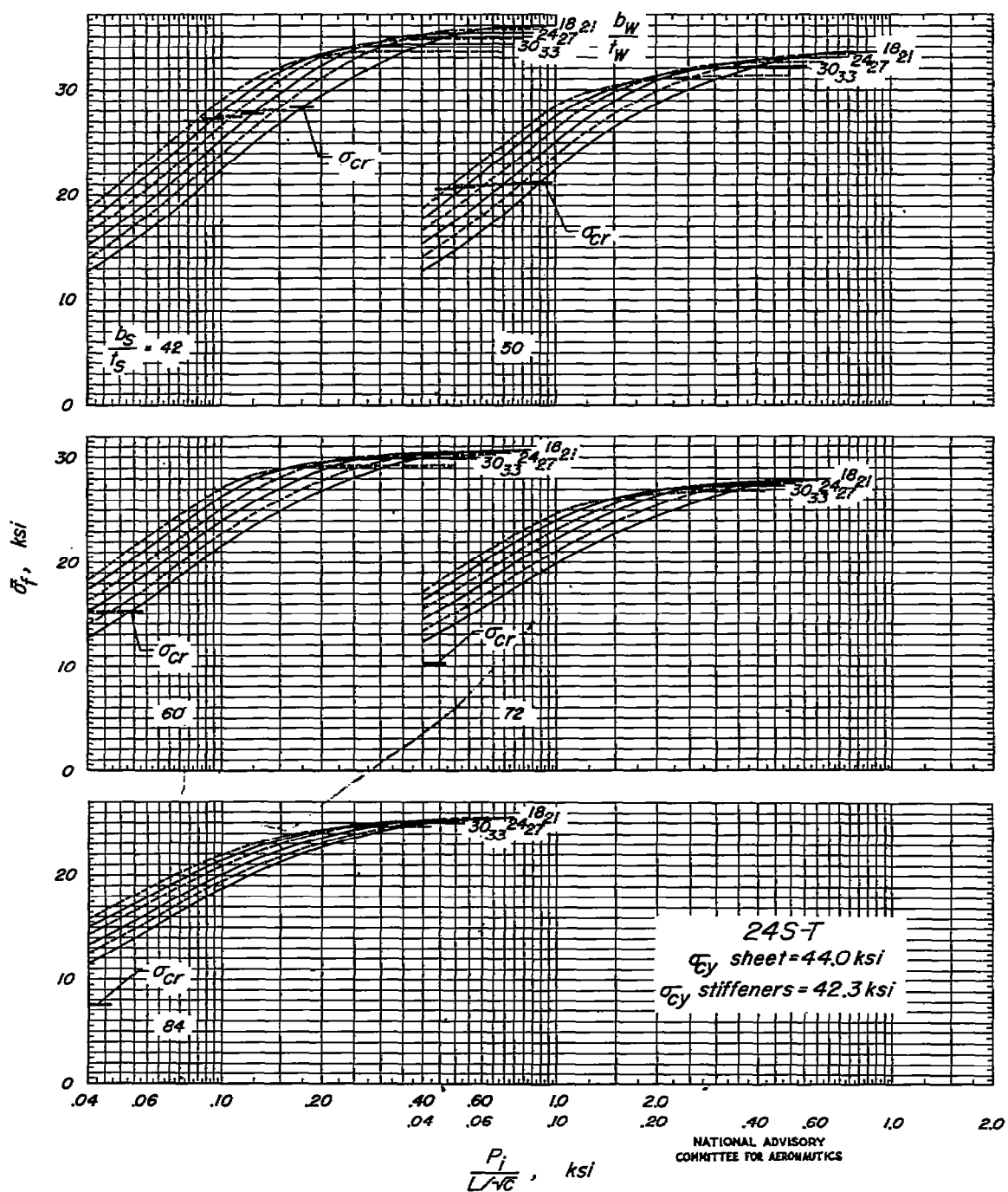


Figure 8.- Concluded.

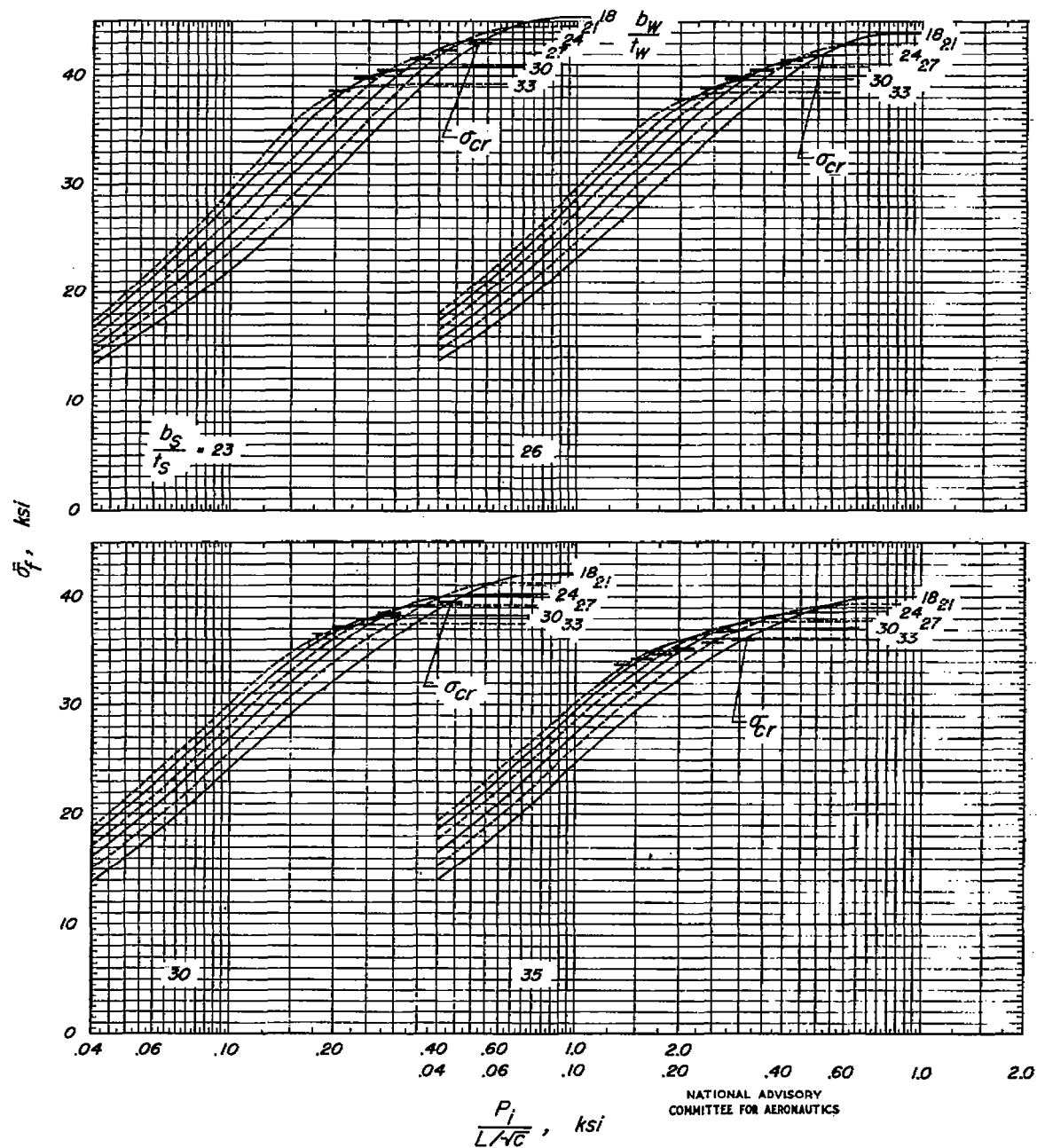


Figure 9.- Design chart for 24S-T Y-panels of the proportions tested.  $\frac{t_w}{t_s} = 0.63$ .

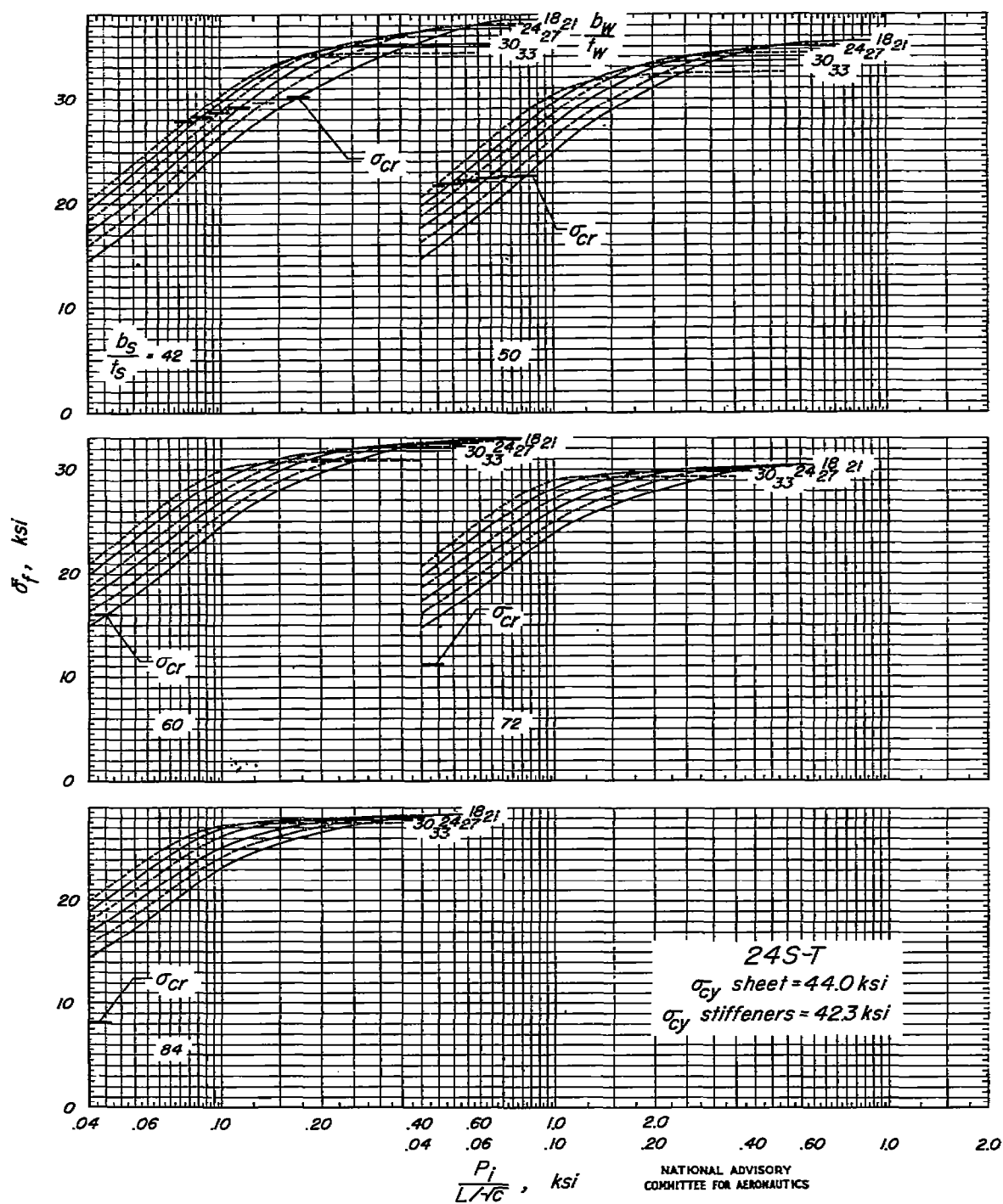


Figure 9.- Concluded.



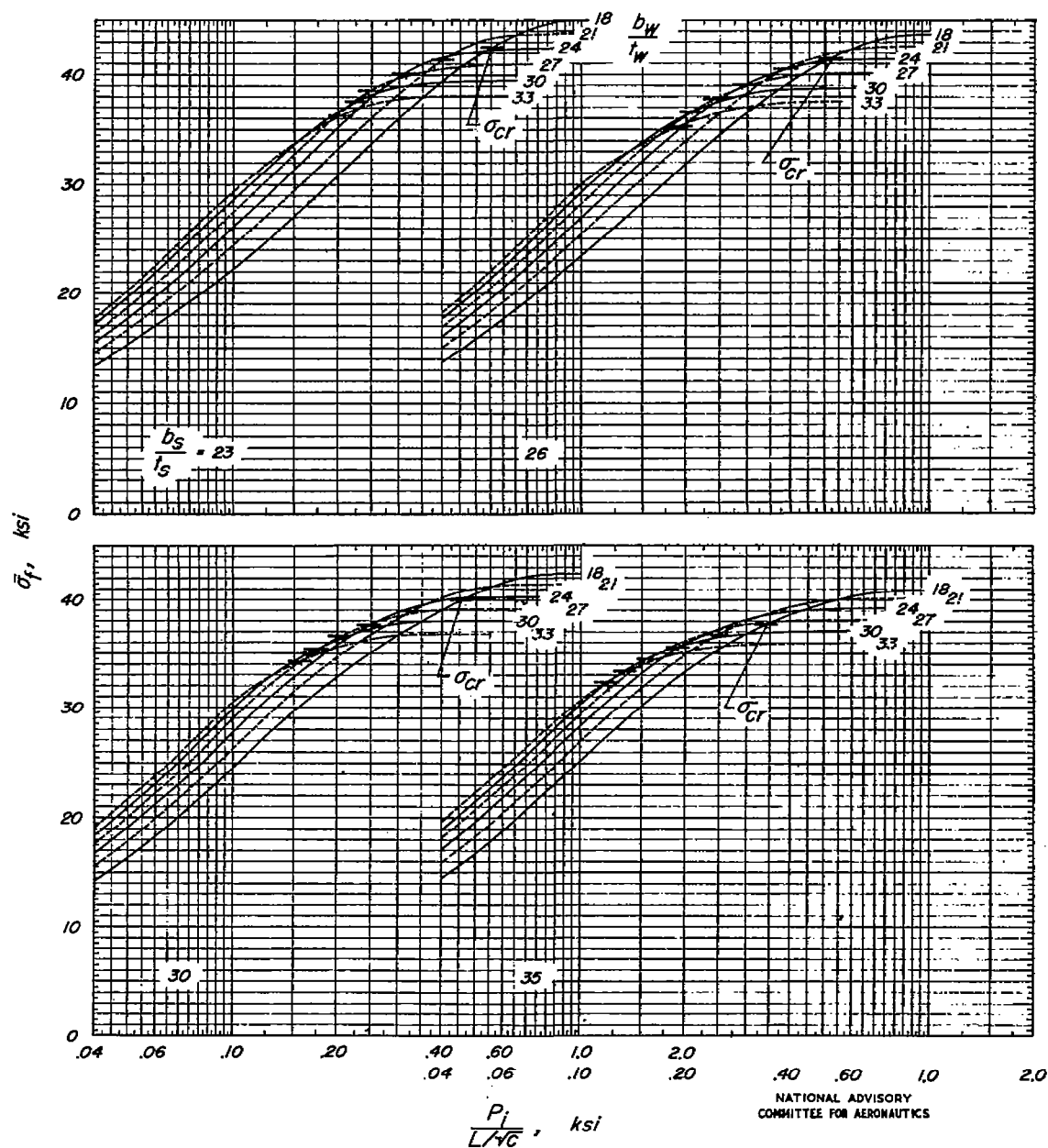


Figure 10.- Design chart for 24S-T Y-panels of the proportions tested.  $\frac{t_w}{t_s} = 0.79$ .

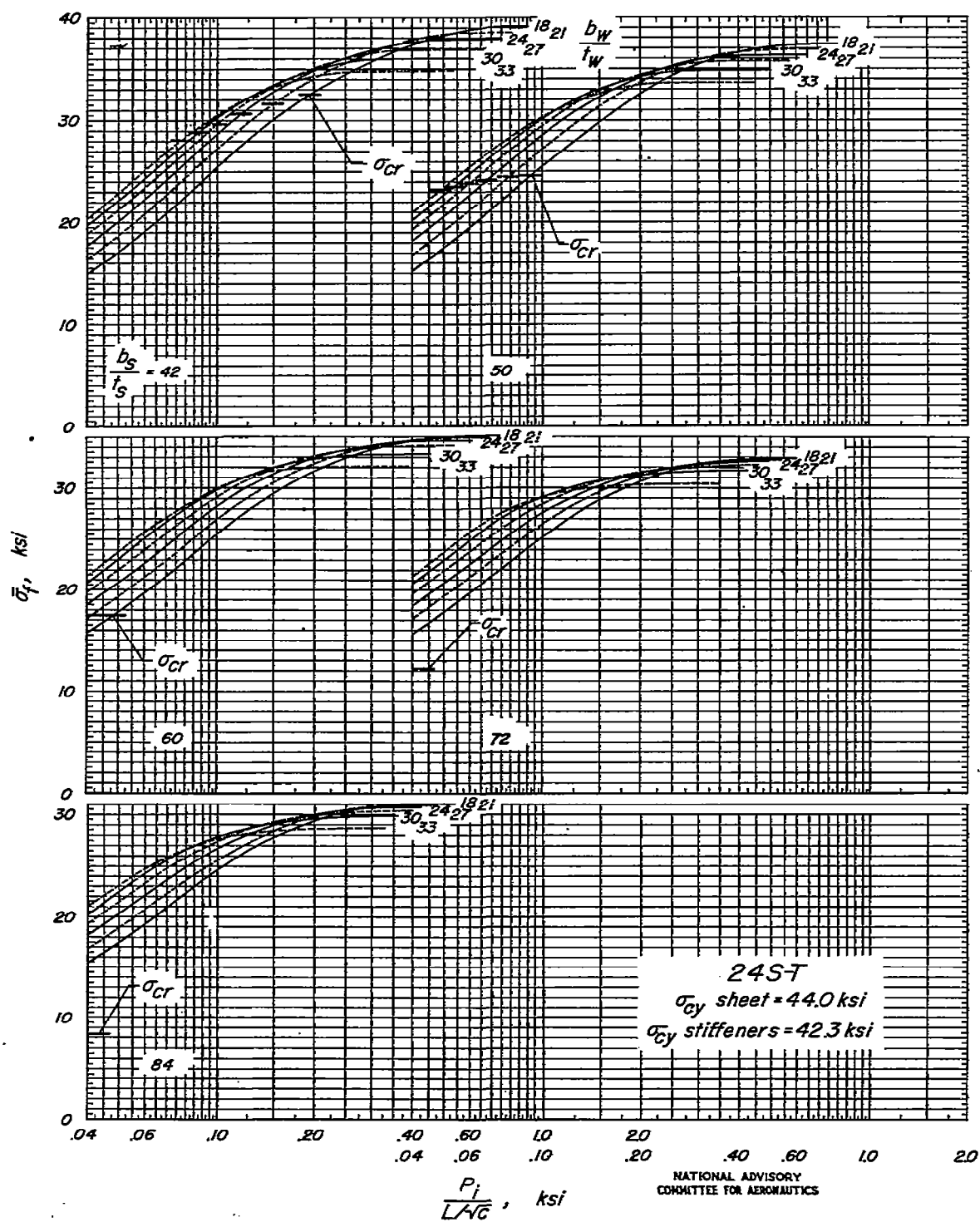


Figure 10. - Concluded.

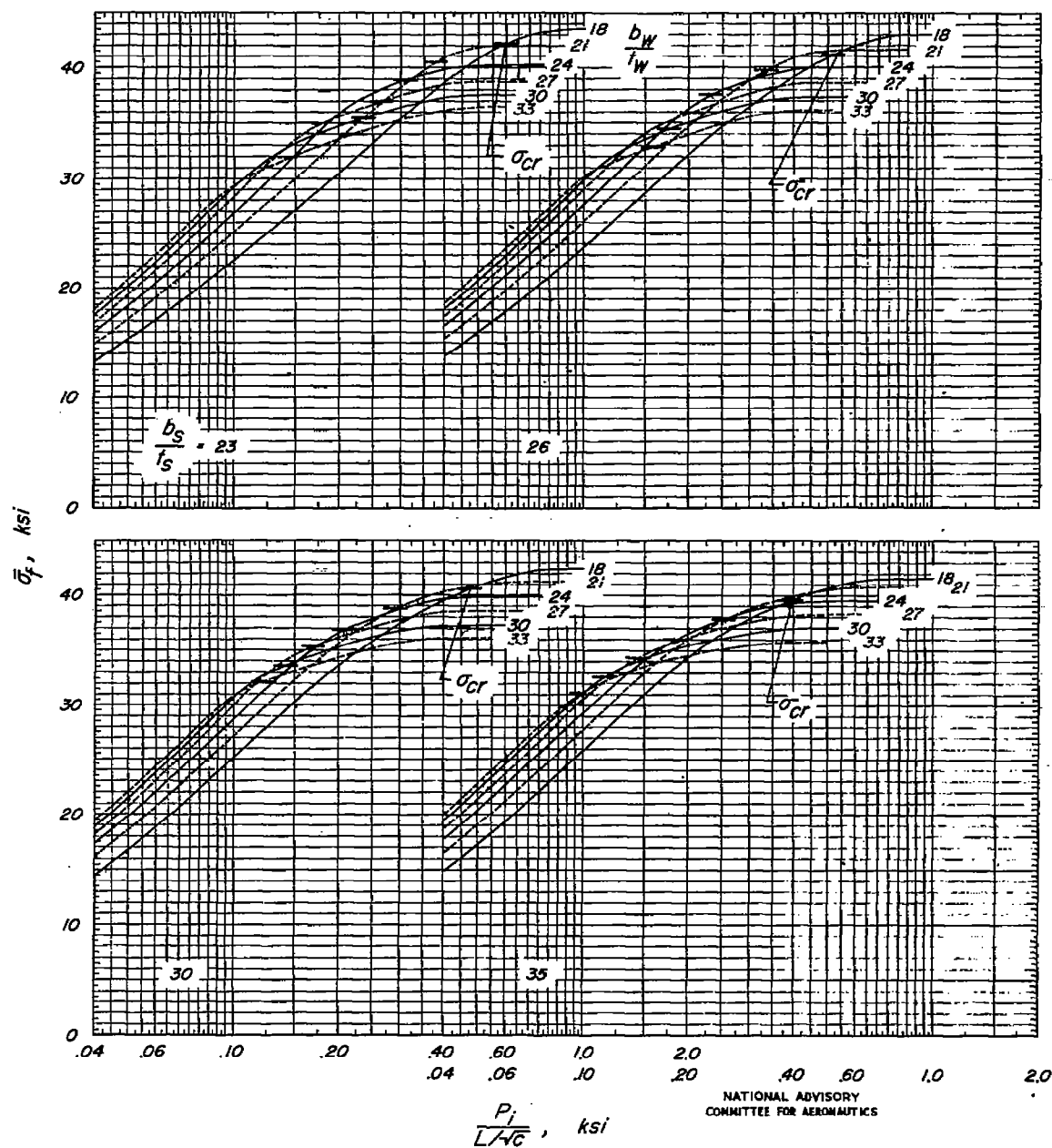


Figure 11.- Design chart for 24S-T Y-panels of the proportions tested.  $\frac{t_W}{t_S} = 1.00$ .

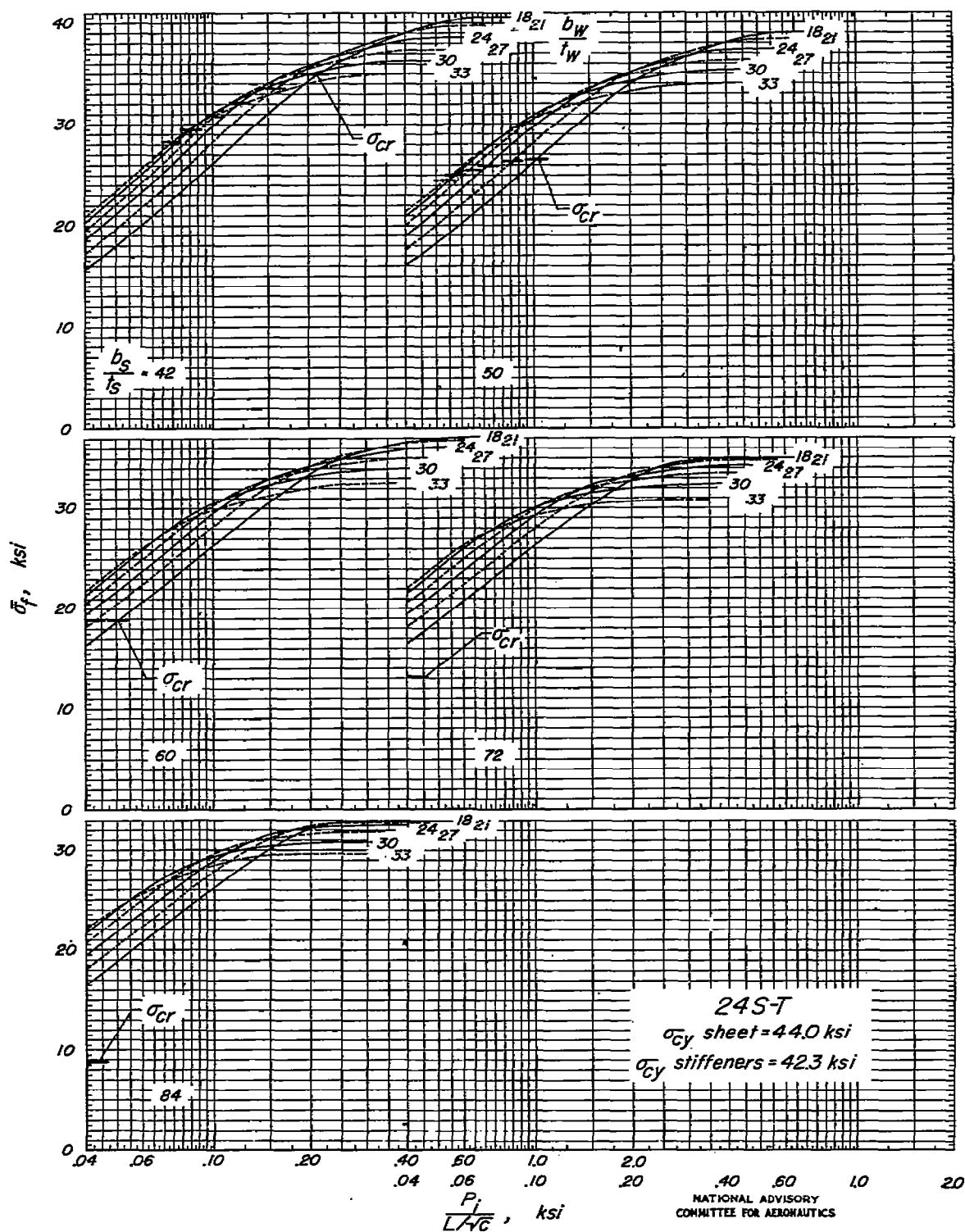


Figure 11.- Concluded.

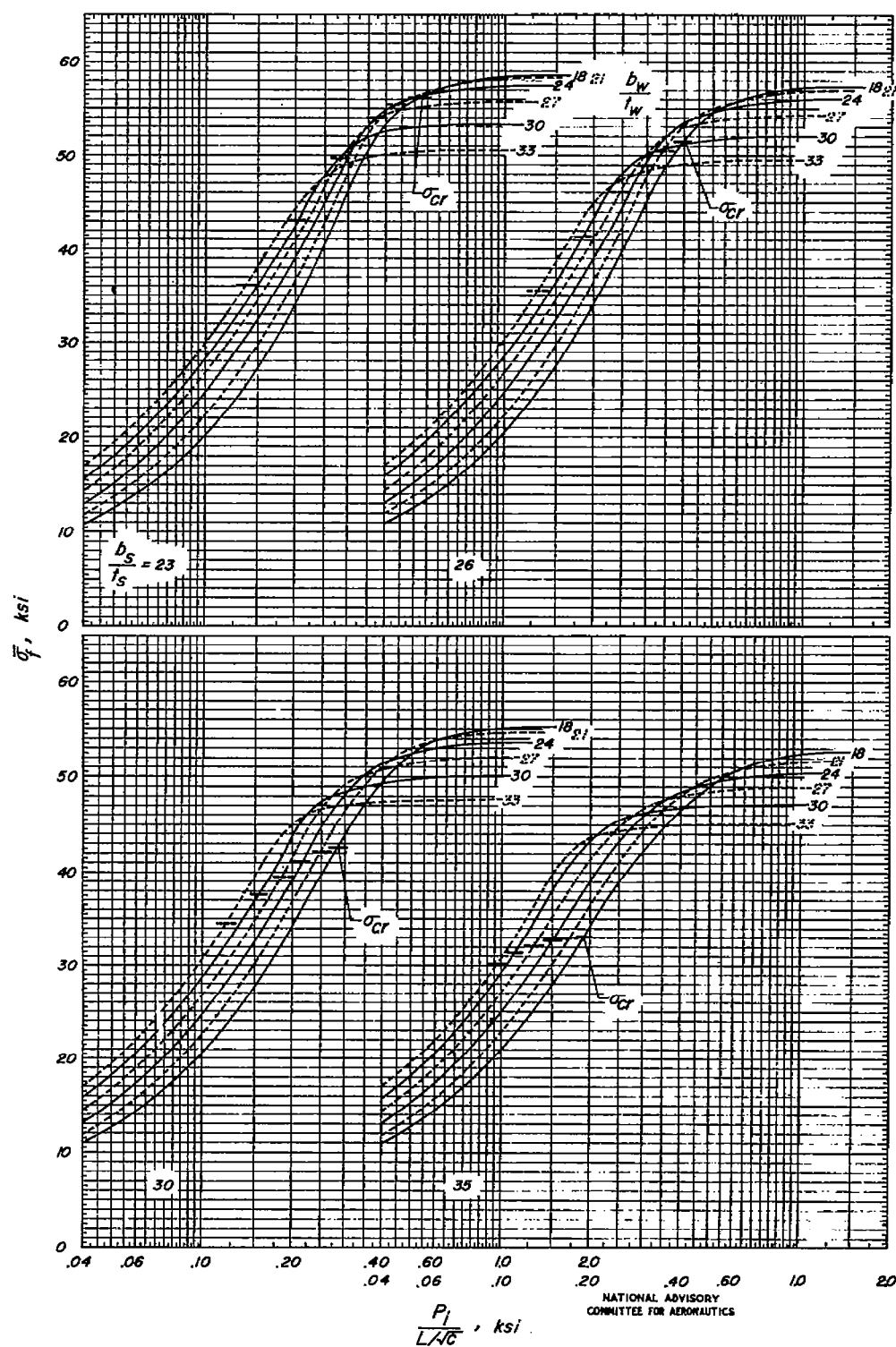


Figure 12.- Design chart for 75S-T Y-panels of the proportions tested,  $\frac{t_w}{t_s} = 0.40$ .

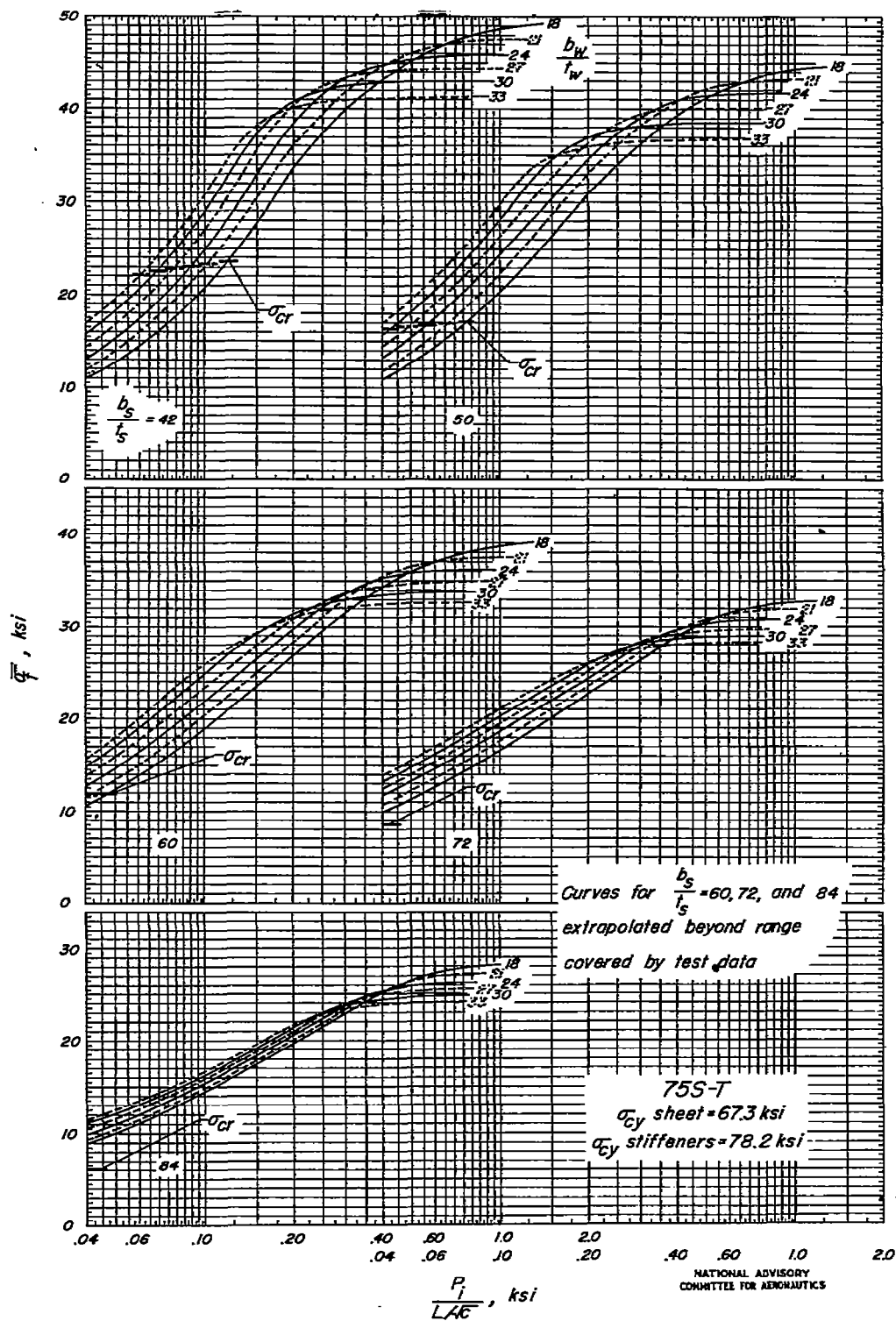


Figure 12.-Concluded.

Fig. 13

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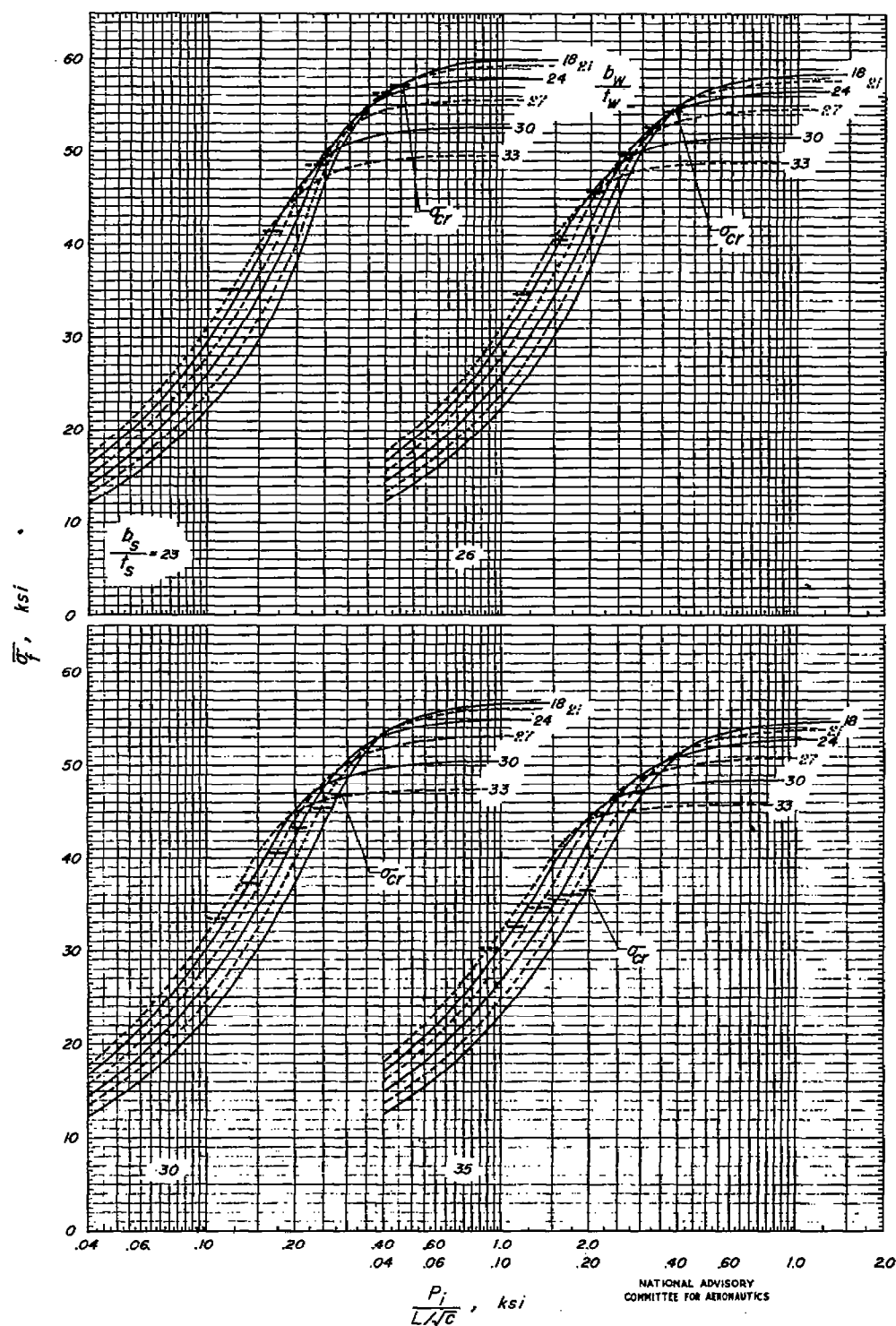


Figure 13.- Design chart for 75S-T Y-panels of the proportions tested,  $\frac{t_w}{t_s} = 0.51$ .

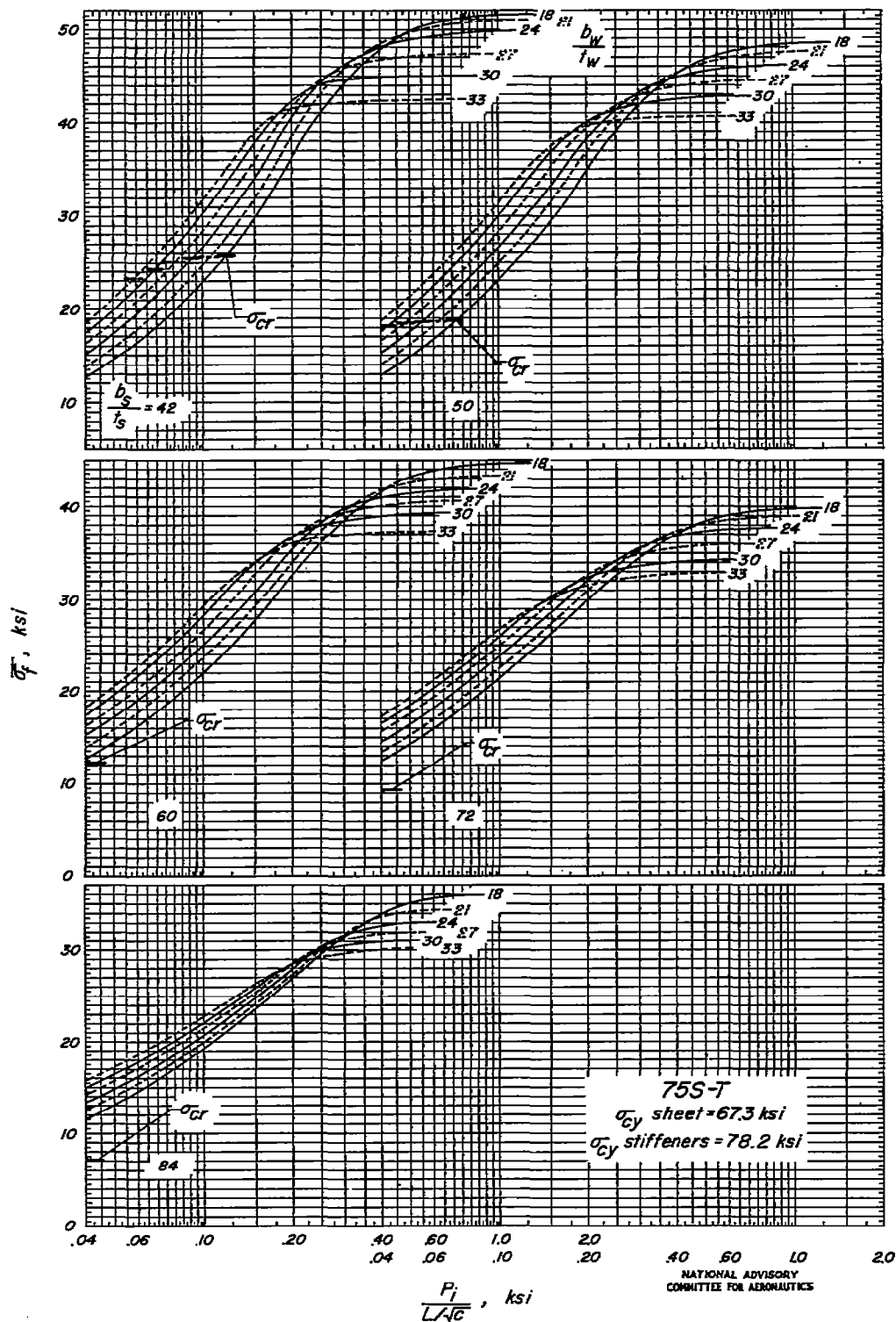


Figure 13.—Concluded.



Fig. 14

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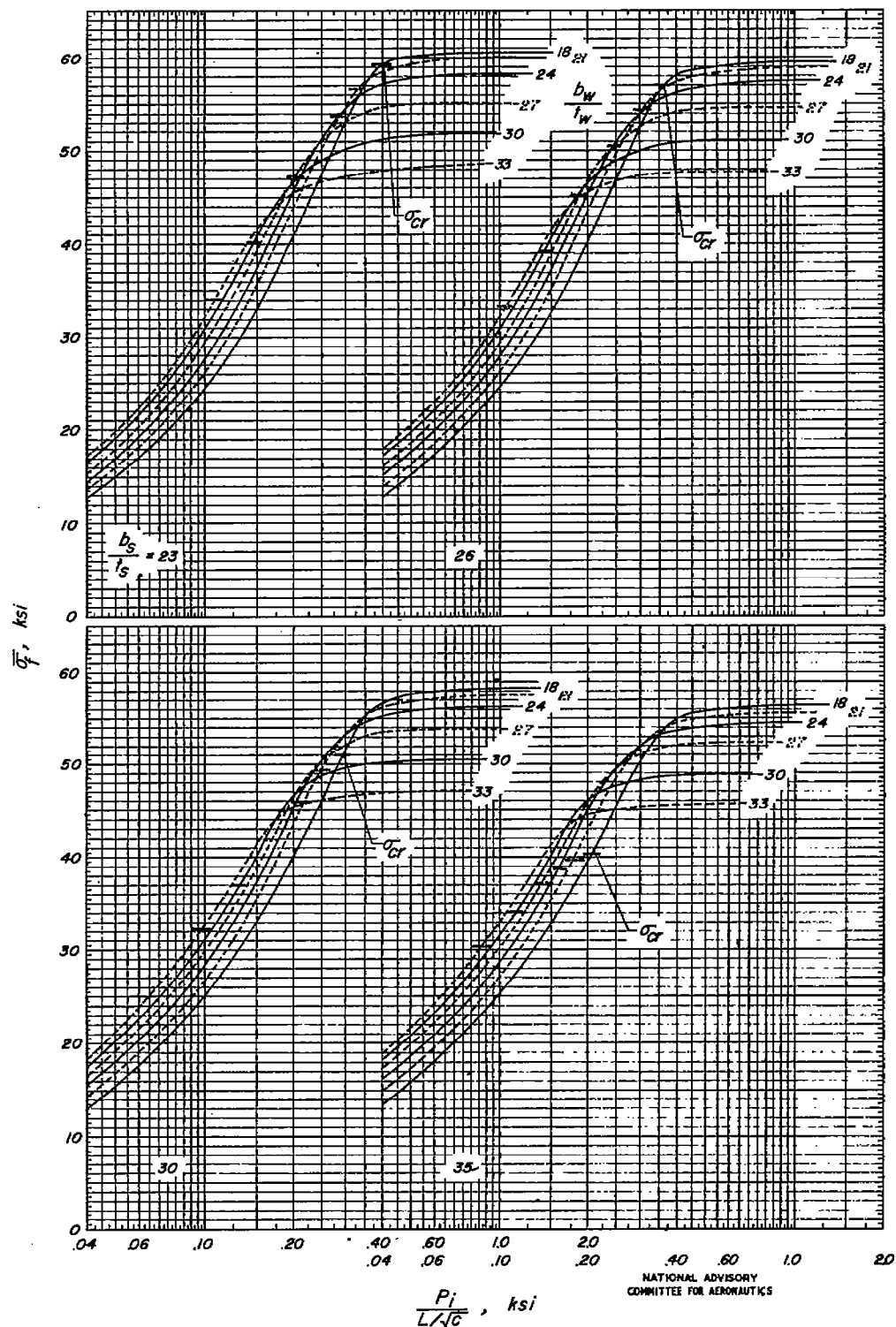


Figure 14.-Design chart for 75S-T Y-panels of the proportions tested.  $\frac{t_w}{t_s} = 0.63$ .

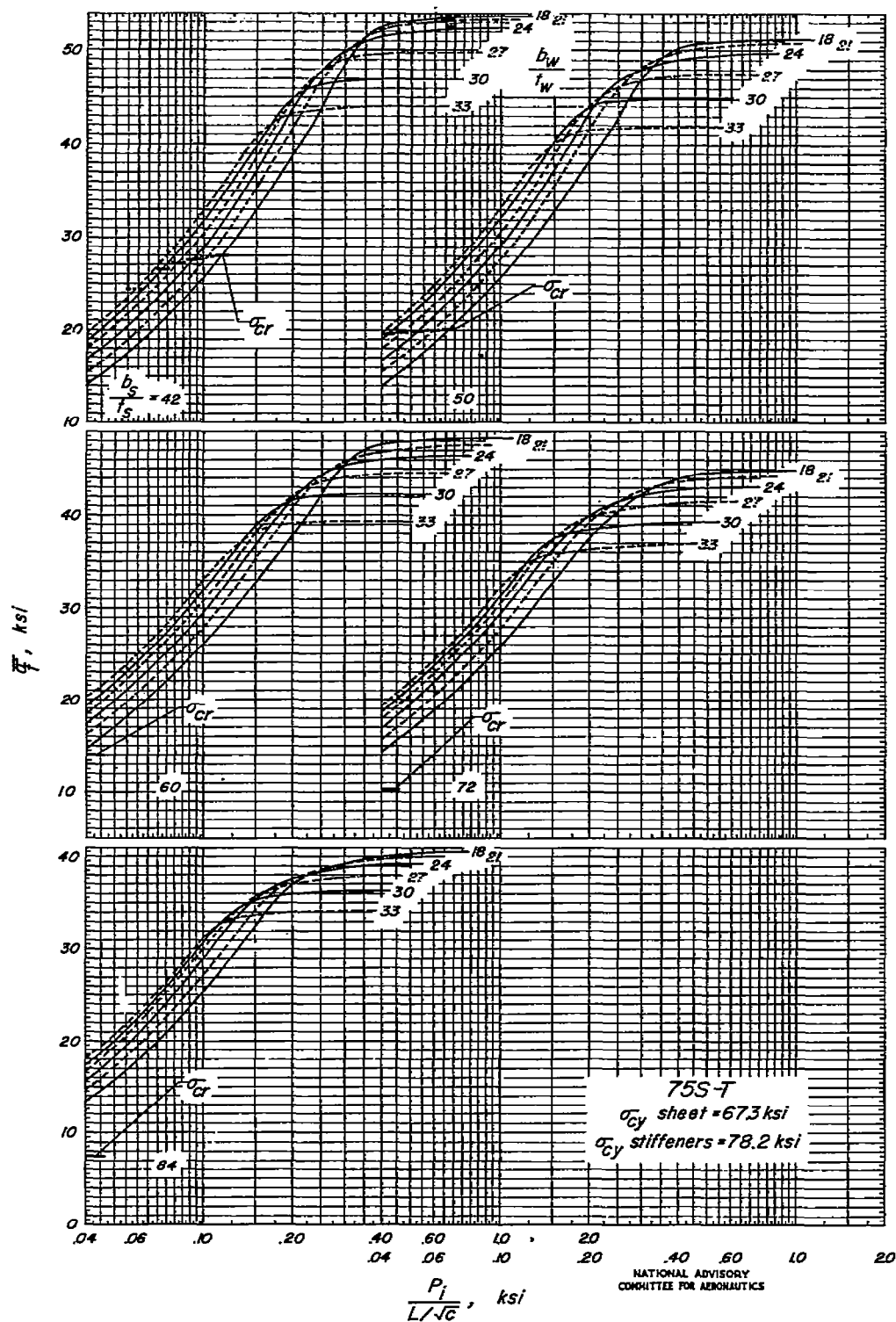


Figure 14.—Concluded.

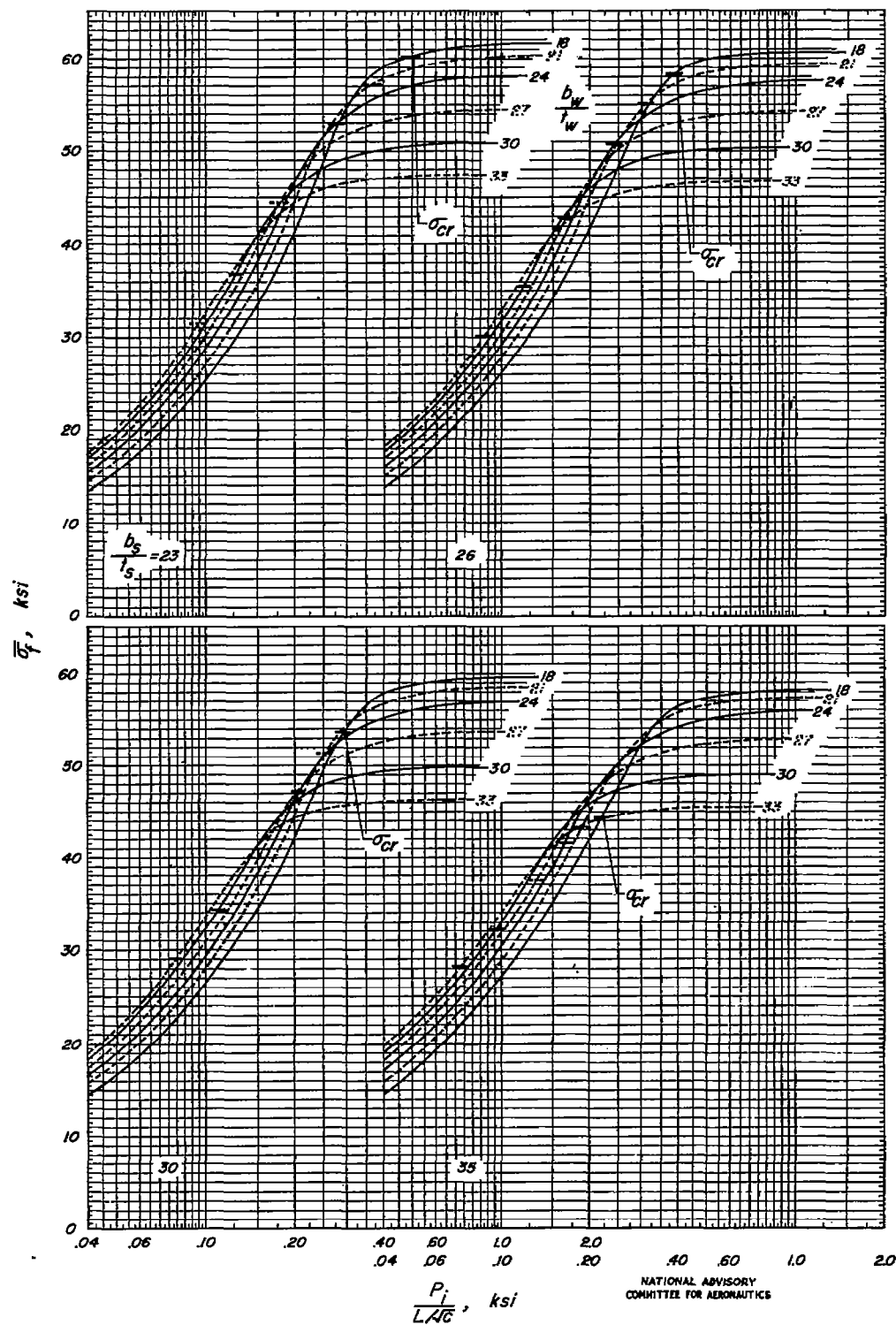


Figure 15.—Design chart for 75S-T Y-panels of the proportions tested.  $\frac{t_w}{t_s} = 0.79$ .

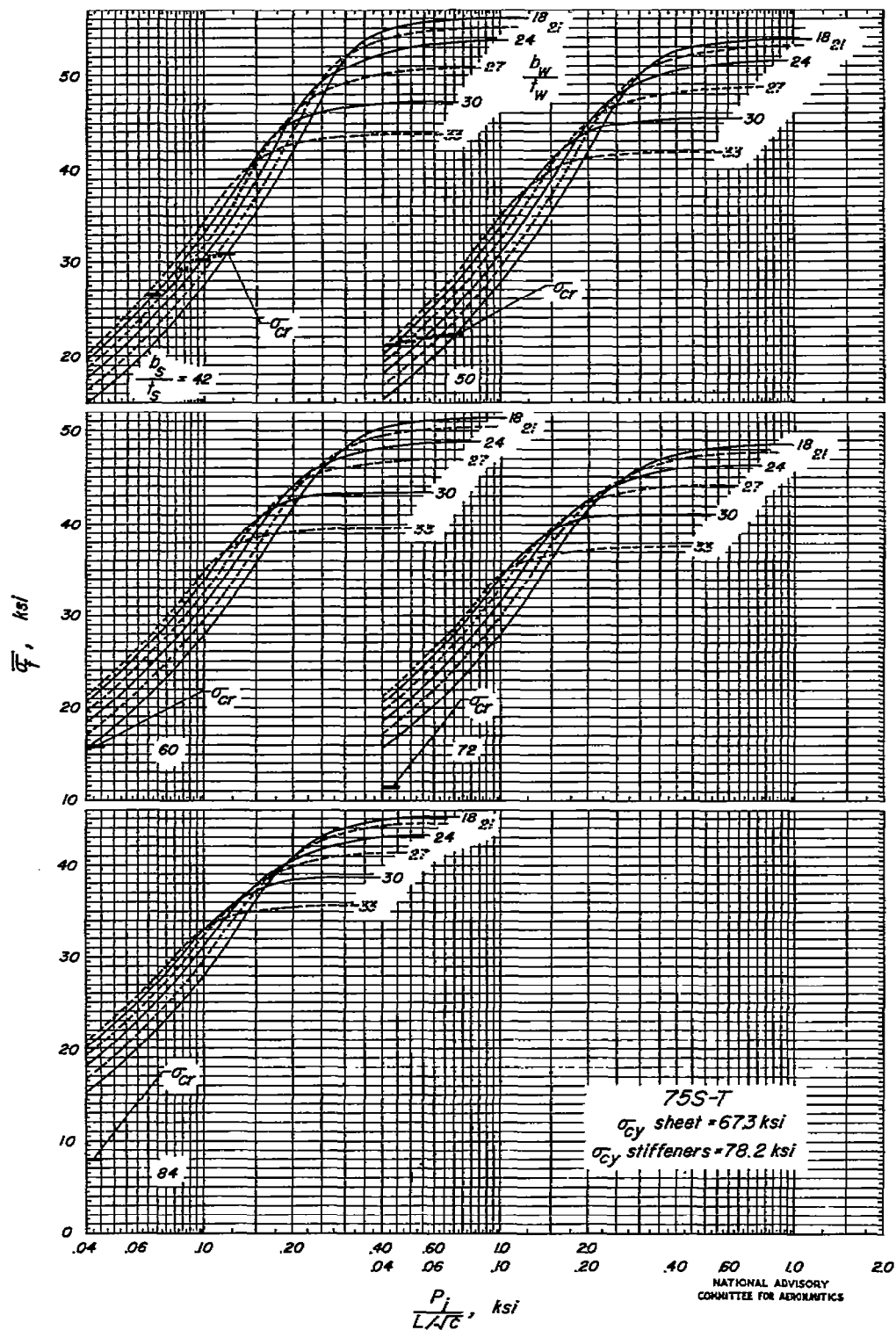


Figure 15.-Concluded.

Fig. 16

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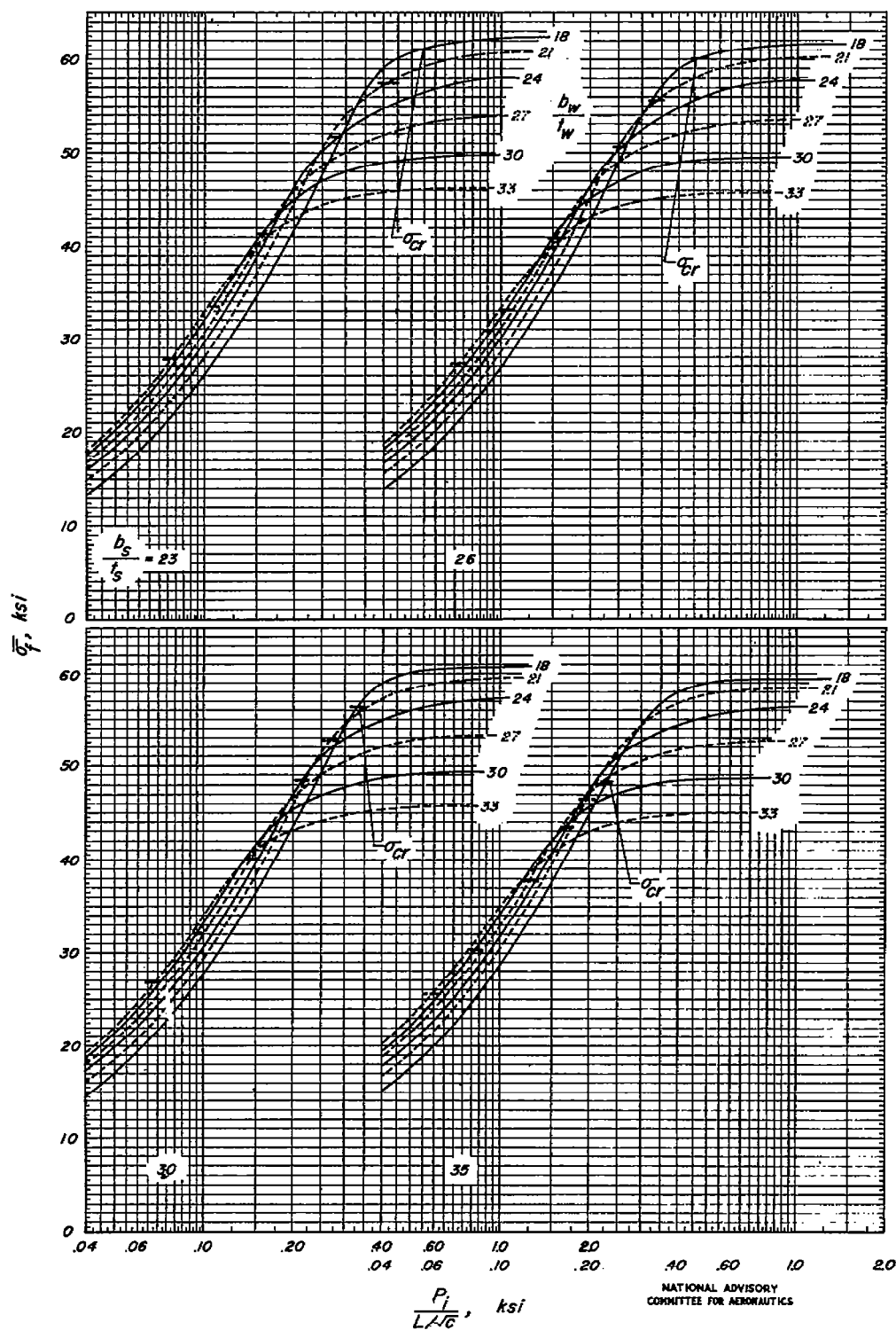


Figure 16.—Design chart for 75S-T Y-panels of the proportions tested,  $\frac{t_w}{t_s} = 1.00$ .

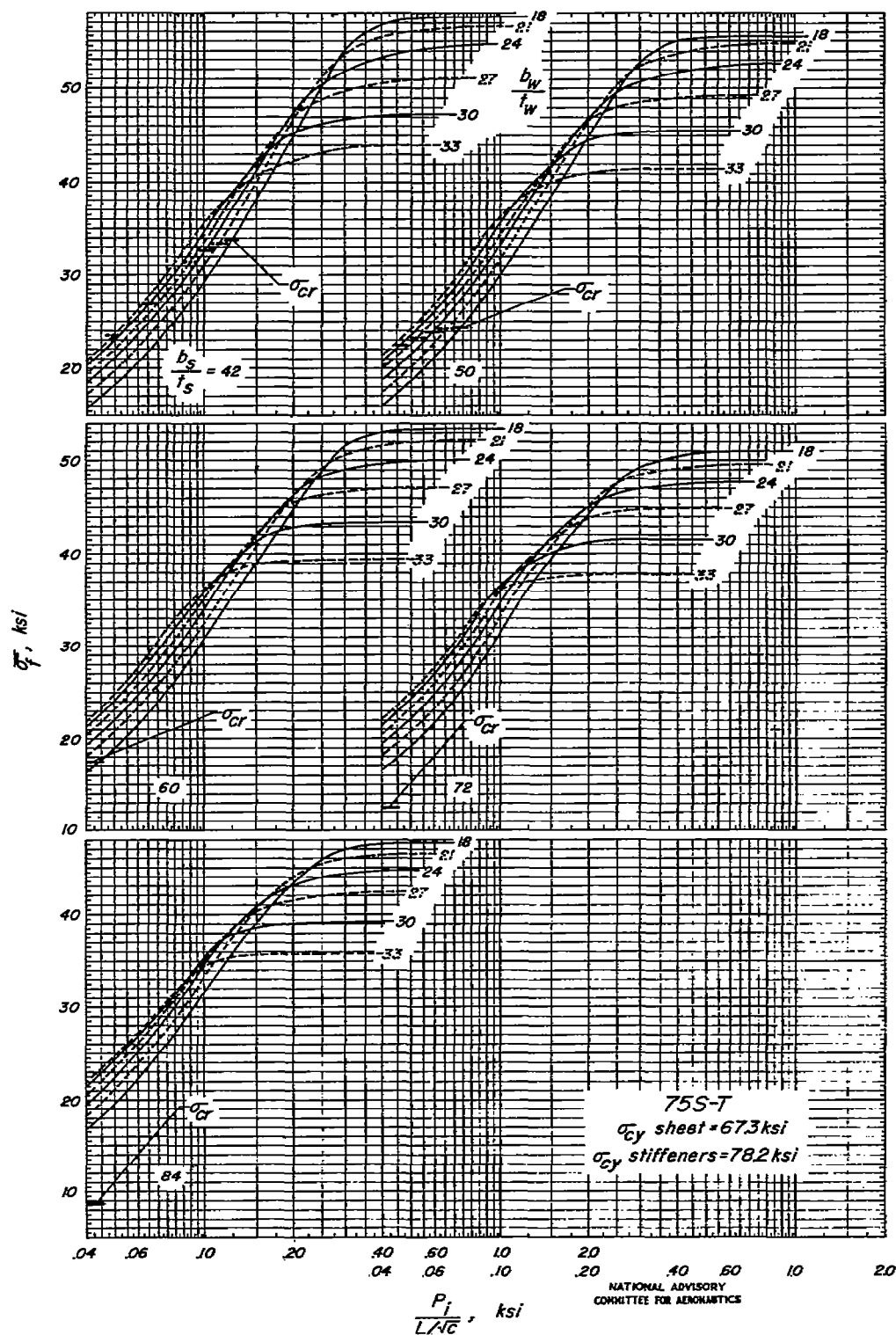


Figure 16.-Concluded.

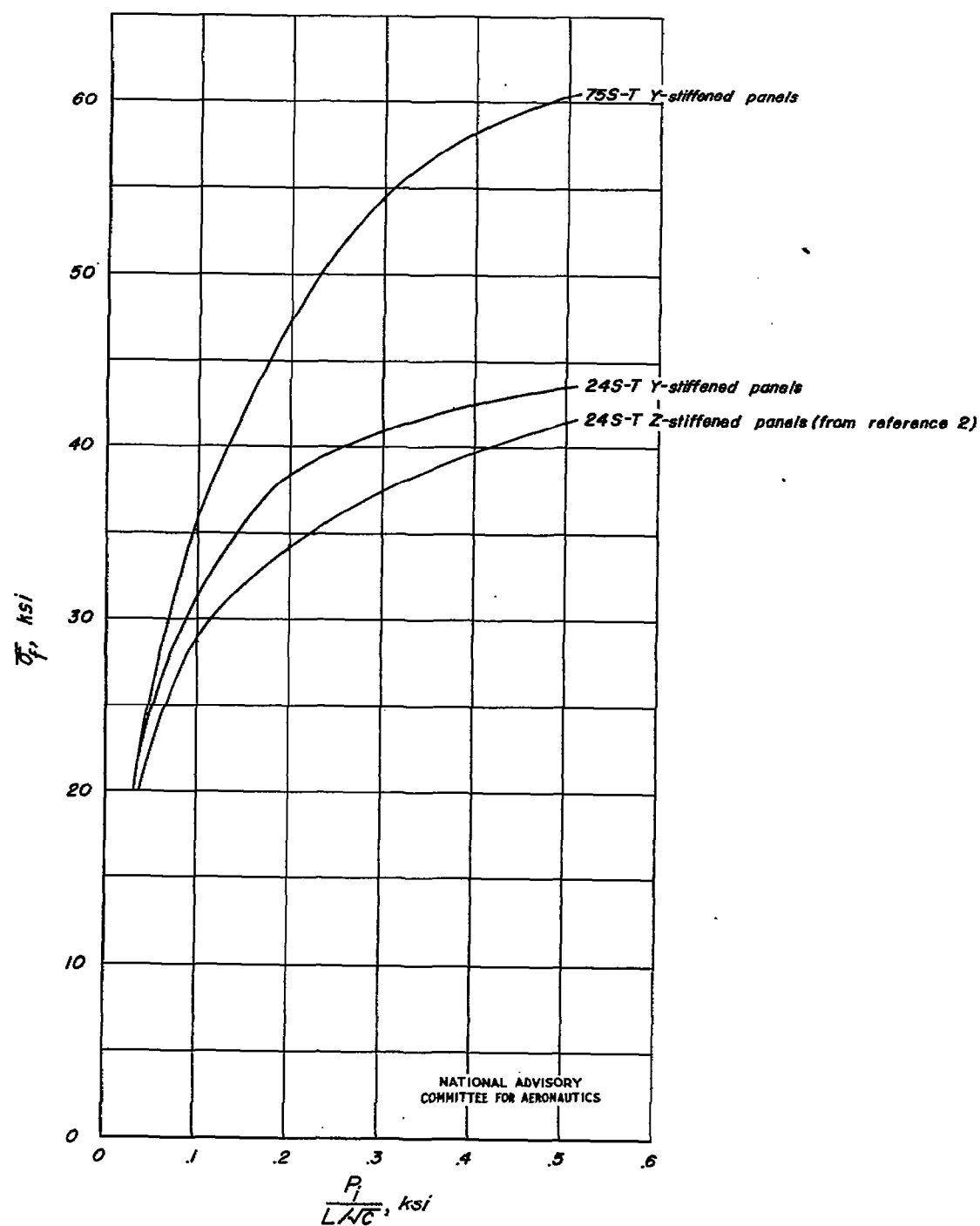


Figure 17.-Comparison of envelope curves for 24S-T Z-stiffened panels (from reference 2) and for 24S-T and 75S-T Y-stiffened panels.

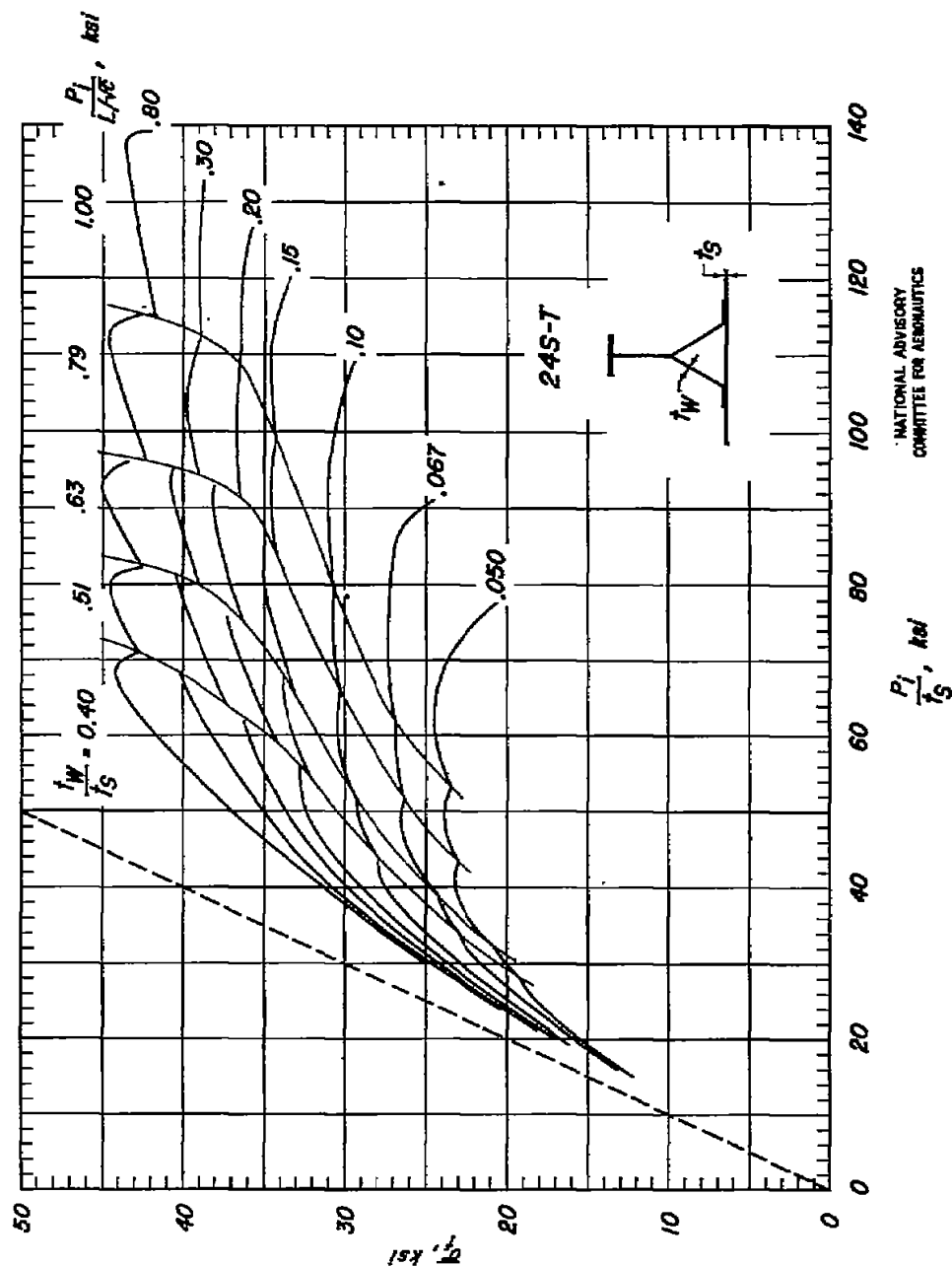


Figure 18. - Design chart for the determination of the average stress at failure that can be carried by minimum-weight designs of 24S-T aluminum-alloy flat compression panels having extruded Y-section stiffeners.



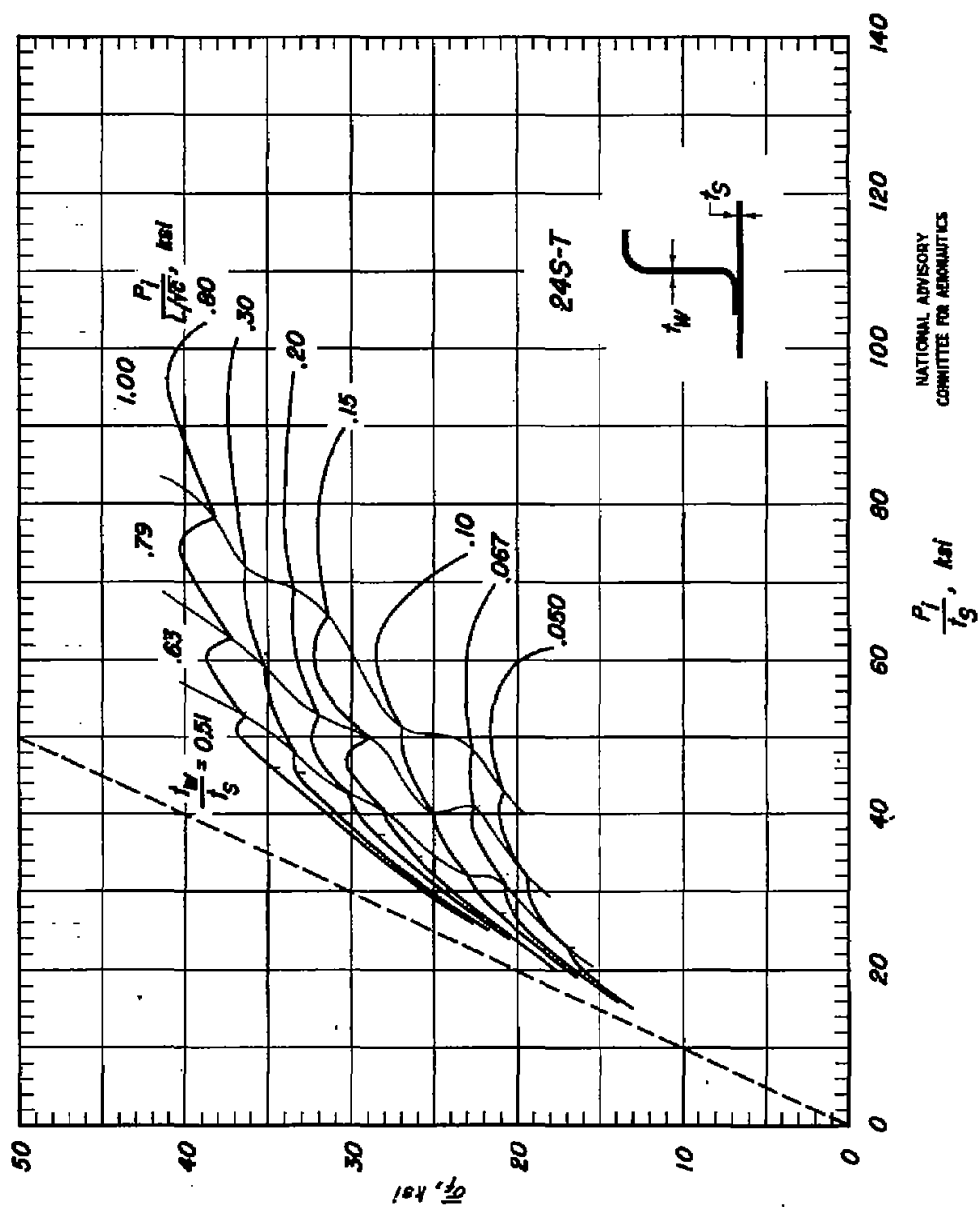


Figure 19.- Design chart for the determination of the average stress at failure that can be carried by minimum-weight designs of 24S-T aluminum-alloy flat compression panels having formed Z-section stiffeners.

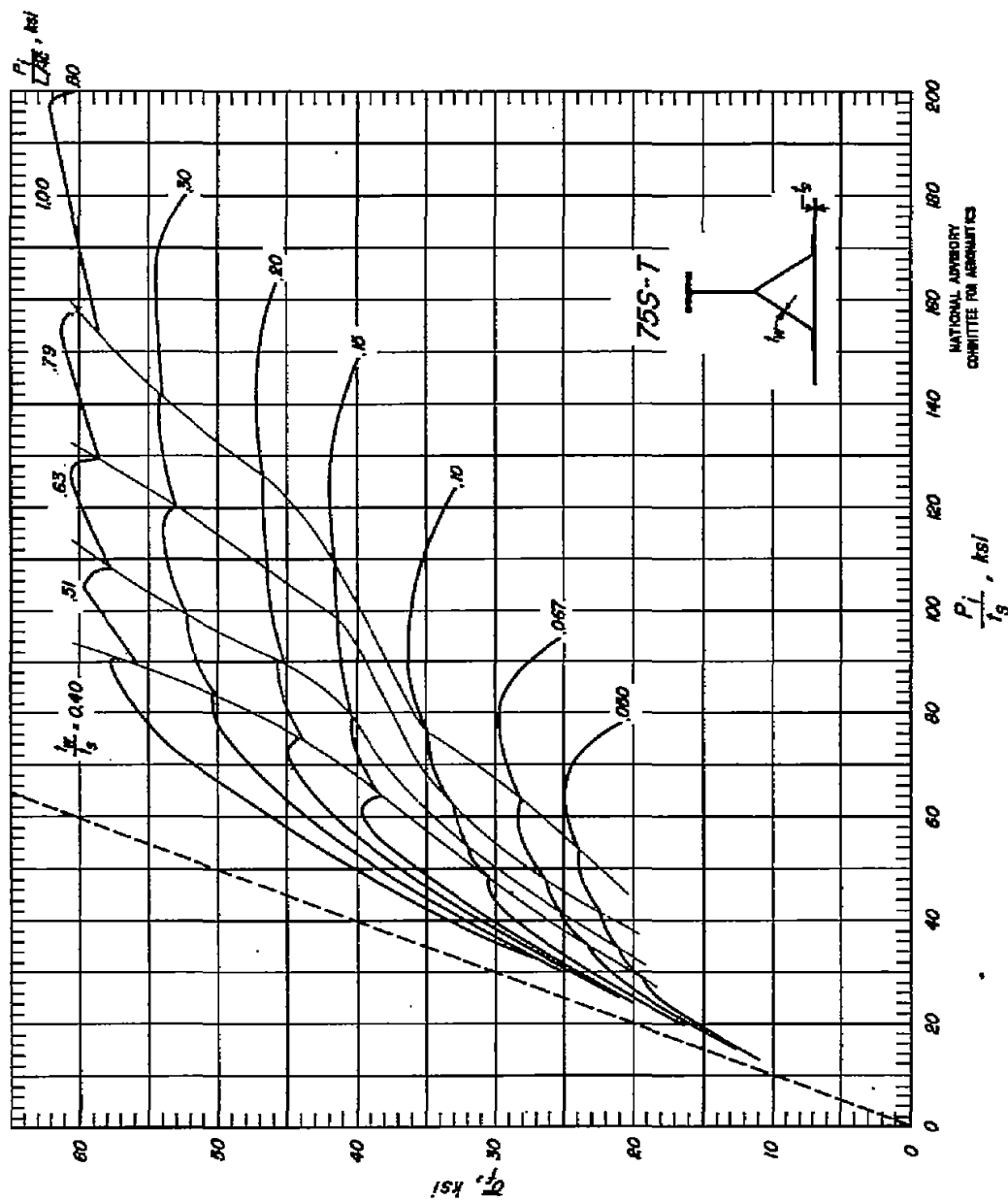
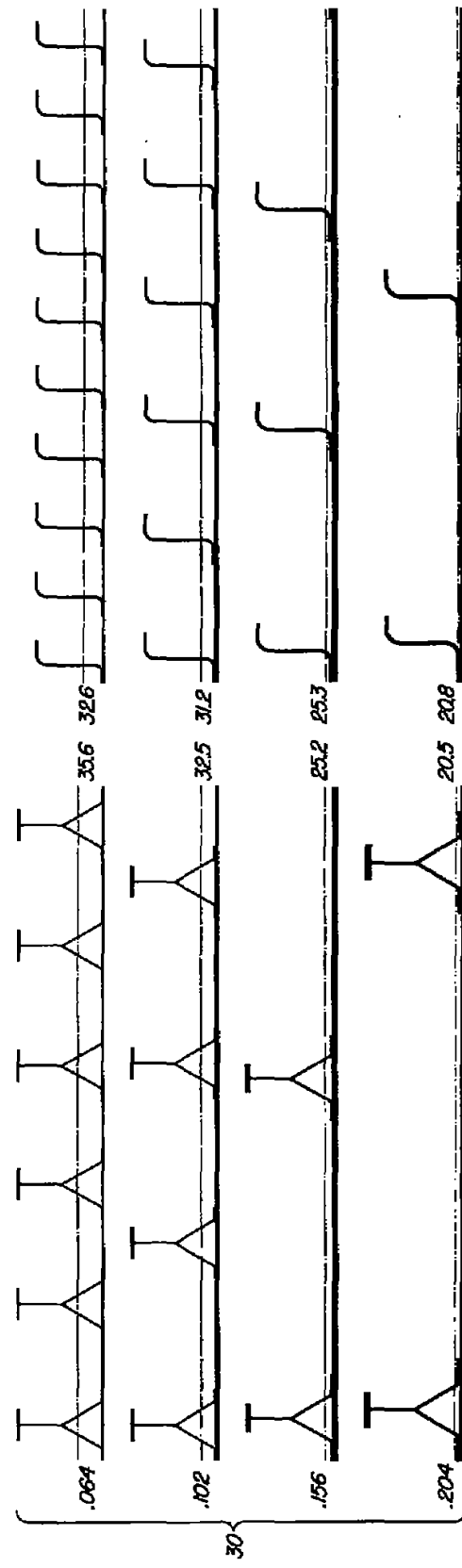
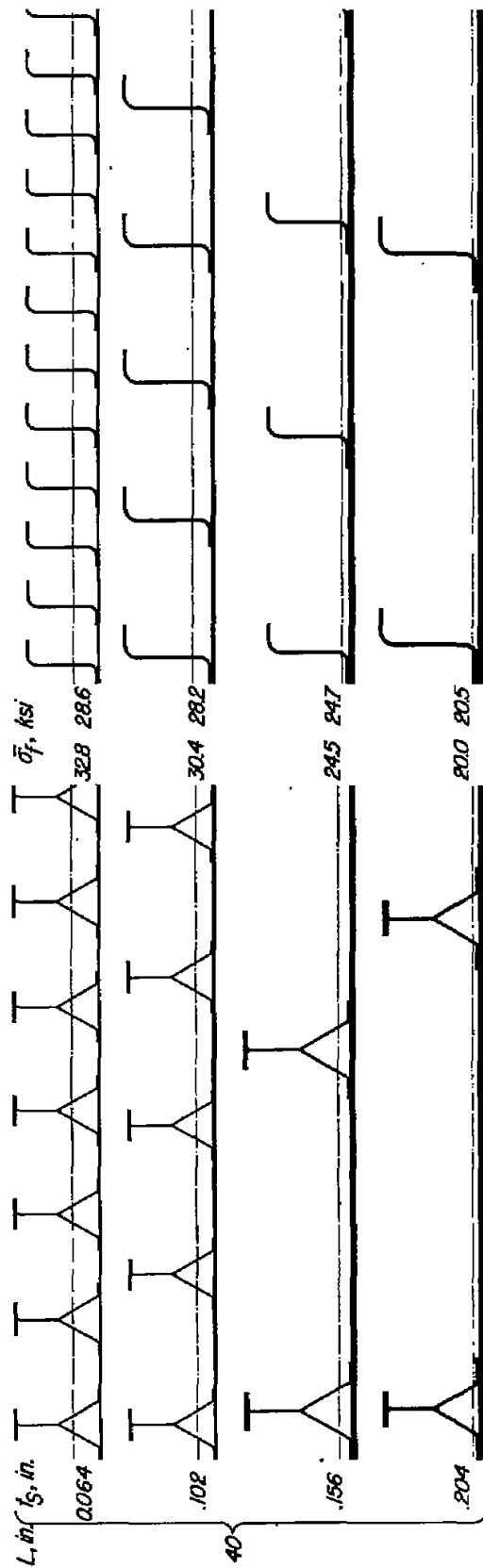


Figure 20-Design chart for the determination of the average stress at failure that can be carried by minimum-weight designs of flat compression panels having Alclad 75S-T aluminum alloy sheet and extruded 75S-T Y-section stiffeners.

Fig. 21

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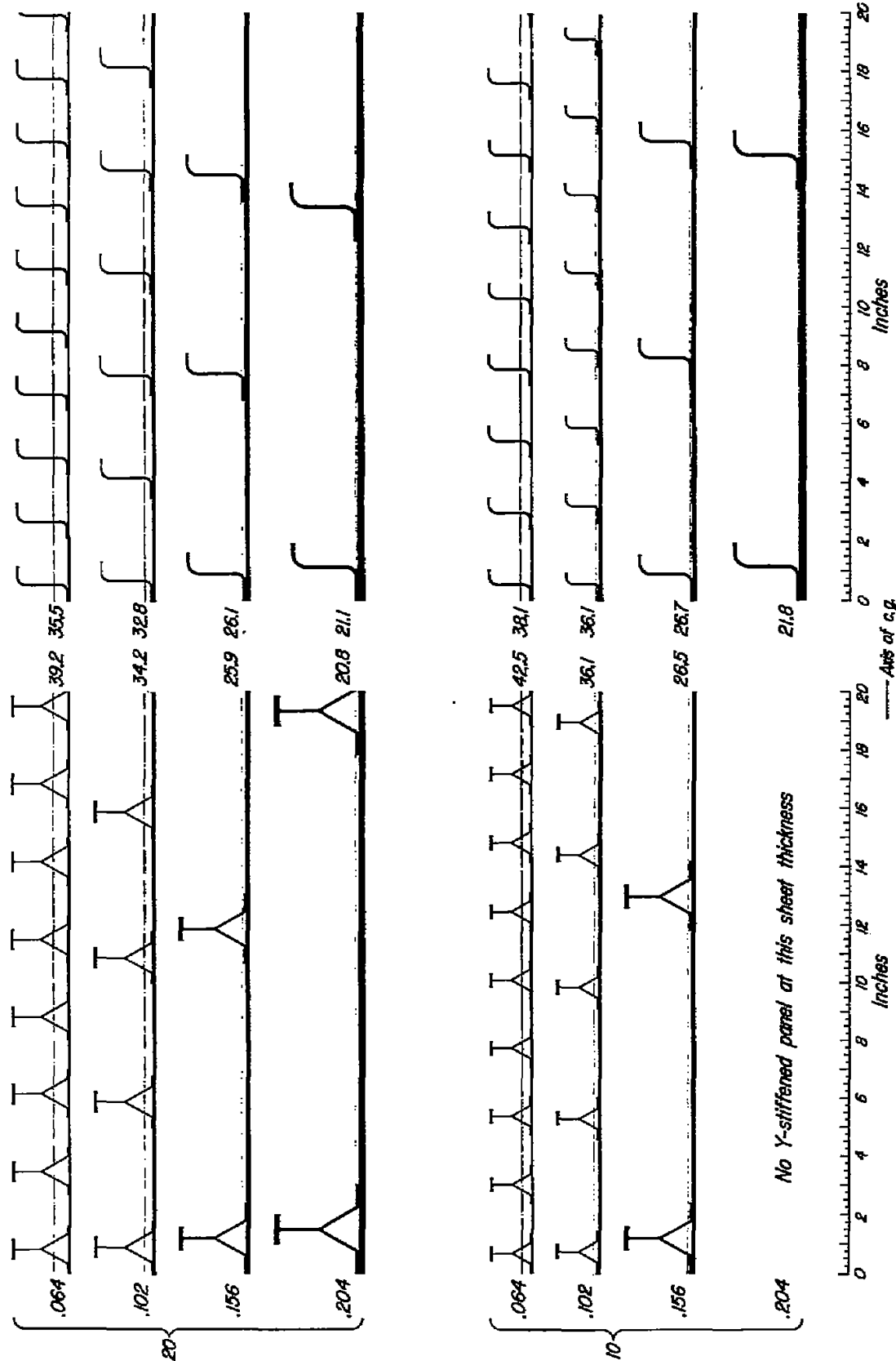
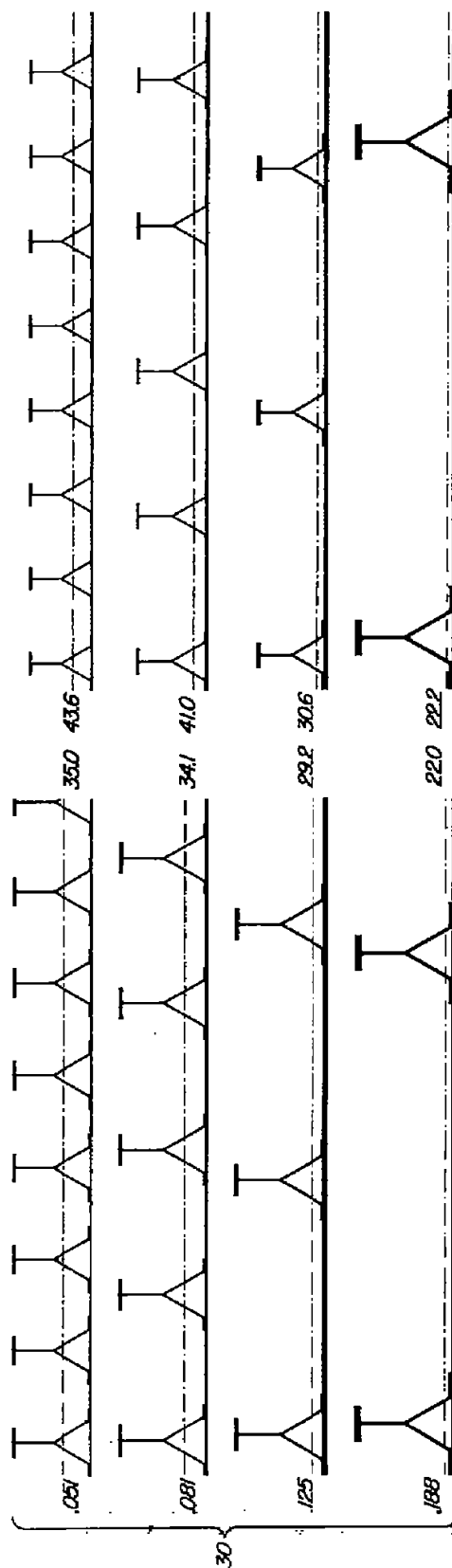
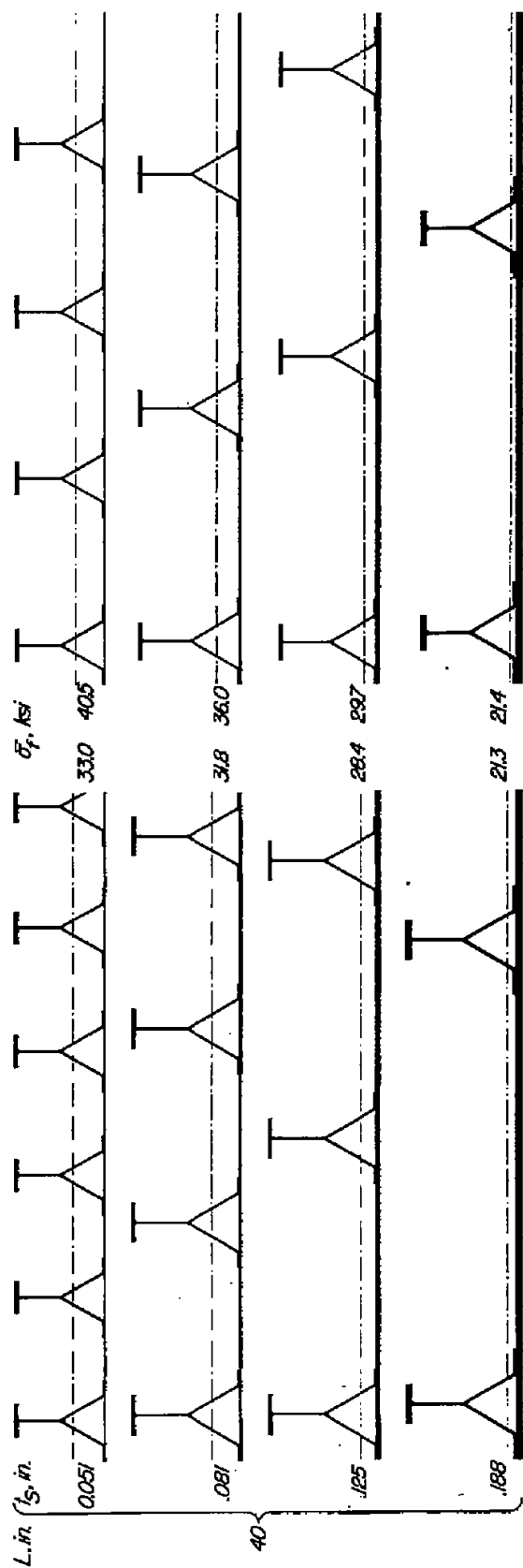


Figure 21.- Comparative minimum-weight designs of Y- and Z-stiffened panels.  $P_1 = 5.0$  kips per inch;  $c = 1$ .

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Fig. 22

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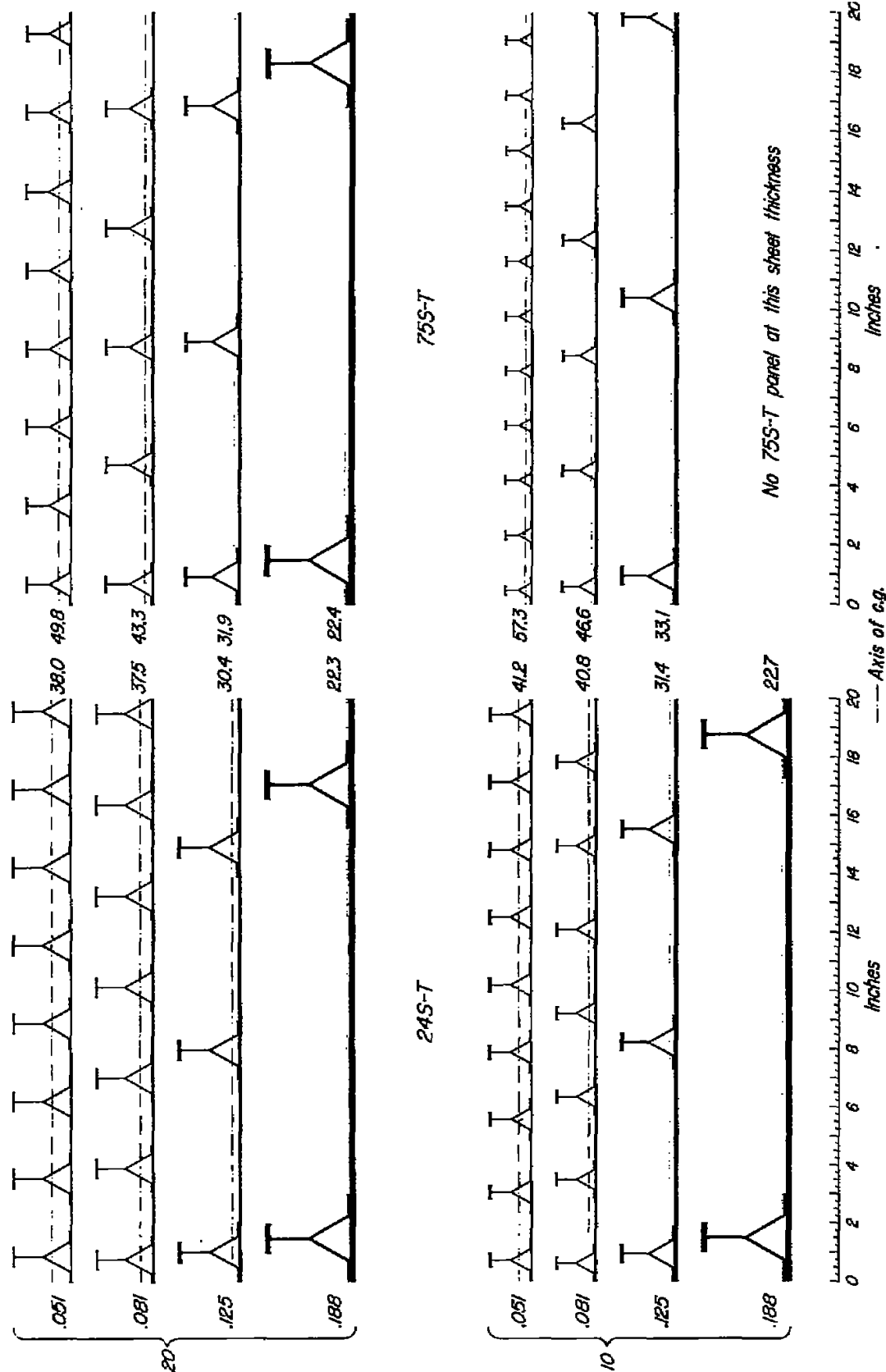


Figure 22.—Comparative minimum-weight designs of 24S-T and 75S-T Y-stiffened panels.  $P_f = 5.0$  kips per inch;  $c = 1$ .

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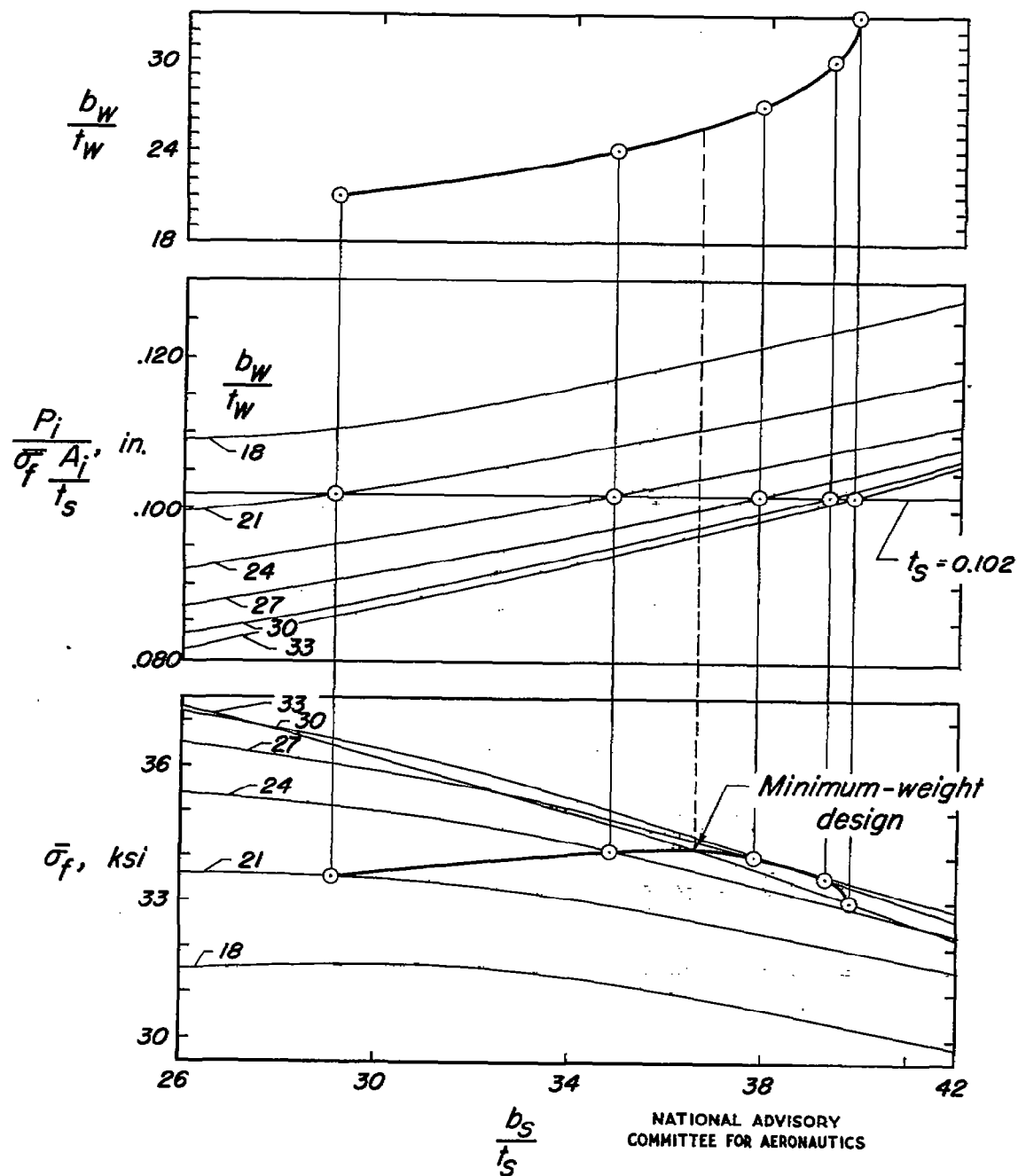
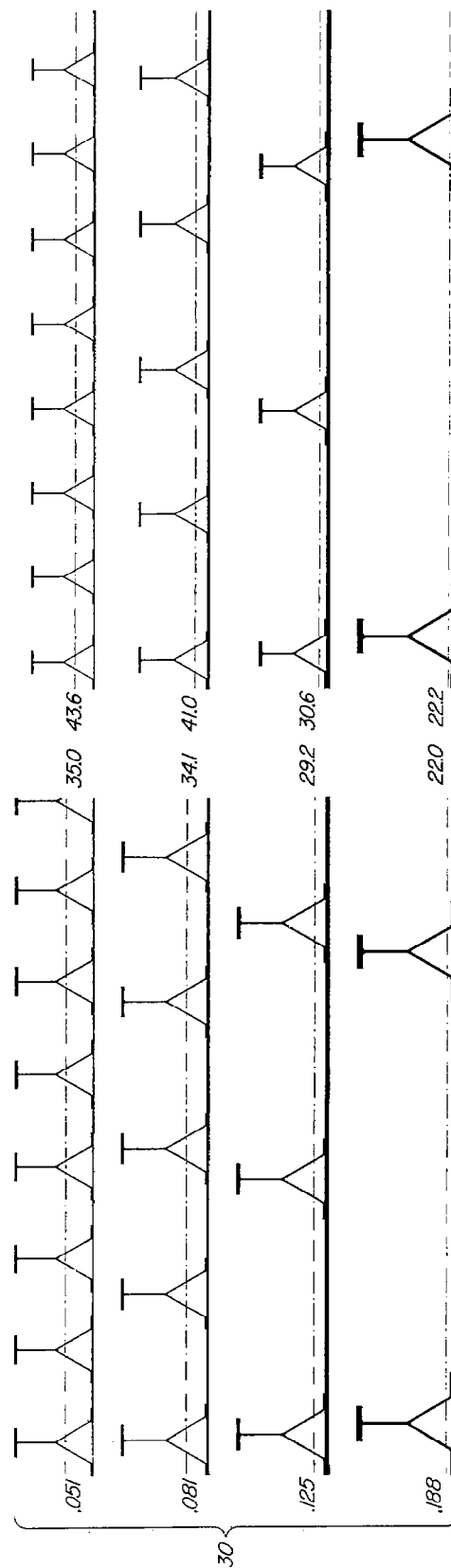
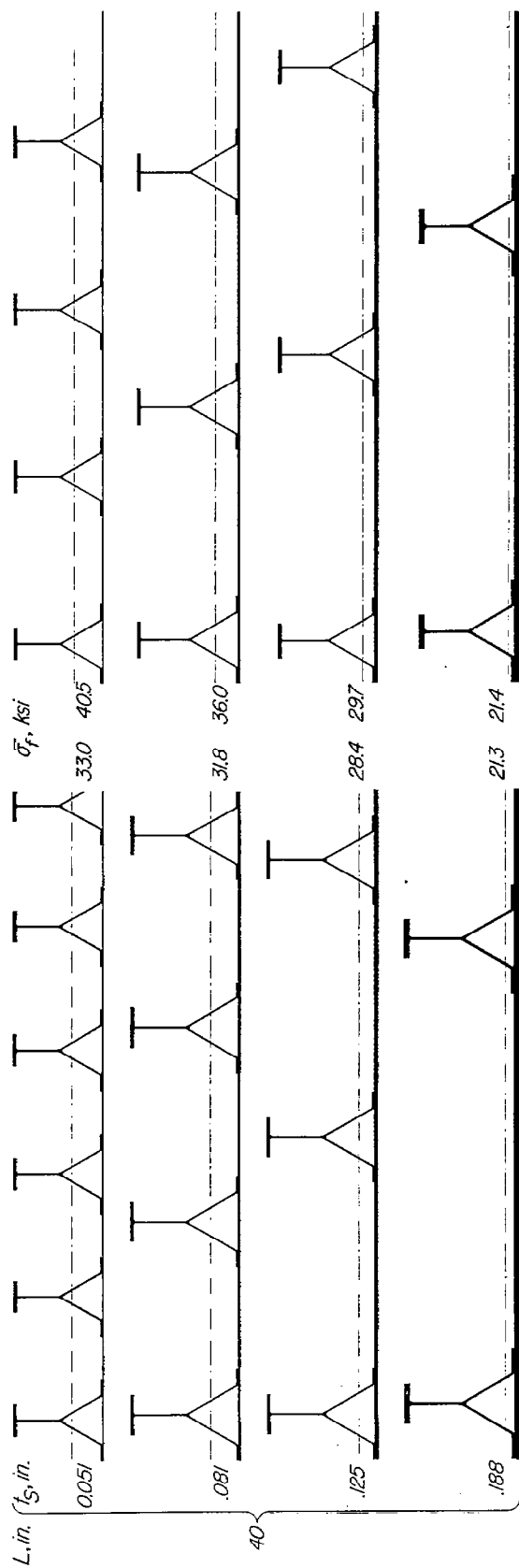


Figure 23- Plot for obtaining design for maximum structural efficiency (minimum weight).  $P_i = 5.0$  kips per inch;  $L = 20$  inches;  $c = 1$ ;  $t_s = 0.102$  inch;  $\frac{t_W}{t_S} = 0.40$ .

Fig. 22

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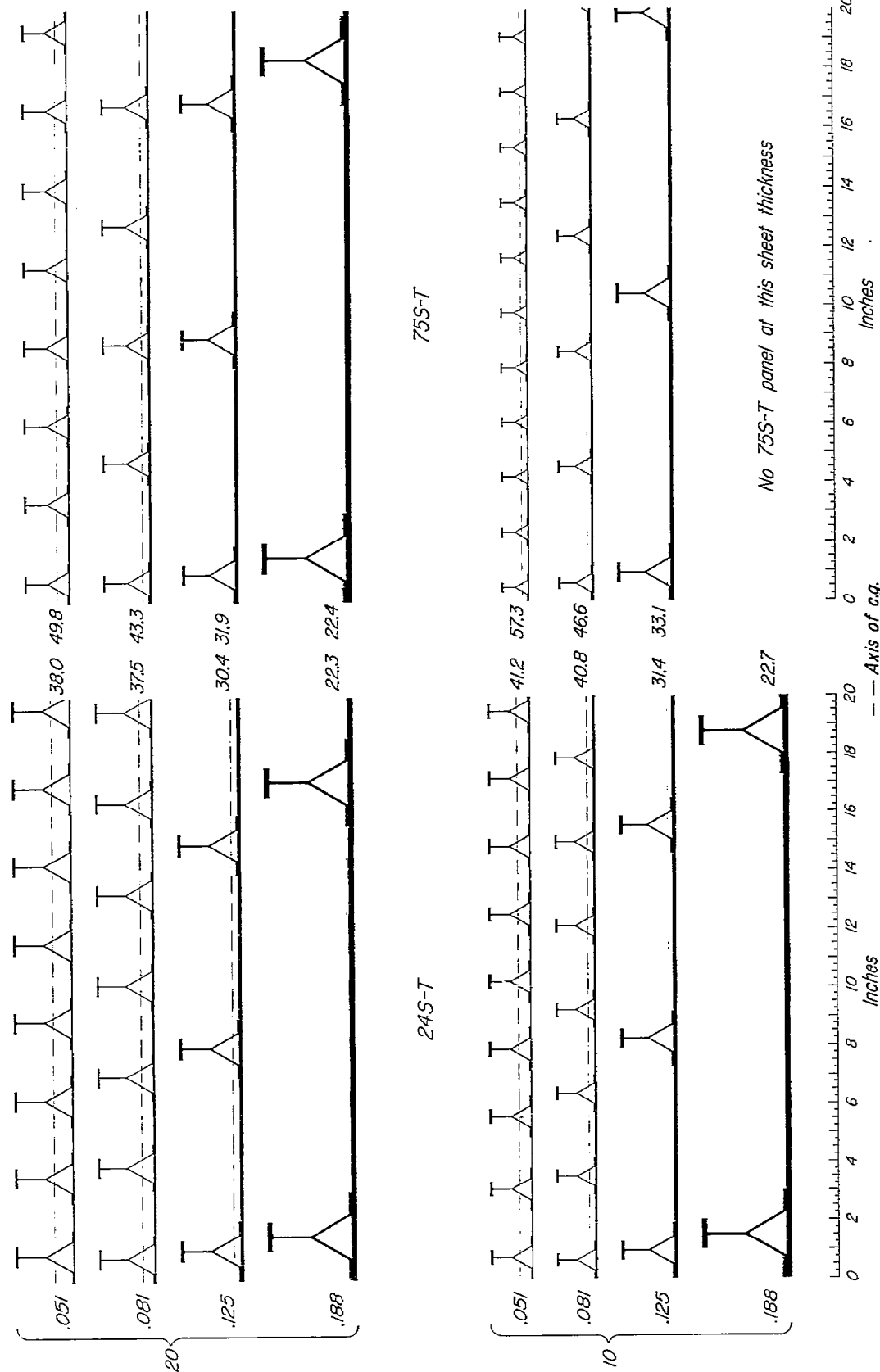


Figure 22.—Comparative minimum-weight designs of 24S-T and 75S-T Y-stiffened panels.  $P_1 = 5.0$  kips per inch;  $c = l$ .

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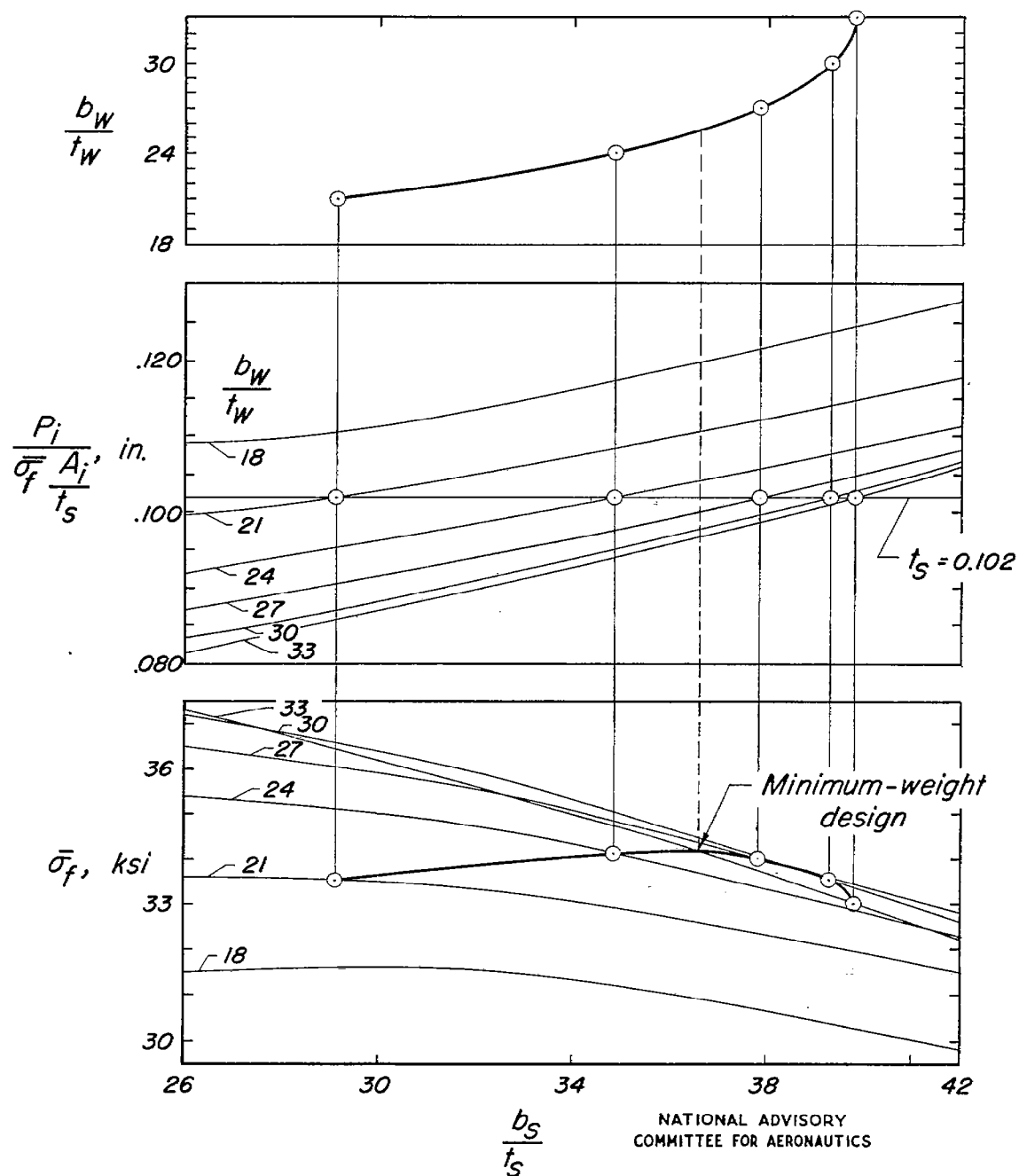


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**ORIGINATING AGENCY:** National Advisory Committee for Aeronautics, Washington, D. C.  
**PUBLISHED BY:** (Same)

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